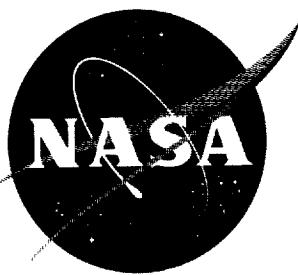


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TECHNICAL MEMORANDUM

X-680

WIND-TUNNEL INVESTIGATION OF THE
DYNAMIC AND STRUCTURAL-DEFLECTION CHARACTERISTICS
OF AN INFLATABLE AIRPLANE

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NATIONAL AERONAUTICS AND SPACE ADMINISTRATION
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SUMMARY

An investigation has been conducted in the Langley full-scale tunnel to determine the aerodynamic and structural-deflection characteristics of a single-place inflatable airplane over a range of test velocities from about 77 ft/sec to 135 ft/sec. Aerodynamic force data, wing-distortion photographs, and wing-guy-cable loads were recorded at each test speed for a range of angles of attack to the stall or to wing buckle.

At a normal inflation pressure of 7.0 lb/sq in. load factors of the order of 4.50 could be obtained for a test velocity of 134.3 ft/sec. For a reduced inflation pressure of 4.8 lb/sq in. a maximum load factor of 3.09 was obtained at a test velocity of 113.0 ft/sec. At an inflation pressure of 2.0 lb/sq in. a maximum load factor of 1.33 was obtained for a test velocity of 78.5 ft/sec, and the airplane would not be safe for flight at such a low inflation pressure.

INTRODUCTION

Because of wing-buckling failures during flight tests of inflatable airplanes, evaluating teams have jointly recommended some changes in the restraint-cable attachment points. These changes are intended to increase the wing load factor before wing buckle. The inflatable airplane is certain to be torn by the propeller and deflated if buckling occurs during flight; therefore, a configuration having a wing cable and structural relationship sufficiently strong to preclude a wing buckle within the normal operating limits of the airplane would certainly be desirable.

The inflatable airplane of reference 1 (identified herein as Inflatoplane I) was previously tested in the Langley full-scale tunnel where the largest load factor obtainable prior to buckle was found to be

about 2.5 at the normal inflation pressure of 7.0 lb/sq in. The intent of the present series of tests of an inflatable airplane (identified herein as Inflatoplane II), which were also made in the Langley full-scale tunnel, was to determine whether load factors of the order of 4.50 to 5.00 could be obtained with appropriate modifications.

The tests of the present investigation were conducted at angles of attack varying from about -8° to that required for maximum lift or for wing buckle. Measurements of the surface-pressure distributions were made for most of the test configurations, and for one of the best configurations the wing structural integrity was determined with full aileron deflection at high wing loadings. The tests were conducted for a range of Reynolds numbers, based on wing chord, varying from 2.57×10^6 to 4.57×10^6 which corresponds to a test velocity range of about 77 ft/sec to 135 ft/sec.

SYMBOLS

The data of the subject report are referred to the stability system of axes the origin of which is located at the model center of gravity located longitudinally at fuselage station 72.7 and vertically at water line 45.3. See figure 1.

b	wing span, ft
c	wing chord, ft
\bar{c}	mean aerodynamic chord, ft
C_D	drag coefficient, Drag/ $q_{\infty} S$
C_L	lift coefficient, Lift/ $q_{\infty} S$
C_l	rolling-moment coefficient, Rolling moment/ $q_{\infty} S b$
C_m	pitching-moment coefficient, Pitching moment/ $q_{\infty} S \bar{c}$
C_n	yawing-moment coefficient, Yawing moment/ $q_{\infty} S b$
C_p	pressure coefficient, $\frac{p_l - p_{\infty}}{q_{\infty}}$

C_Y	side-force coefficient, Side force/ $q_\infty S$
c_n	wing-section normal-force coefficient, $\int_0^{1.0} c_{p,l} d(x/c) - \int_0^{1.0} c_{p,u} d(x/c)$
i_w	mean incidence of wing (angle between wing chord and water line 50, see fig. 1), deg
n	load factor, $\frac{C_L q_\infty}{W/S}$
p	wing and fuselage inflation pressure, lb/sq in.
p_l	local static pressure, lb/sq ft
p_∞	free-stream static pressure, lb/sq ft
q_∞	free-stream dynamic pressure, lb/sq ft
S	wing area, sq ft
V_∞	free-stream velocity, ft/sec
W	weight of airplane, 550 lb
x	chordwise distance measured parallel to the plane of symmetry, ft
y	lateral distance measured perpendicular to the plane of symmetry, ft
α_f	angle of attack of fuselage (angle between relative wind and water line 50, see fig. 1), deg
δ_a	right-hand aileron deflection (positive when trailing edge deflected down), deg
δ_r	rudder deflection (positive when trailing edge deflected toward the pilot's right hand), deg

Subscripts:

u	upper surface
l	lower surface
max	maximum

AIRPLANE AND APPARATUS

The inflatable airplane used in the present investigation was composed of pneumatic structure throughout with the exceptions of the engine, engine mount, landing gear, support cables, and miscellaneous short control members. All inflatable components were interconnected in a manner allowing, during flight operation, a small compressor on the 40-horsepower air-cooled engine to maintain a constant regulated pressure in the system even with moderate leakage. An external air supply was substituted for the normal engine compressor because the engine was not operated during the tests. The wing and tail surfaces are woven in a manner such that the upper and lower airfoil surfaces were connected internally by nylon drop threads varying in length to produce the shape desired in any surface when inflated. A circular cross-sectional fuselage was utilized with a fuel bag internally mounted, and the cockpit section was constructed of sections of an inflatable material 2 inches thick.

Each wing panel was restrained by two guy cables on the upper surface and by either three or four cables attached to the lower surface. The two upper cables were anchored to the engine pylon and the lower cables were attached to the landing gear and to the fuselage rearward of the wing trailing edge. Each guy cable had a calibrated strain-gage link inserted in the cable and the instrumentation was such that all cable loads were recorded simultaneously. A general layout with pertinent geometric data is shown in figure 1, and a photograph of a similar airplane, Inflatoplane I, mounted in the Langley full-scale tunnel is given as figure 2. No such photograph of the present airplane was obtained as the tests were terminated, because of wing failure, before any general photographs were made. The principal external differences between Inflatoplane I and Inflatoplane II are that the configuration used in this investigation has a slightly larger wing, no cockpit canopy, and better shape control of the airfoil contour. The propeller was not installed for the present tests.

The airplane was mounted for tests on the conventional six-component tunnel mechanical balance system. A special yoke was utilized to mount

the airplane so that restraining loads were transmitted to the fuselage through strap attachments located beneath the wing quarter-chord point; thus, the wings were free to deflect while being restrained, as in flight, only by the normal wing-fuselage and guy-cable attachments. Provision was made to change cable configurations by means of extra attachment points beneath the wing. The various lower surface cable attachment points with their number designations are shown in figure 3. Cable configuration was designated in terms of the numbers of the attachment points. For example, a cable configuration designation of 2-5-8 means that three lower surface cables were used and these cables were attached to the wing at positions 2, 5, and 8. The rear mounting strut was attached to a saddle strapped to the rearward portion of the fuselage and was connected by cables to the front support yoke to prevent longitudinal tail strut loads from being transmitted into the fuselage. Restraint cables were mounted above each wing to control the amount of deflection during buckling and, thus, minimize structural damage caused by the buckles.

An actuator system was installed in the cockpit to allow remote operation of the ailerons. Movie and still cameras were set up to record the deflection of the right-hand wing panel under various loading conditions. The right-hand wing panel was chosen for photographic study because the wing contours and geometry were more uniform than those of the left-hand panel, and in the event of wing buckle, the right-hand panel was expected to buckle first.

The right-hand wing panel was equipped with five (10- and 12-tube) plastic belts glued to the wing surface in a chordwise direction with a hole punched in each tube to provide wing-surface orifices for determining the pressure-distribution characteristics of the wing under various loading conditions. The spanwise locations of the pressure-distribution belts were 16.21, 28.95, 50.00, 68.00, and 89.20 percent of the semispan and are referred to hereinafter in this report as stations 1 to 5, respectively. (See fig. 4.) The chordwise locations of the orifices at the 5 spanwise stations are also given in figure 4. The orifice pressures were measured on a multiple-tube manometer and photographically recorded.

TESTS AND CORRECTIONS

The primary objective of the present tests was to determine the maximum loading, prior to wing buckle, of the unmodified Inflatoplane II (cable configuration 2-5-8) and if possible to improve the loading characteristics of the airplane by the addition of a wing guy cable or by relocation of some or all of the wing restraint guy cables. In addition

to the aerodynamic force, the wing-deflection characteristics and the pressure-distribution characteristics were determined for all the test configurations; however, camera malfunction during the tests caused some portions of the photographs to be unreadable. Consequently, complete pressure-distribution data for all stations and test configurations are not presented. The tests were conducted at various airspeeds ranging from approximately 77 ft/sec to 135 ft/sec corresponding to Reynolds numbers, based on wing chord, of 2.57×10^6 to 4.57×10^6 . The angle of attack of the fuselage increased at each airspeed from about -8° to the angle at which the wing stalled or buckled. The wing had an average incidence of 7.2° relative to water line 50. (See fig. 1.) Most of the tests were conducted at the normal inflation pressure of 7.0 lb/sq in.; however, a few tests were conducted at inflation pressures of 4.8 lb/sq in. and 2.0 lb/sq in.

Wing cable loads were recorded for all test configurations, and the wing deflections during most of the tests were photographically recorded by still and movie cameras. The cable positions available for use during the tests are shown in figure 3. The method used for all the tests, after force data on the original configuration were obtained, was to first visually observe the wing stalling or buckling characteristics and the cable loads for a given test velocity. Further force and pressure measurements were then made for only the better configurations. The cable positions shown in figure 3 were utilized in the following manner: one or two cables attached to position 1, 2, or 3; one to position 4, 5, or 6; and one to position 7, 8, or 9. A few visual tests were made for modified cable arrangements having the attachment points at the landing gear, moved forward but these tests did not warrant further investigation because of poor loading characteristics.

For the aileron tests, only the right-hand aileron was deflected so that the effect of individual aileron movement, up or down, could be determined. For the tests with the rudder deflected, the rudder was locked by cables to a fully deflected position of full right rudder.

All the data presented in this paper have been corrected for air-stream misalignment ($\Delta\alpha = -0.5^\circ$; $\Delta C_L = -0.0087 C_L$; $\Delta C_D = -0.0087 C_D$), buoyancy ($\Delta C_D = 0.0$), and jet boundary ($\Delta\alpha = -0.760 C_L$; $\Delta C_D = -0.013 C_L^2$). Support strut tares were not measured since major emphasis was placed on obtaining loads information. All drag results, therefore, include the tare drag of the support system.

RESULTS AND DISCUSSION

A motion-picture film supplement to this paper has been prepared and is available on loan. A request card form and a description of the

film will be found at the back of this paper on the page immediately preceding the abstract and index pages.

Static Longitudinal and Wing-Buckling Characteristics

The variations of the lift, drag, and pitching-moment characteristics of the inflatable airplane for the normal inflation pressure of 7.0 lb/sq in., for several test airspeeds and two wing-guy-cable configurations are shown in figures 5 and 6. The data of figure 5 were obtained by utilizing the basic airplane as received from the manufacturer. In general the data of figure 5 show that the airplane is longitudinally stable through the stall. The right-hand wing buckled just prior to $C_{L,\max}$ at the test velocity of 124.3 ft/sec which resulted in a load factor of 4.19. The maximum load factor achieved for a speed at which the airplane reached stall was 3.63 at a velocity of 112.7 ft/sec.

In an attempt to increase the load on the wing prior to stall or buckle, qualitative tests were made of various cable arrangements. Apparently, the best of the arrangements tested was configuration 1-3-6-8. During these tests, however, the largest load factor obtained was only about 4.22, which was below the desired value of 4.50 to 5.00.

It was noted that the landing-gear structure was deflecting into the belly of the fuselage just prior to wing buckle. This deflection was attributed to the wing-guy-cable load because the guy cable was anchored to the landing-gear structure. This deflection would effectively lengthen the guy cable which would in turn allow the wing to deflect more and thus buckle more readily. A 9-inch-square, 1/2-inch-thick plate of aluminum was placed between the landing-gear tubular structure and the inflated belly of the fuselage to provide additional load-bearing area and, thus, minimize the landing-gear deflection under cable load. In addition to the belly plate, the guy cables were heavily tightened in the static condition. The results of the tests made with these modifications to cable configuration 1-3-6-8 are shown in figure 6. The maximum load factor at a velocity of 123.6 ft/sec remained at 4.22 which indicated that the cable and plate modification had not improved the load-carrying capabilities of the airplane. The wing deflection occurred in a different manner with the belly plate installed; however, to be sure of the best cable configuration, the cables were rerigged to the original cable configuration 2-5-8 with the belly plate installed and with the guy cables heavily tightened. For this configuration at a velocity of 122.0 ft/sec the wing did not buckle and the load factor at $C_{L,\max}$ was 4.17. The test velocity was arbitrarily increased to 134.3 ft/sec to produce a buckle in order to determine the maximum load factor of cable configuration 2-5-8. The data for this configuration are shown in figure 6. The load factor for this test condition was 4.50 which was within the desired load range so no further attempts were made to increase the airplane load factor.

Two types of buckles were noticed during the tests. A root buckle occurred when the larger of two shear wrinkles at the outboard edge of the fuselage bulkhead straps lengthened to about the quarter chord. These wrinkles first progressed rearward and outward at about 45° and then straight rearward. As the wrinkles neared the quarter chord several short wrinkles appeared parallel to the long wrinkle just before buckle occurred at the root section. A patch buckle (buckle at cable attachment point) occurred just over one of the patches that supported the lower cables. Depressions in the wing upper surface above the cable attachment points can be seen in the film supplement. As the cable load increased the depressions deepened. A patch buckle occurred when numerous short chordwise wrinkles appeared in the depressed areas of the inboard cable attachment points. A patch buckle was usually preceded by an erratic rise and fall of the portion of the wing outboard of the outboard cable. An example of a typical load buildup and buckle is shown in the photographs of figure 7. This figure also shows the effect of the forward and rearward cables over the wing in limiting the upward movement of the wing which thus minimized damage when the wing buckled.

In order to determine whether the fuselage and wing structure would be strong enough to withstand flight at some reduced speed with a reduced inflation pressure, several tests were conducted at reduced inflation pressures. The results of these tests are shown in figure 8. At an inflation pressure of 4.8 lb/sq in. and a test speed of 93.6 ft/sec the wing reached $C_{L,\max}$ and a load factor of 2.32 without buckling. The velocity was arbitrarily increased to 113.0 ft/sec and the maximum load factor obtained at buckle was 3.09. At an inflation pressure of 2.0 lb/sq in. the maximum load factor reached at buckle was 1.33 at a velocity of 78.5 ft/sec. At this speed one would be flying very close to buckle in unaccelerated level flight; therefore, an inflation pressure of 2.0 lb/sq in. should be considered to be too low for flight.

During some of the high-loading tests, which were repeated to obtain pressure-distribution data that were not obtained earlier because of camera malfunction, the wing tore loose from the wing-fuselage bulkhead. The wing was damaged severely and the test program was terminated. It was later found during study of the motion-picture film that the failure resulted from a broken right-hand wing guy cable which had probably been damaged during previous buckle tests. The broken cable was attached to the wing in the number 5 position. Incidentally, when the cable broke the load factor was estimated to be about 4.32, and the cable load was estimated to be approximately 1,100 pounds, which is well below the design breaking strength of about 3,700 pounds.

Static Lateral Characteristics

The static lateral characteristics of the airplane with right-hand aileron deflections of -15° , 0° , and 26° are shown in figures 9 and 10. Deflection of one aileron was done to isolate the effects of up and down deflection. The reason for the reduced rolling moment with aileron deflection and with increased speed is that the deflected aileron was twisting the wing. Incidentally, the motion-picture film of the aileron tests shows considerable aileron flutter or buffet. One contributing factor to the small amount of flutter which occurred during some of the tests simulating normal flight conditions was weak bungee chords in the aileron restraint system. The problems that could arise in the event of aileron flutter should definitely be recognized and appropriate steps should be taken to prevent their occurrence in flight.

Aileron deflection is seen to produce adverse yaw (fig. 10). Tests for rudder effectiveness were not made, but visual tests for fuselage torsional stiffness were made for several airspeeds with the rudder fully deflected and single-point data were obtained for each airspeed. The data, figure 11, showed that the rudder would provide adequate moments to counteract the adverse yaw and still have sufficient power for maneuvering.

Aerodynamic wing-guy-cable loads for several of the test conditions are given without analysis in figures 12 to 15. The initial loading of the cables for the zero-speed condition was not determined. For each test the load recording instruments were set at zero load for the zero-speed condition; thus, the data show the change in cable loading caused by aerodynamic forces and moments.

Pressure-Distribution Characteristics

A complete listing of the pressure coefficients obtained during the subject investigation is given in tables 1 to 16. A listing of the tables with pertinent information concerning them is as follows:

Table	Cable configuration	Guy-cable tension	Belly plate	q_{∞} , lb/sq ft	P_r , lb/sq in.	α_r , deg	δ_a , deg
1	2-5-8	Light	Off	6.7	7.0	-8.5 to 11.8	0
2	2-5-8	Light	Off	9.9	7.0	-8.5 to 12.9	0
3	2-5-8	Light	Off	14.2	7.0	-8.5 to 11.9	0
4	1-3-6-8	Light	Off	17.1	7.0	-8.4 to 3.7	0
5	1-3-6-8	Heavy	On	17.0	7.0	-6.6 to 4.7	0
6	2-5-8	Heavy	On	6.9	7.0	-8.5 to 11.8	0
7	2-5-8	Heavy	On	10.1	7.0	-8.5 to 11.9	0
8	2-5-8	Heavy	On	14.1	7.0	-8.5 to 10.9	0
9	2-5-8	Heavy	On	16.9	7.0	-6.5 to 9.4	0
10	2-5-8	Heavy	On	20.1	7.0	-6.6 to 2.2	0
11	2-5-8	Light	Off	9.9	4.8	-8.5 to 9.8	0
12	2-5-8	Light	Off	11.6	4.8	-8.6 to 1.8	0
13	2-5-8	Light	Off	6.9	2.0	-8.6 to -1.1	0
14	2-5-8	Heavy	On	7.0	7.0	-4.8 to 10.9	-15 to 26
15	2-5-8	Heavy	On	10.2 or 10.8	7.0	-4.8 to 6.7	-15 to 26
16	2-5-8	Heavy	On	14.4 or 14.7	7.0	-4.8 to 7.4	-15 to 26

The wing-surface-pressure distributions are shown in figure 16 for two cable configurations. The data of figure 16 are not intended to show detailed differences in the loading characteristics of the two configurations but are intended to be representative diagrams of the loading of the inflatable wing for varying angles of attack and air-speeds. For detailed analysis, use must be made of the tabulated values of the pressure coefficients.

The pressure-distribution characteristics of the wing with aileron deflection and for two test velocity conditions are shown in figure 17. As one might expect of an inflatable wing, aileron deflection is seen to adversely affect the chordwise loading on the wing forward of the aileron which is the result of wing twist caused by the aileron deflection.

Span Loading Characteristics

Complete span loading plots cannot be made because the spanwise orifice stations were not close enough to the airplane vertical plane of symmetry to accurately determine the fairing of the curves for the inboard locations and because of camera malfunction some of the data for the inboard stations were not obtained; however, the general span loading characteristics of the configurations are shown in figures 18 and 19. Apparently, the reason that the loading shown in figure 18(c) at $\alpha_f = 2.2^\circ$ is lower than that at $\alpha_f = 1.8^\circ$ is that the manometer was photographed just as the wing buckled, because the data of figure 6 and visual observations of the wing showed that maximum lift should have been experienced at $\alpha_f = 2.2^\circ$.

The general span loading characteristics of the wing with the right-hand aileron deflected are shown in figure 19 for test velocities of 94.0 ft/sec and 113.7 ft/sec.

CONCLUDING REMARKS

The results of wind-tunnel tests of an inflatable airplane in the Langley full-scale tunnel indicate that by proper selection of the attachment points of the wing restraint guy cables, by restricting the movement of the lower cable attachment point into the belly of the fuselage, and by heavily tightening the lower guy cables in the static condition, a load factor of 4.50 at a normal inflation pressure of 7.0 lb/sq in. can be obtained prior to buckle for a test velocity of 134.3 ft/sec. For an inflation pressure of 4.8 lb/sq in. a maximum load factor of 3.09 was obtained at a test velocity of 113.0 ft/sec but

for an inflation pressure of 2.0 lb/sq in. the maximum load factor obtained at a test velocity of 78.5 ft/sec was 1.33 which is considered to be too low for flight.

Langley Research Center,
National Aeronautics and Space Administration,
Langley Air Force Base, Va., February 13, 1962.

REFERENCE

1. Cocke, Bennie W., Jr.: Wind-Tunnel Investigation of the Aerodynamic and Structural Deflection Characteristics of the Goodyear Inflatable plane. NACA RM L58E09, 1958.

TABLE I

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.5^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper				
	.0000	.848	.938	.893
	.0100	.125	.304	.411
	.0250	-.714	-.348	-.089
	.0500	-.518	-.366	-.214
	.1000	-.625	-.482	-.393
	.2250	-.607	-.473	-.384
	.4500	-.554	-.357	-.295
Lower	.7500	-.304	-.464	-.321
	.0500	-.732	-.750	-.696
	.1500	-.607	-.580	-.491
	.4000	-.348	-.286	-.188
	.7000	-.143	.045	.036
Aileron				
Upper	.8050	-.384	-.482	-.357
	.8150	-.357	-.313	-.339
	.8350	-.268	-.259	-.411
	.8750	-.179	-.268	-.250
	.9250	-.089	-.107	-.063
	.9750	-.000	-.027	-.018
Lower	.8100	-.000	.125	.223
	.8340	.152	.259	.179
	.9250	-.036	.089	.080
	.9750	-.018	.027	.045

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened

(b) $\alpha_f = -4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper		.0000	1.045	.973	.884	
		.0100	-.339	-.205	-.054	
		.0250	-.902	-.964	-.554	
		.0500	-.902	-.964	-.679	
		.1000	-.625	-.813	-.643	
		.2250	-.473	-.571	-.554	
		.4500	-.366	-.446	-.384	
		.7500	-.188	-.509	-.393	
Lower		.0500	-.098	-.339	-.393	
		.1500	-.107	-.366	-.366	
		.4000	-.107	-.286	-.188	
		.7000	.125	.071	-.036	
Aileron						
Upper		.8050	-.071	-.482	-.402	
		.8150	-.054	-.339	-.411	
		.8350	-.107	-.286	-.527	
		.8750	-.036	-.304	-.321	
		.9250	.125	-.152	-.179	
		.9750	.125	-.018	-.054	
Lower		.8100	.250	.214	.196	
		.8340	.161	.232	.152	
		.9250	.161	.063	.018	
		.9750	.196	-.018	-.018	

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $a_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(c) \quad \alpha_f = -0.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper		.0000	.295	.759	.813	
		.0100	-1.688	-1.304	-0.777	
		.0250	-2.188	-2.054	-1.241	
		.0500	-1.911	-1.893	-1.205	
		.1000	-1.473	-1.205	-0.938	
		.2250	-0.839	-0.813	-0.643	
		.4500	-0.589	-0.500	-0.375	
		.7500	-0.170	-0.464	-0.375	
Lower		.0500	.330	.259	.071	
		.1500	.188	-0.063	-0.098	
		.4000	-0.080	-0.063	-0.071	
		.7000	-0.080	.152	-0.009	
Aileron						
Upper		.8050	-0.223	-0.402	-0.348	
		.8150	-0.223	-0.286	-0.357	
		.8350	-0.223	-0.188	-0.473	
		.8750	-0.116	-0.179	-0.295	
		.9250	-0.045	-0.098	-0.188	
		.9750	-0.045	-0.054	-0.098	
Lower		.8100	.143	.286	.205	
		.8340	.134	.277	.152	
		.9250	.080	.080	-0.045	
		.9750	-0.063	.045	-0.063	

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(d) \quad \alpha_f = 2.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing					Wing	
Upper	.0000		-.446	.143	.393	
	.0100		-3.098	-2.500	-1.679	
	.0250		-3.652	-3.071	-1.929	
	.0500		-3.179	-2.875	-1.804	
	.1000		-2.107	-1.929	-1.196	
	.2250		-1.321	-1.259	-.938	-.768
	.4500		-.714	-.705	-.509	-.366
	.7500		-.339	-.321	-.321	-.446
	.8350		.161			
	.8750		.107			
	.9250		.116			
	.9750		.071			
Lower	.0500		.661			
	.1500		.286			
	.4000		.071			
	.7000		-.054			
	.8530		-.036			
	.9250		-.071			
	.9750		-.071			
Aileron						
Upper	.8050		-.321	-.384	-.464	
	.8150		-.196	-.080	-.429	
	.8350		-.205	-.098	-.509	
	.8750		-.089	-.107	-.366	
	.9250		-.018	-.098	-.250	
	.9750		-.089	-.098	-.196	
Lower	.8100		.161	.313	.170	
	.8340		.214	.366	.161	
	.9250		-.036	.098	-.089	
	.9750		-.089	.000	-.170	

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(e) \quad \alpha_f = 6.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.634	-1.598	Upper	.0000	-1.741	-.741	-.223
	.0100	-4.643	-4.857		.0100	-4.473	-3.830	-2.625
	.0250	-4.482	-5.063		.0250	-4.786	-4.196	-2.696
	.0500	-4.089	-4.366		.0500	-4.107	-3.866	-2.464
	.1000	-2.304	-2.393		.1000	-2.589	-2.071	-1.482
	.2250	-1.357	-1.339		.2250	-1.393	-1.116	-.839
	.4500	-.920	-.741		.4500	-.732	-.563	-.438
	.7500	-.268	-.464		.7500	-.304	-.277	-.589
	.8350	-.063	-.054					
	.8750	.018	-.107					
	.9250	-.107	-.089					
	.9750	.018	.018					
Lower	.0500	.866	.866	Lower	.0500	.714	.857	.634
	.1500	.589	.384		.1500	.393	.482	.241
	.4000	.232	.089		.4000	.045	.196	.027
	.7000	.152	.125		.7000	.018	.268	-.063
	.8530	.063	-.107					
	.9250	-.188	-.152					
	.9750	-.071	-.170					
Aileron								
Upper	.8050	-.446	-.259	-.670				
	.8150	-.241	-.098	-.598				
	.8350	-.232	-.080	-.750				
	.8750	-.161	-.107	-.589				
	.9250	-.054	-.080	-.420				
	.9750	-.205	-.071	-.313				
Lower	.8100	.080	.330	.080				
	.8340	.232	.366	.116				
	.9250	-.143	.152	-.098				
	.9750	-.080	.045	-.179				

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(f) \quad \alpha_f = 8.6^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000	-2.464	-2.420	
	.0100	-5.652	-5.991	
	.0250	-5.205	-5.848	
	.0500	-4.268	-4.821	
	.1000	-2.384	-2.777	
	.2250	-1.589	-1.393	
	.4500	-.679	-.741	
	.7500	-.286	-.188	
	.8350	-.045	.152	
	.8750	-.036	-.125	
	.9250	-.143	-.143	
	.9750	-.080	-.036	
Lower	.0500	.813	.929	
	.1500	.411	.732	
	.4000	.330	.161	
	.7000	.179	.080	
	.8530	-.045	.143	
	.9250	-.196	-.045	
	.9750	-.107	-.080	
Wing				
Upper	.0000	-2.500	-1.268	-.527
	.0100	-5.607	-4.616	-3.143
	.0250	-5.598	-4.875	-3.116
	.0500	-4.643	-4.518	-2.821
	.1000	-2.661	-2.295	-1.634
	.2250	-1.482	-1.232	-.911
	.4500	-.768	-.607	-.491
Lower	.7500	-.286	-.286	-.652
	.0500	.830	.938	.714
	.1500	.607	.554	.313
	.4000	.268	.196	.018
	.7000	.125	.259	-.107
Aileron				
Upper	.8050	-.250	-.268	-.821
	.8150	-.196	-.143	-.661
	.8350	-.170	-.098	-.875
	.8750	-.143	-.125	-.714
	.9250	-.125	-.107	-.446
	.9750	-.098	-.107	-.357
Lower	.8100	.295	.348	.036
	.8340	.330	.348	.036
	.9250	-.018	.125	-.179
	.9750	-.000	.009	-.250

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(g) \quad \alpha_f = 9.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper						
	.0000	-2.223	-1.348	-.643		
	.0100	-5.277	-4.723	-3.205		
	.0250	-5.679	-4.893	-3.125		
	.0500	-4.420	-4.607	-2.848		
	.1000	-2.571	-2.304	-1.670		
	.2250	-1.402	-1.313	-.955		
	.4500	-.634	-.634	-.500		
Lower	.7500	-.098	-.366	-.732		
	.0500	1.018	.875	.652		
	.1500	.670	.482	.277		
	.4000	.366	.161	.009		
Lower	.7000	-.098	.161	-.125		
Aileron						
Upper						
	.8050	-.098	-.304	-.813		
	.8150	-.161	-.161	-.643		
	.8350	.045	-.107	-.839		
	.8750	-.054	-.143	-.723		
	.9250	-.063	-.161	-.482		
	.9750	-.036	-.179	-.357		
Lower						
	.8100	.170	.250	-.027		
	.8340	.295	.375	.054		
	.9250	.152	.045	-.161		
Lower	.9750	-.116	-.036	-.259		

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(h) \quad \alpha_f = 10.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.804	-2.125			
	.0100	-4.518	-5.402			
	.0250	-4.277	-5.411			
	.0500	-3.661	-4.545			
	.1000	-2.179	-2.455			
	.2250	-1.295	-1.295			
	.4500	-.804	-.571			
	.7500	-.473	-.143			
	.8350	-.491	-.009			
	.8750	-.339	-.188			
	.9250	-.339	-.232			
	.9750	-.232	-.134			
Lower	.0500	.875	.821			
	.1500	.696	.518			
	.4000	.420	.161			
	.7000	.045	.098			
	.8530	.018	-.009			
	.9250	-.339	-.009			
	.9750	-.214	-.134			
Aileron						
Upper	.8050	-.313	-.232	-.857		
	.8150	-.277	-.134	-.554		
	.8350	-.205	-.089	-.866		
	.8750	-.152	-.143	-.750		
	.9250	-.098	-.125	-.455		
	.9750	-.098	-.116	-.330		
Lower	.8100	.170	.277	-.027		
	.8340	.277	.366	.036		
	.9250	-.036	.098	-.161		
	.9750	-.116	.009	-.277		

TABLE I.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.7 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(i) $\alpha_f = 11.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:			C_p for values of y/b of:
		0.1621	0.2895		
Wing					
Upper	.0000	-2.036	-2.232		
	.0100	-4.643	-5.661		
	.0250	-4.277	-5.625		
	.0500	-3.518	-4.438		
	.1000	-2.125	-2.527		
	.2250	-1.393	-1.250		
	.4500	-.580	-.688		
	.7500	-.688	-.446		
	.8350	-.393	-.089		
	.8750	-.679	-.179		
	.9250	-.536	-.134		
	.9750	-.482	-.196		
Lower	.0500	.652	.866		
	.1500	.250	.634		
	.4000	.223	.268		
	.7000	-.071	.125		
	.8530	-.143	-.089		
	.9250	-.509	-.170		
Aileron					
	Upper	.8050	-.286	-.241	-.839
		.8150	-.259	-.098	-.563
		.8350	-.196	-.036	-.857
		.8750	-.179	-.143	-.732
		.9250	-.098	-.089	-.482
		.9750	-.134	-.098	-.321
	Lower	.8100	.179	.286	-.027
		.8340	.250	.420	.018
		.9250	-.027	.098	-.161
		.9750	-.107	.000	-.277

TABLE II

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.5^0$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper				.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.849 .349 -.193 -.205 -.572 -.464 -.404 -.373	.880 .512 -.114 -.199 -.349 	.873 .536 .030 -.072 -.301 -.337 -.259 -.349
Lower				.0500 .1500 .4000 .7000	-.729 -.536 -.398 -.084	-.855 -.572 -.283 .054	-.783 -.482 -.199 .066
Aileron							
Upper				.8050 .8150 .8350 .8750 .9250 .9750	-.386 -.361 -.289 -.193 -.108 -.064	-.476 -.289 -.066 -.283 -.084 -.006	-.343 -.319 -.422 -.217 -.054 .024
Lower				.8100 .8340 .9250 .9750	.006 .012 .024 .060	.175 .253 .096 .078	.193 .181 .066 .066

TABLE II.- Continued
 CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON
 $[\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9$ lb/sq ft; belly plate off;
 $p = 7.0$ lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = -0.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper		-.000 -2.024 -2.922 -2.289 -1.681 -.886 -.663 -.331 -.060 -.084 -.060 .030		.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.157 -1.898 -2.410 -2.199 -1.633 -.964 -.620 -.235	.645 -1.349 -2.054 -1.910 -1.259 -.801 -.518 -.446	.771 -1.807 -1.253 -1.241 -.946 -.663 -.373 -.380
				.0500 .1500 .4000 .7000	.470 .151 -.096 .024	.271 .024 -.054 .120	.066 -1.090 -1.090 -.006
Lower		.470 .084 -.181 -.157 .000 -.042 .024					
Aileron							
Upper				.8050 .8150 .8350 .8750 .9250 .9750	-.325 -.235 -.211 -.199 -.036 -.024	-.392 -.331 -.090 -.175 -.096 -.060	-.349 -.355 -.500 -.343 -.199 -.151
Lower				.8100 .8340 .9250 .9750	.018 .169 .036 .024	.241 .319 .078 .018	.193 .163 -.030 -.114

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(d) $\alpha_f = 2.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-3.120	-3.410	.0000	-3.084	.084
	.0100	-3.355	-3.867	.0100	-3.440	-2.518
	.0250	-2.861	-3.313	.0250	-2.934	-3.139
	.0500	-1.795	-1.946	.0500	-1.922	-2.819
	.1000	-.831	-1.006	.1000	-.922	-1.657
	.2250	-.548	-.566	.2250	-.458	-.970
	.4500	-.271	-.295	.4500	-.127	-.542
	.7500	.500	.873	.7500	.831	-.361
	.8350	-.078	.054			
	.8750	-.048	.072			
	.9250	.084	.145			
	.9750	-.072	.145			
Lower	.0500	.536	.476	.0500	.416	.608
	.1500	.440	.295	.1500	.127	.259
	.4000	.199	.193	.4000	.139	.042
	.7000	.000	.139	.7000	-.259	.205
	.8530	-.048	.127			
	.9250	-.084	.078			
	.9750	-.030	.090			
Aileron						
Upper	.8050	-.223	-.325	.8050	-.223	-.464
	.8150	-.133	-.325	.8150	-.133	-.392
	.8350	-.078	-.108	.8350	-.078	-.602
	.8750	-.018	-.120	.8750	-.018	-.428
	.9250	.036	-.072	.9250	.036	-.253
	.9750	.193		.9750	.193	-.181
Lower	.8100	.247	.271	.8100	.247	.127
	.8340	-.018	.331	.8340	-.018	.139
	.9250	.054	.102	.9250	.054	-.127
	.9750	.006	.024	.9750	.006	-.193

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(e) $\alpha_f = 6.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing						Wing		
Upper	.0000	-1.849	-1.753	Upper	.0000	-1.886	-.813	-.265
	.0100	-4.747	-5.241		.0100	-4.795	-3.964	-2.699
	.0250	-4.578	-5.217		.0250	-4.934	-4.361	-2.699
	.0500	-3.970	-4.319		.0500	-4.175	-4.036	-2.476
	.1000	-2.337	-2.542		.1000	-2.530	-2.120	-1.506
	.2250	-1.277	-1.331		.2250	-1.301	-1.169	-.880
	.4500	-.663	-.639		.4500	-.711	-.584	-.446
	.7500	-.133	-.205		.7500	-.253	-.301	-.608
	.8350	-.181	.024					
	.8750	-.078	-.108					
	.9250	-.139	-.060					
	.9750	-.084	.024					
Lower	.0500	.783	1.006	Lower	.0500	.855	.807	.639
	.1500	.488	.633		.1500	.536	.434	.217
	.4000	.169	.277		.4000	.108	.151	-.012
	.7000	.157	.120		.7000	.139	.175	-.120
	.8530	.072	.024					
	.9250	-.084	.054					
	.9750	-.060	-.042					
Aileron								
Upper	.8050	-.241	-.295	Upper	.8050	-.241	-.295	-.645
	.8150	-.175	-.319		.8150	-.175	-.319	-.584
	.8350	-.163	-.114		.8350	-.163	-.114	-.735
	.8750	-.120	-.114		.8750	-.120	-.114	-.614
	.9250	-.096	-.084		.9250	-.096	-.084	-.380
	.9750	-.096	-.078		.9750	-.096	-.078	-.289
Lower	.8100	.169	.301	Lower	.8100	.169	.301	.030
	.8340	.301	.265		.8340	.301	.265	.096
	.9250	.006	.078		.9250	.006	.078	-.120
	.9750	-.030	.012		.9750	-.030	.012	-.229

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(f) $\alpha_f = 8.6^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.470	-1.735	Upper	.0000	-1.867	-.976	-.367
	.0100	-4.163	-4.970		.0100	-4.572	-4.084	-2.741
	.0250	-4.030	-4.934		.0250	-4.976	-4.380	-2.717
	.0500	-3.392	-4.090		.0500	-4.410	-4.102	-2.458
	.1000	-1.970	-2.295		.1000	-2.452	-2.066	-1.464
	.2250	-1.127	-1.175		.2250	-1.247	-1.127	-.825
	.4500	-.614	-.608		.4500	-.693	-.548	-.446
	.7500	-.428	-.223		.7500	-.223	-.259	-.524
	.8350	-.331	.018					
	.8750	-.289	-.133					
Lower	.9250	-.247	-.096					
	.9750	-.211	-.060					
				Lower	.0500	.880	.831	.620
					.1500	.542	.494	.247
					.4000	.199	.169	.018
					.7000	.163	.175	-.133
Aileron								
Upper	.8050	-.223	-.205	Upper	.8050	-.223	-.205	-.596
	.8150	-.193	-.283		.8150	-.193	-.283	-.512
	.8350	-.151	-.048		.8350	-.151	-.048	-.657
	.8750	-.114	-.102		.8750	-.114	-.102	-.560
	.9250	-.072	-.078		.9250	-.072	-.078	-.361
	.9750	-.060	-.042		.9750	-.060	-.042	-.235
Lower	.8100	.193	.229	Lower	.8100	.193	.229	.018
	.8340	.277	.416		.8340	.277	.416	.090
	.9250	-.012	.054		.9250	-.012	.054	-.157
	.9750	-.006	.024		.9750	-.006	.024	-.205

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(g) $\alpha_f = 9.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.584	-2.108	.0000	-2.717	-1.271
	.0100	-4.349	-5.476	.0100	-5.783	-4.681
	.0250	-4.157	-5.410	.0250	-5.729	-4.886
	.0500	-3.343	-4.343	.0500	-4.657	-4.614
	.1000	-1.904	-2.428	.1000	-2.651	-2.289
	.2250	-1.078	-1.199	.2250	-1.367	-1.193
	.4500	-.693	-.596	.4500	-.651	-.560
	.7500	-.488	-.205	.7500	-.193	-.247
	.8350	-.488	-.066			
	.8750	-.512	-.193			
	.9250	-.404	-.157			
	.9750	-.373	-.096			
Lower	.0500	.970	1.000	.0500	.922	.898
	.1500	.663	.693	.1500	.614	.548
	.4000	.163	.151	.4000	.163	.211
	.7000	.066	.090	.7000	.163	.175
	.8530	-.066	-.042			
	.9250	-.440	-.066			
	.9750	-.289	-.169			
Aileron						
Upper	.8050	-.223	-.241	.8050	-.223	-.241
	.8150	-.169	-.114	.8150	-.169	-.114
	.8350	-.151	-.133	.8350	-.151	-.133
	.8750	-.120	-.114	.8750	-.120	-.114
	.9250	-.084	-.084	.9250	-.084	-.084
	.9750	-.096	-.066	.9750	-.096	-.066
Lower	.8100	.205	.247	.8100	.205	.247
	.8340	.175	.367	.8340	.175	.367
	.9250	-.054	.084	.9250	-.054	.084
	.9750	-.078	.000	.9750	-.078	.000

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(h) \quad \alpha_f = 11.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper				.0000	-2.807	-1.548
				.0100	-5.946	-5.036
				.0250	-6.006	-5.145
				.0500	-4.807	-4.861
				.1000	-2.693	-2.337
				.2250	-1.235	-1.217
				.4500	-5.96	-6.27
				.7500	-3.07	-2.83
Lower				.0500	.982	.898
				.1500	.735	.512
				.4000	.307	.187
				.7000	.127	.211
Aileron						
Upper				.8050	-.181	-.229
				.8150	-.108	-.114
				.8350	-.127	-.120
				.8750	-.114	-.133
				.9250	-.078	-.102
				.9750	-.036	-.096
Lower				.8100	-.120	.253
				.8340	-.175	.355
				.9250	.000	.036
				.9750	-.036	.000

TABLE II.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(i) \quad \alpha_f = 12.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.608	-1.675	Upper	.0000	-.946	-.223	-.199
	.0100	-4.151	-4.657		.0100	-2.645	-2.289	-2.452
	.0250	-3.964	-4.536		.0250	-2.211	-2.331	-2.452
	.0500	-3.223	-3.687		.0500	-1.428	-2.018	-2.211
	.1000	-1.771	-1.952		.1000	-.801	-.807	-1.295
	.2250	-.952	-.982		.2250	-.789	-.801	-.705
	.4500	-.608	-.627		.4500	-.789	-.843	-.416
	.7500	-.584	-.602		.7500	-.699	-.639	-.596
	.8350	-.506	-.072					
	.8750	-.494	-.398					
	.9250	-.488	-.355					
	.9750	-.422	-.301					
Lower	.0500	.916	.910	Lower	.0500	.837	.795	.602
	.1500	.578	.542		.1500	.488	.416	.217
	.4000	.193	.187		.4000	.133	.102	-.024
	.7000	-.000	.000		.7000	.030	.084	-.163
	.8530	-.133	-.078					
	.9250	-.446	-.187					
	.9750	-.349	-.319					
Aileron								
Upper	.8050	-.584	-.548	Upper	.8050	-.584	-.548	-.590
	.8150	-.554	-.494		.8150	-.554	-.494	-.452
	.8350	-.590	-.524		.8350	-.590	-.524	-.693
	.8750	-.542	-.506		.8750	-.542	-.506	-.602
	.9250	-.506	-.464		.9250	-.506	-.464	-.470
	.9750	-.512	-.422		.9750	-.512	-.422	-.349
Lower	.8100	.030	.151	Lower	.8100	.030	.151	-.018
	.8340	.145	.331		.8340	.145	.331	.127
	.9250	-.283	-.090		.9250	-.283	-.090	-.151
	.9750	-.380	-.205		.9750	-.380	-.205	-.229

TABLE III

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.5^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		.752			
	.0100		-.059			
	.0250		-.353			
	.0500		-.336			
	.1000		-.328			
	.2250		-.366			
	.4500		-.420			
	.7500		-.521			
	.8350		.059			
	.8750		-.067			
Lower	.9250		-.059			
	.9750		.046			
	.0500		-.622			
	.1500		-.424			
	.4000		-.395			
	.7000		-.155			
	.8530		-.080			
Upper	.9250		-.063			
	.9750		.008			
Wing						
Upper	.0000		.933			
	.0100		.538			
	.0250		-.008			
	.0500		-.092			
	.1000		-.416			
	.2250		-.290			
	.4500		-.340			
Lower	.7500		-.223			
	.0500		-.798			
	.1500		-.513			
	.4000		-.282			
	.7000		-.008			
Aileron						
Upper	.8050		-.345			
	.8150		-.324			
	.8350		-.143			
	.8750		-.088			
	.9250		-.013			
	.9750		.013			
	.8100					
Lower	.8340		.080			
	.9250		.080			
	.9750		.017			
	.8100		.097			
	.9750					

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(b) \quad \alpha_f = -4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:				C_p for values of $y/b/2$ of:		
		0.1621	0.2895			0.5000	0.6800	0.8920
Wing								
Upper					.0000	.756	.966	.971
					.0100	-.277	.013	.176
					.0250	-.958	-.723	-.349
					.0500	-.992	-.756	-.416
					.1000	-.937	-.685	-.525
					.2250	-.672	-.517	-.471
					.4500	-.559	-.374	-.324
					.7500	-.324	-.492	-.340
Lower					.0500	-.210	-.458	-.492
					.1500	-.269	-.374	-.366
					.4000	-.319	-.239	-.193
					.7000	-.092	.080	.034
Aileron								
Upper					.8050	-.336	-.475	-.361
					.8150	-.336	-.475	-.332
					.8350	-.231	-.399	-.466
					.8750	-.126	-.277	-.261
					.9250	-.034	-.101	-.109
					.9750	-.063	-.021	-.029
Lower					.8100	.080	.185	.210
					.8340	.080	.261	.206
					.9250	-.025	.088	.034
					.9750	.004	.055	.017

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(c) $\alpha_f = 2.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.013			
	.0100		-3.996			
	.0250		-4.315			
	.0500		-3.765			
	.1000		-2.227			
	.2250		-1.261			
	.4500		-0.702			
	.7500		-0.206			
	.8350		-0.071			
	.8750		-0.101			
	.9250		-0.080			
	.9750		-0.004			
Lower	.0500		.681			
	.1500		.391			
	.4000		.071			
	.7000		.008			
	.8530		-0.084			
	.9250		-0.067			
	.9750		-0.059			
Aileron						
Upper	.8050		-.273			
	.8150		-.172			
	.8350		-.172			
	.8750		-.147			
	.9250		-.139			
	.9750		-.101			
Lower	.8100		.122			
	.8340		.063			
	.9250		.008			
	.9750		.004			

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened

(d) $\alpha_f = 4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing					Wing	
Upper		.0000	-1.630	-.550	-.063	
		.0100	-4.786	-3.731	-2.508	
		.0250	-5.084	-4.244	-2.571	
		.0500	-4.134	-3.929	-2.307	
		.1000	-2.525	-2.113	-1.496	
		.2250	-1.239	-1.176	-.861	
		.4500	-.697	-.592	-.471	
Lower		.7500	-.118	-.336	-.542	
		.0500	.819	.798	.567	
		.1500	.563	.403	.189	
		.4000	.143	.122	-.008	
		.7000	-.013	.151	-.139	
Aileron						
Upper		.8050	-.210	-.290	-.542	
		.8150	-.176	-.286	-.529	
		.8350	-.139	-.210	-.689	
		.8750	-.071	-.143	-.605	
		.9250	-.042	-.088	-.374	
		.9750	-.034	-.063	-.235	
Lower		.8100	.151	.294	.021	
		.8340	.038	.340	.088	
		.9250	-.004	.101	-.139	
		.9750	.017	.059	-.206	

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(e) \quad \alpha_f = 5.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper				.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-.2113 -.5126 -.5235 -.4353 -.2576 -.1298 -.765 -.080	-.824 -.4118 -.4534 -.4210 -.2181 -.1168 -.618 -.315	-.315 -.2807 -.2761 -.2420 -.1496 -.861 -.454 -.546
Lower				.0500 .1500 .4000 .7000	.803 .563 .202 .067	.828 .462 .160 .168	.613 .218 .038 .143
Aileron							
Upper				.8050 .8150 .8350 .8750 .9250 .9750	-.223 -.155 -.118 -.067 -.071 -.071	-.294 -.244 -.193 -.118 -.067 -.046	-.571 -.521 -.634 -.550 -.366 -.239
Lower				.8100 .8340 .9250 .9750	.147 .025 -.059 .013	.261 .349 .092 .025	.013 .063 -.143 -.214

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(f) \quad \alpha_f = 6.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper				
	.0000	-1.739	-.718	-.298
	.0100	-4.891	-3.891	-2.693
	.0250	-5.080	-4.298	-2.664
	.0500	-4.143	-3.996	-2.328
	.1000	-2.483	-2.080	-1.471
	.2250	-1.252	-1.113	-.857
	.4500	-.685	-.567	-.433
	.7500	-.059	-.324	-.492
Lower	.0500	.870	.828	.609
	.1500	.584	.433	.223
	.4000	.193	.143	-.021
	.7000	.160	.172	-.134
Aileron				
Upper	.8050	-.151	-.252	-.508
	.8150	-.151	-.223	-.466
	.8350	-.109	-.164	-.609
	.8750	-.105	-.101	-.508
	.9250	-.038	-.038	-.340
	.9750	-.038	-.017	-.206
Lower	.8100	.122	.231	.025
	.8340	.088	.357	.088
	.9250	.008	.067	-.130
	.9750	.008	.029	-.189

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(g) $\alpha_f = 7.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.979			
	.0100		-5.420			
	.0250		-5.424			
	.0500		-4.433			
	.1000		-2.450			
	.2250		-1.311			
	.4500		-0.731			
	.7500		-0.252			
	.8350		-0.046			
	.8750		-0.092			
	.9250		-0.105			
	.9750		-0.130			
Lower	.0500		.857			
	.1500		.529			
	.4000		.269			
	.7000		.168			
	.8530		-0.050			
	.9250		-0.050			
	.9750		-0.080			
Aileron						
Upper	.8050		-0.218			
	.8150		-0.181			
	.8350		-0.105			
	.8750		-0.042			
	.9250		-0.046			
	.9750		0.008			
Lower	.8100		.202			
	.8340		.231			
	.9250		-0.076			
	.9750		-0.084			

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(h) $\alpha_f = 8.8^0$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.412	-1.878			
	.0100	-3.769	-5.361			
	.0250	-4.118	-5.256			
	.0500	-3.353	-4.218			
	.1000	-1.303	-2.399			
	.2250	-1.298	-1.197			
	.4500	-0.710	-0.601			
	.7500	-0.366	-0.282			
	.8350	-0.353	-0.126			
	.8750	-0.332	-0.109			
Lower	.9250	-0.269	-0.076			
	.9750	-0.336	-0.063			
Aileron						
Upper	.0500	.870	.979			
	.1500	.601	.605			
	.4000	.244	.256			
	.7000	.143	.076			
	.8530	-.063	.025			
	.9250	-.332	-.025			
Lower	.9750	-.206	-.088			

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(i) \alpha_f = 9.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper				
	.0000	-2.765	-1.370	-.777
	.0100	-6.210	-4.870	-3.336
	.0250	-5.870	-5.080	-3.080
	.0500	-4.622	-4.706	-2.639
	.1000	-2.689	-2.294	-1.597
	.2250	-1.399	-1.185	-.870
	.4500	-.773	-.559	-.450
Lower	.7500	-.324	-.261	-.492
Aileron				
Upper	.8050	-.218	-.185	-.542
	.8150	-.210	-.155	-.433
	.8350	-.189	-.139	-.584
	.8750	-.059	-.092	-.567
	.9250	-.084	-.080	-.357
	.9750	-.042	-.046	-.214
Lower	.8100	.004	.218	-.029
	.8340	.034	.387	.004
	.9250	-.122	.046	-.176
	.9750	-.025	.029	-.206

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(j) \quad \alpha_f = 10.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.433	-2.193	Upper	-3.046	-1.706
	.0100	-4.034	-5.689		-6.084	-5.366
	.0250	-3.979	-5.521		-6.000	-5.521
	.0500	-3.134	-4.420		-4.782	-5.008
	.1000	-1.836	-2.370		-2.714	-2.429
	.2250	-1.050	-1.151		-1.345	-1.218
	.4500	-.689	-.655		-.626	-.567
	.7500	-.567	-.445		-.185	-.235
	.8350	-.521	-.248			
	.8750	-.437	-.210			
Lower	.9250	-.437	-.210	Lower		
	.9750	-.429	-.160			
	.0500					
	.1500	.929	.979			
	.4000	.613	.697			
	.7000	.244	.366			
	.8530	.013	.118			
Aileron						
Upper	.8050			Upper		
	.8150					
	.8350					
	.8750					
	.9250					
	.9750					
Lower	.8100			Lower		
	.8340					
	.9250					
	.9750					

TABLE III.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.2 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(k) \quad \alpha_f = 11.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.744	-2.445	.0000	-3.227	-1.702
	.0100	-4.416	-5.899	.0100	-6.273	-5.374
	.0250	-4.261	-5.697	.0250	-6.063	-5.450
	.0500	-3.319	-4.517	.0500	-4.870	-5.008
	.1000	-1.798	-2.454	.1000	-2.748	-2.378
	.2250	-1.130	-1.181	.2250	-1.357	-1.185
	.4500	-0.727	-0.689	.4500	-0.622	-0.513
	.7500	-0.634	-0.525	.7500	-0.168	-0.185
	.8350	-0.542	-0.475			
	.8750	-0.521	-0.311			
	.9250	-0.571	-0.218			
	.9750	-0.567	-0.185			
Lower	.0500	.945	.920	.0500	.916	.929
	.1500	.660	.626	.1500	.588	.618
	.4000	.357	.248	.4000	.197	.239
	.7000	.113	.088	.7000	-.013	.160
	.8530	-.197	-.063			
	.9250	-.517	-.155			
	.9750	-.445	-.366			
Aileron						
Upper	.8050	-.223	-.134	.8050	-.223	-.134
	.8150	-.185	-.109	.8150	-.185	-.109
	.8350	-.185	-.113	.8350	-.185	-.113
	.8750	-.126	-.084	.8750	-.126	-.084
	.9250	-.139	-.059	.9250	-.139	-.059
	.9750	-.126	-.046	.9750	-.126	-.046
Lower	.8100	.088	.189	.8100	.088	.189
	.8340	.147	.311	.8340	.147	.311
	.9250	-.172	.055	.9250	-.172	.055
	.9750	-.134	.017	.9750	-.134	.017

TABLE IV

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(a) \alpha_f = -8.4^0$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing					Wing	
Upper		.0000	.846	.723	.705	
		.0100	.467	.751	.733	
		.0250	.039	.182	.305	
		.0500	-.182	.081	.168	
		.1000	-.442	-.175	-.165	
		.2250	-.274	-.228	-.284	
		.4500	-.267	-.253	-.298	
		.7500	-.267	-.470	-.372	
Lower		.0500	-.944	-1.274	-1.140	
		.1500	-.698	-.782	-.632	
		.4000	-.449	-.389	-.235	
		.7000	-.109	.046	.074	
Aileron						
Upper		.8050	-.393	-.467	-.316	
		.8150	-.316	-.491	-.316	
		.8350	-.151	-.432	-.407	
		.8750	-.116	-.326	-.274	
		.9250	-.053	-.116	-.077	
		.9750	-.053	.021	.007	
Lower		.8100	.070	.130	.165	
		.8340	.070	.200	.182	
		.9250	-.039	.088	.105	
		.9750	.018	.088	.077	

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(b) \quad \alpha_f = -4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000			.947	1.000	.947
	.0100			-.028	.207	.344
	.0250			-.789	-.509	-.172
	.0500			-.723	-.575	-.281
	.1000			-.821	-.575	-.442
	.2250			-.554	-.460	-.432
	.4500			-.477	-.365	-.323
	.7500			-.302	-.477	-.319
	.8350					
	.8750					
	.9250					
	.9750					
Lower	.0500			-.323	-.611	-.660
	.1500			-.372	-.495	-.421
	.4000			-.260	-.288	-.168
	.7000			.046	.074	.056
Aileron						
Upper	.8050			-.351	-.484	-.323
	.8150			-.337	-.488	-.319
	.8350			-.270	-.396	-.435
	.8750			-.165	-.274	-.256
	.9250			-.077	-.112	-.074
	.9750			-.004	.035	-.007
Lower	.8100			.070	.189	.165
	.8340			.116	.267	.211
	.9250			-.035	.074	.053
	.9750			.011	.056	.028

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(c) \quad \alpha_f = -2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						Wing
Upper	.0000		.786			
	.0100		-1.028			
	.0250		-1.691			
	.0500		-1.533			
	.1000		-1.228			
	.2250		-0.782			
	.4500		-0.596			
	.7500		-0.516			
	.8350		-0.042			
	.8750		-0.070			
	.9250		-0.111			
	.9750		-0.004			
Lower	.0500		.021			
	.1500		-0.077			
	.4000		-0.256			
	.7000		-0.025			
	.8530		-0.049			
	.9250		-0.046			
	.9750		.018			
Aileron						
Upper	.8050		-0.372			
	.8150		-0.323			
	.8350		-0.193			
	.8750		-0.098			
	.9250		-0.021			
	.9750		-0.014			
Lower	.8100		.119			
	.8340		.119			
	.9250		-0.077			
	.9750		.018			

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(d) $\alpha_f = -0.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper	.0000		.309				
	.0100		-1.933				
	.0250		-2.540				
	.0500		-2.193				
	.1000		-1.632				
	.2250		-1.011				
	.4500		-.677				
	.7500		-.526				
	.8350		-.179				
	.8750		-.154				
	.9250		-.039				
	.9750		.011				
Lower	.0500		.337				
	.1500		.021				
	.4000		-.130				
	.7000		-.130				
	.8530		-.095				
	.9250		-.063				
Aileron	Wing						
	Upper	.0000	.491	.898	.940		
		.0100	-1.449	-.828	-.404		
		.0250	-2.196	-1.646	-.867		
		.0500	-1.986	-1.568	-.895		
		.1000	-1.565	-1.056	-.818		
		.2250	-.916	-.681	-.596		
		.4500	-.677	-.456	-.372		
		.7500	-.361	-.435	-.365		
	Lower	.0500	.246	.102	-.119		
		.1500	-.049	-.098	-.175		
		.4000	-.123	-.102	-.130		
		.7000	.014	.140	-.000		
Aileron	Aileron						
	Upper	.8050	-.375	-.407	-.323		
		.8150	-.319	-.414	-.319		
		.8350	-.246	-.344	-.463		
		.8750	-.147	-.214	-.323		
		.9250	-.077	-.091	-.154		
		.9750	-.035	-.004	-.067		
	Lower	.8100	.105	.232	.200		
		.8340	.172	.330	.193		
		.9250	-.046	.077	-.011		
		.9750	-.028	.063	-.042		

TABLE IV..- Continued
CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(e) $\alpha_f = 0.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper						
	.0000			.554		.754
	.0100	-2.407		-1.726		-1.021
	.0250	-2.961		-2.586		-1.439
	.0500	-2.544		-2.305		-1.400
	.1000	-1.747		-1.456		-1.056
	.2250	-.968		-.849		-.758
	.4500	-.593		-.551		-.407
	.7500	-.270		-.418		-.446
Lower						
	.0500		.442		.323	.116
	.1500		.168		.088	-.091
	.4000		.095		-.112	-.123
	.7000		.004		.102	-.067
Aileron						
Upper						
	.8050		-.179		-.418	-.404
	.8150		-.232		-.393	-.418
	.8350		-.204		-.305	-.646
	.8750		-.133		-.207	-.411
	.9250		-.074		-.140	-.228
Lower						
	.9750		-.042		-.004	-.147
Lower						
	.8100		-.049		.214	.102
	.8340		-.109		.319	.088
	.9250		-.039		.032	-.105
	.9750		-.035		.000	-.154

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened

(f) $\alpha_f = 1.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.530			
	.0100		-3.502			
	.0250		-3.751			
	.0500		-3.358			
	.1000		-2.021			
	.2250		-1.147			
	.4500		-.712			
	.7500		-.418			
	.8350		-.144			
	.8750		-.088			
	.9250		-.007			
	.9750		.070			
Lower	.0500		.768			
	.1500		.344			
	.4000		.091			
	.7000		.053			
	.8530		.025			
	.9250		.021			
	.9750		-.053			
Aileron						
Upper	.8050		-.277			
	.8150		-.232			
	.8350		-.168			
	.8750		-.109			
	.9250		-.053			
	.9750		.011			
Lower	.8100		.179			
	.8340		.200			
	.9250		-.046			
	.9750		.007			

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(g) \quad \alpha_f = 2.3^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing					Wing	
Upper				.0000	-1.000	-.119
				.0100	-4.379	-3.067
				.0250	-4.372	-3.695
				.0500	-3.628	-3.372
				.1000	-2.288	-1.860
				.2250	-1.214	-1.021
				.4500	-.747	-.575
				.7500	-.225	-.358
Lower				.0500	.681	.681
				.1500	.288	.302
				.4000	.000	.046
				.7000	-.007	.123
Aileron						
Upper				.8050	-.274	-.302
				.8150	-.175	-.298
				.8350	-.126	-.232
				.8750	-.077	-.161
				.9250	-.028	-.067
				.9750	-.007	-.035
Lower				.8100	.056	.214
				.8340	.154	.368
				.9250	-.011	.060
				.9750	-.056	.035

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(h) $\alpha_f = 2.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.260			
	.0100		-4.551			
	.0250		-4.558			
	.0500		-3.839			
	.1000		-2.358			
	.2250		-1.316			
	.4500		-.719			
	.7500		-.333			
	.8350		-.088			
	.8750		-.084			
Lower	.9250		-.067			
	.9750		-.014			
	.0500			.754		
	.1500			.347		
	.4000			.123		
	.7000			.035		
Aileron						
Upper	.8050		-.200		.305	-.421
	.8150		-.193		.277	-.382
	.8350		-.147		.218	-.575
	.8750		-.112		.172	-.498
	.9250		-.049		.074	-.298
	.9750		-.046		.032	-.168
Lower	.8100		.098		.193	.063
	.8340		.095		.393	.067
	.9250		-.053		.046	-.154
	.9750		-.007		.039	-.161

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(i) \quad \alpha_f = 3.3^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.435			
	.0100		-4.786			
	.0250		-4.723			
	.0500		-3.846			
	.1000		-2.323			
	.2250		-1.256			
	.4500		-.660			
	.7500		-.232			
	.8350		-.035			
	.8750		.000			
Lower	.9250		.063			
	.9750		.105			
	.0500		.909			
	.1500		.509			
	.4000		.232			
	.7000		.130			
Aileron	.8530		.077			
	.9250		.077			
	.9750		-.032			
	.8050		-.172			
	.8150		-.196			
	.8350		-.095			
	.8750		-.060			
Upper	.9250		-.014			
	.9750		-.004			
	.8100		.175			
	.8340		.126			
	.9250		-.035			
Lower	.9750		-.000			
	.8100		.214			
	.8340		.375			
	.9250		.049			
Aileron	.9750		.028			
	.8100		.007			
Aileron	.8340		.063			
	.9250		-.200			
Aileron	.9750		-.189			

TABLE IV.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.1 \text{ lb/sq ft}$; belly plate off;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(j) $\alpha_f = 3.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.828			
	.0100		-5.337			
	.0250		-5.158			
	.0500		-4.140			
	.1000		-2.516			
	.2250		-1.274			
	.4500		-.621			
	.7500		-.189			
	.8350		.028			
	.8750		.028			
	.9250		.028			
	.9750		.081			
Lower	.0500		.944			
	.1500		.565			
	.4000		.326			
	.7000		.077			
	.8530		.060			
	.9250		-.007			
	.9750		-.007			
Aileron						
Upper	.8050		-.140			
	.8150		-.137			
	.8350		-.095			
	.8750		-.039			
	.9250		.021			
	.9750		.042			
Lower	.8100		.144			
	.8340		.084			
	.9250		.039			
	.9750		.032			

TABLE V

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(a) \quad \alpha_f = -6.6^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		
		0.1621	0.2895	
Wing				
Upper				
	.0000	.919	.958	.922
	.0100	.219	.442	.491
	.0250	-.435	-.286	-.004
	.0500	-.466	-.304	-.110
	.1000	-.678	-.406	-.343
	.2250	-.498	-.350	-.371
	.4500	-.470	-.304	-.300
	.7500	-.290	-.382	-.300
Lower	.0500	-.611	-.837	-.799
	.1500	-.512	-.597	-.498
	.4000	-.389	-.311	-.208
	.7000	-.064	.011	.011
Aileron				
Upper	.8050	-.375	-.389	-.304
	.8150	-.237	-.396	-.261
	.8350	-.134	-.350	-.371
	.8750	-.071	-.261	-.219
	.9250	-.000	-.088	-.042
	.9750	.046	.042	.049
Lower	.8100	-.004	.124	.138
	.8340	.117	.198	.127
	.9250	.018	.074	.057
	.9750	.067	.081	.057

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:				C_p for values of $y/\frac{b}{2}$ of:																															
		0.1621	0.2895																																		
Wing																																					
Upper				<table> <tr><td>.0000</td><td>.806</td><td>1.014</td><td>.993</td></tr> <tr><td>.0100</td><td>-.127</td><td>.088</td><td>.219</td></tr> <tr><td>.0250</td><td>-.972</td><td>-.650</td><td>-.311</td></tr> <tr><td>.0500</td><td>-.979</td><td>-.707</td><td>-.382</td></tr> <tr><td>.1000</td><td>-.926</td><td>-.629</td><td>-.481</td></tr> <tr><td>.2250</td><td>-.640</td><td>-.477</td><td>-.445</td></tr> <tr><td>.4500</td><td>-.534</td><td>-.371</td><td>-.332</td></tr> <tr><td>.7500</td><td>-.512</td><td>-.396</td><td>-.290</td></tr> </table>	.0000	.806	1.014	.993	.0100	-.127	.088	.219	.0250	-.972	-.650	-.311	.0500	-.979	-.707	-.382	.1000	-.926	-.629	-.481	.2250	-.640	-.477	-.445	.4500	-.534	-.371	-.332	.7500	-.512	-.396	-.290	
.0000	.806	1.014	.993																																		
.0100	-.127	.088	.219																																		
.0250	-.972	-.650	-.311																																		
.0500	-.979	-.707	-.382																																		
.1000	-.926	-.629	-.481																																		
.2250	-.640	-.477	-.445																																		
.4500	-.534	-.371	-.332																																		
.7500	-.512	-.396	-.290																																		
Aileron																																					
Upper				<table> <tr><td>.8050</td><td>-.325</td><td>-.389</td><td>-.318</td></tr> <tr><td>.8150</td><td>-.304</td><td>-.410</td><td>-.290</td></tr> <tr><td>.8350</td><td>-.208</td><td>-.346</td><td>-.375</td></tr> <tr><td>.8750</td><td>-.163</td><td>-.290</td><td>-.237</td></tr> <tr><td>.9250</td><td>-.042</td><td>-.085</td><td>-.071</td></tr> <tr><td>.9750</td><td>-.060</td><td>-.039</td><td>.018</td></tr> </table>	.8050	-.325	-.389	-.318	.8150	-.304	-.410	-.290	.8350	-.208	-.346	-.375	.8750	-.163	-.290	-.237	.9250	-.042	-.085	-.071	.9750	-.060	-.039	.018									
.8050	-.325	-.389	-.318																																		
.8150	-.304	-.410	-.290																																		
.8350	-.208	-.346	-.375																																		
.8750	-.163	-.290	-.237																																		
.9250	-.042	-.085	-.071																																		
.9750	-.060	-.039	.018																																		
Lower				<table> <tr><td>.8100</td><td>.078</td><td>.138</td><td>.163</td></tr> <tr><td>.8340</td><td>.067</td><td>.226</td><td>.110</td></tr> <tr><td>.9250</td><td>-.046</td><td>.081</td><td>.021</td></tr> <tr><td>.9750</td><td>-.018</td><td>.085</td><td>.035</td></tr> </table>	.8100	.078	.138	.163	.8340	.067	.226	.110	.9250	-.046	.081	.021	.9750	-.018	.085	.035																	
.8100	.078	.138	.163																																		
.8340	.067	.226	.110																																		
.9250	-.046	.081	.021																																		
.9750	-.018	.085	.035																																		

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $a_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(c) \quad \alpha_f = -2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		
		0.1621	0.2895	
Wing				
Upper				
	.0000	.830	.951	.965
	.0100	-.883	-.477	-.120
	.0250	-1.527	-1.269	-.625
	.0500	-1.403	-1.219	-.682
	.1000	-1.233	-.901	-.700
	.2250	-.700	-.611	-.534
	.4500	-.541	-.417	-.339
Lower	.7500	-.233	-.385	-.322
	.0500	.102	-.127	-.290
	.1500	-.148	-.233	-.279
	.4000	-.120	-.180	-.134
	.7000	-.011	.081	-.021
Aileron				
Upper	.8050	-.226	-.360	-.269
	.8150	-.198	-.357	-.251
	.8350	-.092	-.322	-.438
	.8750	-.049	-.212	-.286
	.9250	-.018	-.053	-.127
	.9750	.053	.039	-.025
Lower	.8100	.099	.170	.155
	.8340	.155	.279	.092
	.9250	-.011	.071	.004
	.9750	.053	.071	.011

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(d) $\alpha_f = -0.90$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		
		0.1621	0.2895	
Wing				
Upper				
	.0000	.544	.862	.929
	.0100	-1.512	-.887	-.495
	.0250	-2.145	-1.710	-.968
	.0500	-1.873	-1.618	-.996
	.1000	-1.537	-1.124	-.883
	.2250	-.894	-.710	-.622
	.4500	-.629	-.459	-.371
Lower	.7500	-.329	-.403	-.325
	.0500	.216	.099	-.078
	.1500	.000	-.113	-.184
	.4000	-.134	-.155	-.138
Aileron	.7000	-.014	.067	-.039
	.8050	-.283	-.375	-.336
	.8150	-.251	-.367	-.290
	.8350	-.159	-.329	-.445
	.8750	-.148	-.194	-.336
	.9250	-.071	-.064	-.155
	.9750	-.046	.021	-.049
	.8100	.092	.170	.134
Lower	.8340	.099	.276	.064
	.9250	-.021	.060	-.064
	.9750	-.049	.057	-.053

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(e) $\alpha_f = 1.0^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.127		.569	.724
	.0100		-2.982		-1.739	-1.057
	.0250		-3.495		-2.523	-1.445
	.0500		-3.025		-2.304	-1.392
	.1000		-1.961		-1.424	-1.053
	.2250		-1.067		-.848	-.686
	.4500		-.678		-.509	-.389
	.7500		-.459		-.357	-.329
	.8350		-.127			
	.8750		-.074			
	.9250		-.078			
	.9750		.035			
Lower	.0500		.530		.378	.163
	.1500		.191		.085	-.046
	.4000		-.004		-.060	-.088
	.7000		-.018		.078	-.071
	.8530		-.042			
	.9250		-.067			
	.9750		-.049			
Aileron						
Upper	.8050		-.230		-.329	-.300
	.8150		-.216		-.325	-.300
	.8350		-.152		-.276	-.456
	.8750		-.067		-.163	-.353
	.9250		-.007		-.057	-.216
	.9750		.071		.042	-.092
Lower	.8100		.099		.180	.120
	.8340		.106		.307	.078
	.9250		-.042		.074	-.110
	.9750		-.007		.057	-.092

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(f) $\alpha_f = 1.4^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-.417	-.180	.0000	-.131	.463
	.0100	-3.152	-3.272	.0100	-2.813	-1.940
	.0250	-3.488	-3.654	.0250	-3.265	-2.703
	.0500	-2.912	-3.230	.0500	-2.792	-2.456
	.1000	-1.830	-1.954	.1000	-1.887	-1.509
	.2250	-1.081	-1.088	.2250	-1.032	-.866
	.4500	-.668	-.657	.4500	-.686	-.505
	.7500	-.198	-.459	.7500	-.219	-.339
	.8350	-.085	-.106			-.346
	.8750	-.071	-.035			
	.9250	-.042	.007			
	.9750	-.067	.071			
Lower	.0500	.664	.686	.0500	.611	.459
	.1500	.318	.272	.1500	.194	.131
	.4000	.049	.028	.4000	-.011	-.025
	.7000	.025	.032	.7000	.039	.092
	.8530	-.035	.028			
	.9250	-.184	-.007			
	.9750	.000	-.035			
Aileron						
Upper	.8050	-.244	-.293	.8050	-.244	-.307
	.8150	-.205	-.279	.8150	-.205	-.297
	.8350	-.173	-.233	.8350	-.173	-.488
	.8750	-.095	-.141	.8750	-.095	-.396
	.9250	-.004	-.049	.9250	-.004	-.230
	.9750	.000	.046	.9750	.000	-.106
Lower	.8100	.102	.201	.8100	.102	.110
	.8340	.113	.339	.8340	.113	.060
	.9250	-.060	.060	.9250	-.060	-.131
	.9750	.007	.064	.9750	.007	-.110

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 1.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.417			
	.0100		-3.382			
	.0250		-3.781			
	.0500		-3.279			
	.1000		-2.110			
	.2250		-1.177			
	.4500		-.788			
	.7500		-.495			
	.8350		-.120			
	.8750		-.124			
	.9250		-.074			
	.9750		.004			
Lower	.0500		.590			
	.1500		.226			
	.4000		.081			
	.7000		-.011			
	.8530		-.021			
	.9250		-.106			
	.9750		-.106			
Aileron						
Upper	.8050		-.261			
	.8150		-.201			
	.8350		-.131			
	.8750		-.117			
	.9250		-.025			
	.9750		-.053			
Lower	.8100		.067			
	.8340		.106			
	.9250		-.078			
	.9750		-.000			

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(h) $\alpha_f = 2.4^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-.777	-.601	.544	.177	.452
	.0100	-3.615	-3.876	-3.431	-2.608	-1.615
	.0250	-3.876	-4.106	-3.820	-3.307	-1.908
	.0500	-3.191	-3.643	-3.254	-3.004	-1.777
	.1000	-1.972	-2.113	-2.071	-1.710	-1.251
	.2250	-1.240	-1.141	-1.092	-.965	-.792
	.4500	-.647	-.664	-.700	-.530	-.410
	.7500	-.265	-.392	-.180	-.297	-.364
	.8350	-.071	-.095			
	.8750	-.049	-.039			
	.9250	-.014	.018			
	.9750	.035	.049			
Lower	.0500	.774	.802	.731	.587	.360
	.1500	.357	.343	.329	.223	.057
	.4000	.092	.124	.067	.014	-.064
	.7000	.064	.060	.078	.088	-.117
	.8530	-.046	.049			
	.9250	-.134	-.011			
	.9750	-.039	-.042			
Aileron						
Upper	.8050	-.173	-.261	-.247	-.332	-.272
	.8150	-.166	-.117	-.198	-.541	
	.8350	-.117	-.057	-.120	-.449	
	.8750	-.000	-.025	-.035	-.318	
	.9250	.042			-.166	
	.9750					
Lower	.8100	.131	.205	.367	.074	
	.8340	.092	.042	.042	.042	
	.9250	-.035	.035		-.170	
	.9750	.032			-.141	

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(i) $\alpha_f = 2.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000		-.636	
	.0100		-3.943	
	.0250		-4.339	
	.0500		-3.753	
	.1000		-2.152	
	.2250		-1.187	
	.4500		-.693	
	.7500		-.353	
	.8350		-.039	
	.8750		.014	
	.9250		.007	
	.9750		.074	
Lower	.0500		.827	
	.1500		.392	
	.4000		.170	
	.7000		.092	
	.8530		.032	
	.9250		.057	
	.9750		.011	
Aileron				
Upper	.8050		-.198	-.251
	.8150		-.170	-.233
	.8350		-.113	-.191
	.8750		-.064	-.120
	.9250		-.018	-.025
	.9750		.021	.039
Lower	.8100		.120	.191
	.8340		.102	.378
	.9250		-.021	.039
	.9750		-.000	.049

TABLE V.—Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

$$(j) \quad \alpha_f = 3.30$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing						Wing	
Upper				.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.000 -.951 -4.074 -4.166 -3.728 -2.290	.011 -.011 -.000 -.184 -3.307 -3.908 -3.569 -1.940	.053 .046 .014 .219 -2.117 -2.297 -2.120 -1.413
Lower				.0500 .1500 .4000 .7000	-1.155 -.710 -.240 .763	-1.060 -.551 -.276 .721	-.855 -.431 -.353 .473
Aileron							
Upper				.8050 .8150 .8350 .8750 .9250 .9750	.378 .134 .088 -.173 -.208 -.159	.311 .057 .085 -.216 -.201 -.166	.148 -.071 -.145 -.336 -.300 -.537
Lower				.8100 .8340 .9250 .9750	-.074 .007 -.039 .018	-.106 -.018 .046 .205	-.495 -.346 -.180 .025

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(k) $\alpha_f = 4.2^0$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper				
	.0000		-.689	-.194
	.0100		-4.152	-2.731
	.0250		-4.608	-2.735
	.0500		-4.265	-2.435
	.1000		-2.180	-1.537
	.2250		-1.155	-.890
	.4500		-.580	-.456
Lower	.7500		-.240	-.385
	.0500		.806	.622
	.1500		.378	.201
Lower	.4000		.102	-.032
	.7000		.102	-.205
Aileron				
Upper	.8050		-.194	-.357
	.8150		-.180	-.371
	.8350		-.117	-.519
	.8750		-.067	-.541
	.9250		-.025	-.424
	.9750		.057	-.269
Lower	.8100		.159	-.039
	.8340		.346	-.028
	.9250		.021	-.205
	.9750		.039	-.028

TABLE V.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 1-3-6-8; $q_\infty = 17.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(1) $\alpha_f = 4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-2.011	-1.922	Upper	.0000	-2.102	-.700	-.187
	.0100	-5.187	-5.827		.0100	-5.431	-4.180	-2.693
	.0250	-5.099	-5.572		.0250	-5.325	-4.618	-2.686
	.0500	-4.057	-4.435		.0500	-4.403	-4.265	-2.403
	.1000	-2.495	-2.608		.1000	-2.608	-2.170	-1.534
	.2250	-1.396	-1.385		.2250	-1.322	-1.127	-.890
	.4500	-.749	-.721		.4500	-.724	-.544	-.449
	.7500	-.230	-.244		.7500	-.166	-.184	-.385
	.8350	-.032	-.110					
	.8750	-.067	-.067					
	.9250	-.046	-.078					
	.9750	-.042	-.032					
Lower	.0500	.788	.919	Lower	.0500	.795	.837	.604
	.1500	.541	.523		.1500	.516	.459	.244
	.4000	.261	.201		.4000	.138	.131	-.021
	.7000	.166	.110		.7000	.071	.110	-.177
	.8530	.011	.057					
	.9250	-.184	-.039					
	.9750	-.110	-.102					
Aileron								
Upper	.8050	-.124	-.148	-.343				
	.8150	-.120	-.127	-.339				
	.8350	-.110	-.113	-.452				
	.8750	-.074	-.071	-.502				
	.9250	-.025	-.007	-.378				
	.9750	.018	.039	-.219				
Lower	.8100	.085	.191	-.014				
	.8340	.088	.367	.014				
	.9250	-.053	.049	-.201				
	.9750	.000	.057	-.170				

TABLE VI

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(a) $\alpha_f = -8.5^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.974	.939	Upper	.965	1.035
	.0100	-.122	-.052		.087	.226
	.0250	-.748	-.661		-.548	-.391
	.0500	-.748	-.696		-.539	-.461
	.1000	-.696	-.687		-.635	-.487
	.2250	-.565	-.557		-.513	-.374
	.4500	-.348	-.400		-.487	-.278
	.7500	-.296	-.539		-.261	-.243
	.8350	-.087	-.113			
	.8750	-.070	-.087			
	.9250	-.139	-.052			
	.9750	-.096	-.000			
Lower	.0500	-.487	-.348	Lower	-.452	-.539
	.1500	-.365	-.435		-.391	-.417
	.4000	-.322	-.313		-.357	-.270
	.7000	-.174	-.122		-.104	-.052
	.8530	-.104	-.087			
	.9250	-.096	-.043			
	.9750	-.096	-.009			
Aileron						
Upper	.8050	-.243	-.139	Upper	-.226	-.139
	.8150	-.139	-.191		-.139	-.104
	.8350	-.096	-.165		-.191	-.191
	.8750	-.000	-.078		-.165	-.157
	.9250	.035	.035		-.078	-.070
	.9750					.000
Lower	.8100	-.009	.122	Lower	.035	-.035
	.8340	.035	.122		-.009	-.009
	.9250	-.052	.026		.035	.035
	.9750	.052	.113		.017	.017

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.70$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.843	.800	.835	.922	.913
	.0100	-.878	-.913	-.765	-.513	-.278
	.0250	-1.513	-1.583	-1.443	-1.226	-.791
	.0500	-1.400	-1.461	-1.322	-1.209	-.774
	.1000	-1.061	-1.139	-1.096	-.843	-.643
	.2250	-.713	-.713	-.678	-.583	-.487
	.4500	-.383	-.443	-.478	-.383	-.270
	.7500	-.261	-.487	-.252	-.226	-.078
	.8350	-.052	-.043			
	.8750	-.009	-.017			
	.9250	-.009	.009			
	.9750	.052	.148			
Lower	.0500	.035	.148	.070	-.061	-.191
	.1500	-.061	-.096	-.148	-.183	-.270
	.4000	-.165	-.200	-.226	-.157	-.183
	.7000	-.035	-.052	-.026	-.009	-.096
	.8530	-.052	.009			
	.9250	-.052	-.017			
	.9750	.017	.043			
Aileron						
Upper	.8050	-.226	-.157	-.157	-.035	
	.8150	-.174	-.139	-.139	-.017	
	.8350	-.104	-.165	-.165	-.096	
	.8750	-.017	-.139	-.139	-.148	
	.9250	.035	-.017	-.017	-.052	
	.9750	.035	.113	.113	-.052	
Lower	.8100	-.035	.200	.200	-.061	
	.8340	.122	.183	.183	-.078	
	.9250	-.043	.043	.043	-.078	
	.9750	.043	.078	.078	-.017	

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(c) $\alpha_f = -0.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.313	.278	Upper	.0000	.278
	.0100	-1.965	-2.087		.0100	-1.896
	.0250	-2.400	-2.661		.0250	-2.374
	.0500	-2.122	-2.374		.0500	-2.139
	.1000	-1.417	-1.548		.1000	-1.591
	.2250	-0.922	-0.939		.2250	-0.922
	.4500	-0.487	-0.539		.4500	-0.583
	.7500	-0.278	-0.435		.7500	-0.200
	.8350	-0.070	-0.087			
	.8750	-0.052	-0.035			
	.9250	-0.035	0.026			
	.9750	0.000	0.070			
Lower	.0500	.461	.530	Lower	.0500	.452
	.1500	.157	.122		.1500	.122
	.4000	-0.043	-0.070		.4000	-0.087
	.7000	.000	.009		.7000	.017
	.8530	-0.052	-0.017			
	.9250	-0.130	0.000			
	.9750	-0.035	0.009			
Aileron						
Upper	.8050	-0.191	-0.157	Upper	.8050	-0.191
	.8150	-0.157	-0.130		.8150	-0.157
	.8350	-0.113	-0.157		.8350	-0.113
	.8750	-0.043	-0.122		.8750	-0.043
	.9250	.000	-0.026		.9250	.000
	.9750	.035	.070		.9750	.035
Lower	.8100	.087	.139	Lower	.8100	.087
	.8340	.191	.209		.8340	.191
	.9250	.000	.009		.9250	.000
	.9750	.035	.035		.9750	.035

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(d) \quad \alpha_f = 2.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.574			
	.0100		-3.713			
	.0250		-3.904			
	.0500		-3.496			
	.1000		-2.043			
	.2250		-1.252			
	.4500		-.652			
	.7500		-.304			
	.8350		-.096			
	.8750		-.122			
	.9250		-.043			
	.9750		-.043			
Lower	.0500		.600			
	.1500		.148			
	.4000		-.043			
	.7000		-.070			
	.8530		-.009			
	.9250		-.009			
	.9750		-.061			
Aileron						
Upper	.8050		-.252			
	.8150		-.270			
	.8350		-.157			
	.8750		-.113			
	.9250		-.078			
	.9750		-.078			
Lower	.8100		.061			
	.8340		.130			
	.9250		-.043			
	.9750		-.043			
				.035		
				.165		
				.009		
				.009		
				-.104		
				-.087		
				-.165		
				-.096		

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(e) \alpha_f = 6.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing						Wing		
Upper	.0000	-1.983	-1.722	Upper	.0000	-1.835	-.809	-.209
	.0100	-4.965	-5.322		.0100	-4.852	-3.974	-2.643
	.0250	-4.713	-5.287		.0250	-4.965	-4.243	-2.670
	.0500	-3.957	-4.557		.0500	-4.174	-3.983	-2.426
	.1000	-2.296	-2.443		.1000	-2.443	-2.009	-1.400
	.2250	-1.304	-1.304		.2250	-1.296	-1.096	-0.783
	.4500	-.574	-.643		.4500	-.661	-.487	-.391
	.7500	-.209	-.191		.7500	-.174	-.139	-.357
	.8350	-.087	-.096					
	.8750	-.070	-.070					
	.9250	-.035	-.043					
	.9750	.000	-.017					
Lower	.0500	.904	.930	Lower	.0500	.870	.870	.661
	.1500	.600	.557		.1500	.496	.478	.261
	.4000	.174	.209		.4000	.165	.165	.009
	.7000	.096	.113		.7000	.113	.113	-.157
	.8530	.078	.017					
	.9250	-.113	-.026					
	.9750	-.035	-.096					
Aileron								
Upper	.8050	-.165	-.096	Upper	.8050	-.165	-.070	.070
	.8150	-.139	-.078		.8150	-.104	-.078	-.209
	.8350	-.061	-.017		.8350	-.061	-.017	-.365
	.8750	-.043	.026		.8750	-.035	.026	-.270
	.9250	-.035	.035		.9250	-.035	.035	-.183
	.9750							
Lower	.8100	.157	.183	Lower	.8100	.278	.243	-.217
	.8340	-.009	.078		.8340	-.017	.078	.026
	.9250							-.148
	.9750							-.104

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(f) $\alpha_f = 8.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.383	-1.496	Upper	.0000	-2.043	-1.183	-.539
	.0100	-3.983	-4.904		.0100	-5.061	-4.443	-3.122
	.0250	-3.965	-4.922		.0250	-5.200	-4.661	-3.009
	.0500	-3.417	-4.252		.0500	-4.261	-4.357	-2.704
	.1000	-1.896	-2.339		.1000	-2.496	-2.139	-1.539
	.2250	-1.043	-1.217		.2250	-1.261	-1.148	-.843
	.4500	-.600	-.661		.4500	-.678	-.539	-.426
	.7500	-.496	-.313		.7500	-.226	-.191	-.530
	.8350	-.417	-.200	Wing				
	.8750	-.383	-.157	Aileron				
	.9250	-.339	-.130	Upper	.0500	.843	.878	.652
	.9750	-.270	-.070		.1500	.522	.496	.226
Lower	.0500	.826	.887		.4000	.148	.139	-.017
	.1500	.504	.504		.7000	.104	.078	-.235
	.4000	.104	.157	Aileron				
	.7000	.009	.070	Upper	.8050	-.183	-.148	-.357
	.8530	.070	-.052		.8150	-.174	-.139	.165
	.9250	-.322	-.104		.8350	-.122	-.130	-.365
	.9750	-.278	-.139		.8750	-.096	-.061	-.496
					.9250	-.026	-.035	-.383
					.9750	-.070	.009	-.243
Lower	.8100		.122	Lower	.8100	.122	.130	-.330
	.8340		.183		.8340	.235	.017	.017
	.9250		-.052		.9250	.017	-.226	-.226
	.9750		-.035		.9750	.026	-.191	-.191

TABLE VI.- Continued
 CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON
 $[\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 9.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.530	-1.765	Upper	.0000	-2.313	-1.217	-.487
	.0100	-4.157	-5.235		.0100	-5.322	-4.496	-3.052
	.0250	-3.991	-5.139		.0250	-5.270	-4.678	-2.930
	.0500	-3.357	-4.296		.0500	-4.417	-4.374	-2.704
	.1000	-1.835	-2.313		.1000	-2.487	-2.148	-1.522
	.2250	-.957	-1.148		.2250	-1.252	-1.122	-.843
	.4500	-.591	-.557		.4500	-.626	-.539	-.400
	.7500	-.513	-.296		.7500	-.174	-.191	-.452
	.8350	-.426	-.183					
	.8750	-.374	-.139					
	.9250	-.374	-.122					
	.9750	-.348	-.104					
Lower	.0500	.930	.957	Lower	.0500	.922	.913	.661
	.1500	.617	.591		.1500	.583	.548	.304
	.4000	.191	.226		.4000	.226	.165	.035
	.7000	.070	.104		.7000	.157	.104	-.183
	.8530	.052	-.026					
	.9250	-.365	-.070					
	.9750	-.313	-.139					
Aileron								
Upper	.8050	-.148	-.104	Upper	.8050	-.148	-.104	-.322
	.8150	-.139	-.104		.8150	-.122	-.096	.130
	.8350	-.070	-.043		.8350	-.052	-.026	-.287
	.8750	-.035	-.035		.8750	-.017	-.017	-.443
	.9250				.9250			-.348
	.9750				.9750			-.252
Lower	.8100	.148	.148	Lower	.8100	.235	.226	-.278
	.8340	.017	.017		.8340	-.017	-.017	.035
	.9250				.9250			-.165
	.9750				.9750			-.139

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(h) \quad \alpha_f = 10.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.687	-2.017	.0000	-2.722	-1.565
	.0100	-4.270	-5.461	.0100	-5.748	-5.000
	.0250	-4.087	-5.330	.0250	-5.704	-5.026
	.0500	-3.357	-4.383	.0500	-4.617	-4.809
	.1000	-1.826	-2.348	.1000	-2.574	-2.278
	.2250	-1.009	-1.148	.2250	-1.270	-1.157
	.4500	-.626	-.591	.4500	-.609	-.530
	.7500	-.522	-.339	.7500	-.148	-.183
	.8350	-.452	-.226			
	.8750	-.435	-.191			
Lower	.9250	-.426	-.165			
	.9750	-.417	-.122			
	.0500	.913	.948	.0500	.904	.922
	.1500	.609	.591	.1500	.600	.557
	.4000	.217	.261	.4000	.252	.200
	.7000	.052	.078	.7000	.148	.113
Aileron						
Upper	.8050	-.157	-.165	.8050	-.157	-.165
	.8150	-.139	-.139	.8150	-.139	-.217
	.8350	-.122	-.130	.8350	-.122	-.339
	.8750	-.096	-.078	.8750	-.096	-.513
	.9250	-.070	-.052	.9250	-.070	-.409
	.9750	-.078	.009	.9750	-.078	-.243
Lower	.8100	.157	.157	.8100	.157	.365
	.8340	.243	.235	.8340	.243	.026
	.9250	-.052	.026	.9250	-.052	-.235
	.9750	-.043	.026	.9750	-.043	-.217

TABLE VI.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(i) $\alpha_f = 11.8^0$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.739	-2.278	Upper	.0000	-3.043	-1.965	-1.122
	.0100	-4.365	-5.730		.0100	-6.139	-5.322	-3.774
	.0250	-4.070	-5.548		.0250	-5.922	-5.313	-3.443
	.0500	-3.574	-4.539		.0500	-4.817	-5.087	-2.965
	.1000	-2.043	-2.287		.1000	-2.609	-2.330	-1.704
	.2250	-1.278	-1.183		.2250	-1.348	-1.165	-.904
	.4500	-.904	-.713		.4500	-.617	-.557	-.470
	.7500	-.470	-.435		.7500	-.226	-.200	-.609
	.8350	-.522	-.383					
	.8750	-.643	-.296					
	.9250	-.530	-.270					
	.9750	-.530	-.096					
Lower	.0500	.617	.800	Lower	.0500	.870	.896	.722
	.1500	.470	.626		.1500	.504	.565	.348
	.4000	.087	.183		.4000	.226	.209	-.009
	.7000	-.122	.122		.7000	.113	.078	-.243
	.8530	-.026	-.096					
	.9250	-.504	-.104					
	.9750	-.383	-.104					
Aileron								
Upper	.8050	-.130	-.174	Upper	.8050	-.130	-.174	-.374
	.8150	-.139	-.122		.8150	-.139	-.122	.157
	.8350	-.139	-.113		.8350	-.139	-.113	-.383
	.8750	-.087	-.078		.8750	-.087	-.078	-.557
	.9250	-.104	-.052		.9250	-.104	-.052	-.443
	.9750	-.157	-.035		.9750	-.157	-.035	-.278
Lower	.8100	.035	.104	Lower	.8100	.035	.104	-.383
	.8340	.235	.217		.8340	.235	.217	-.009
	.9250	-.052	-.017		.9250	-.052	-.017	-.226
	.9750	-.017	-.017		.9750	-.017	-.017	-.226

TABLE VII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(a) \alpha_f = -8.5^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000	1.018	.982	
	.0100	-.095	.107	
	.0250	-.710	-.580	
	.0500	-.698	-.592	
	.1000	-.663	-.669	
	.2250	-.538	-.497	
	.4500	-.325	-.385	
	.7500	-.272	-.538	
	.8350	-.047	-.041	
	.8750	-.071	-.036	
	.9250	-.124	-.018	
	.9750	-.083	.024	
Lower	.0500	-.485	-.391	
	.1500	-.355	-.438	
	.4000	-.284	-.308	
	.7000	-.160	-.101	
	.8530	-.089	-.053	
	.9250	-.071	-.024	
	.9750	-.047	.041	
Wing				
Upper	.0000	.935	.970	.929
	.0100	.260	.349	.361
	.0250	-.391	-.314	-.154
	.0500	-.450	-.373	-.237
	.1000	-.592	-.462	-.391
	.2250	-.473	-.343	-.379
	.4500	-.450	-.260	-.254
Lower	.7500	-.219	-.189	-.071
	.0500	-.556	-.657	-.633
	.1500	-.473	-.485	-.444
	.4000	-.373	-.296	-.213
	.7000	-.089	-.059	-.047
Aileron				
Upper	.8050	-.249	-.154	.041
	.8150	-.237	-.142	.154
	.8350	-.166	-.178	-.219
	.8750	-.089	-.154	-.130
	.9250	-.030	-.071	-.041
	.9750	.018	.047	.036
Lower	.8100	-.036	.071	.000
	.8340	.065	.071	-.024
	.9250	-.053	.024	-.006
	.9750	.041	.071	.047

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(b) \quad \alpha_f = -4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing					Wing			
Upper	.0000	.686	.734	Upper	.0000	.781	.882	.923
	.0100	-1.065	-1.041		.0100	-.805	-.592	-.278
	.0250	-1.615	-1.669		.0250	-1.450	-1.325	-1.746
	.0500	-1.473	-1.538		.0500	-1.385	-1.225	-1.763
	.1000	-1.124	-1.183		.1000	-1.178	-.893	-.669
	.2250	-.775	-.716		.2250	-.710	-.598	-.521
	.4500	-.432	-.473		.4500	-.544	-.379	-.302
	.7500	-.314	-.515		.7500	-.260	-.237	-.095
	.8350	-.071	-.095					
	.8750	-.071	-.071					
	.9250	-.077	-.030					
	.9750	-.006	.047					
Lower	.0500	.018	.148	Lower	.0500	.041	-.053	-.183
	.1500	-.083	-.148		.1500	-.130	-.172	-.249
	.4000	-.201	-.172		.4000	-.207	-.183	-.166
	.7000	-.083	-.083		.7000	-.036	-.053	-.071
	.8530	-.101	-.065					
	.9250	-.112	-.053					
	.9750	-.024	.000					
Aileron								
Upper	.8050		-.237	Upper	.8050	-.237	-.160	-.000
	.8150		-.213		.8150	-.213	-.178	.012
	.8350		-.178		.8350	-.178	-.183	-.112
	.8750		-.071		.8750	-.071	-.142	-.136
	.9250		-.036		.9250	-.036	-.053	-.065
	.9750		.030		.9750	.030	.036	-.024
Lower	.8100		.024	Lower	.8100	.024	.089	-.041
	.8340		.112		.8340	.112	.112	-.065
	.9250		-.012		.9250	-.012	.012	-.095
	.9750		.030		.9750	.030	.065	-.024

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(c) $\alpha_f = -0.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper							
Upper	.0000	.124	.183	.207	.485	.645	
	.0100	-2.178	-2.314	-2.101	-1.663	-1.124	
	.0250	-2.574	-2.876	-2.598	-2.385	-1.467	
	.0500	-2.290	-2.550	-2.325	-2.178	-1.408	
	.1000	-1.497	-1.627	-1.651	-1.367	-1.030	
	.2250	-.929	-.905	-.929	-.799	-.651	
	.4500	-.503	-.568	-.592	-.450	-.325	
	.7500	-.243	-.414	-.219	-.260	-.148	
	.8350	-.065	-.077				
	.8750	-.036	-.030				
	.9250	-.012	-.030				
	.9750	.041	-.077				
Lower							
Lower	.0500	.479	.592	.467	.355	.195	
	.1500	.207	.142	.154	.071	-.041	
	.4000	-.030	-.006	-.036	-.089	-.107	
	.7000	.024	.024	.041	-.018	-.118	
	.8530	.024	.006				
	.9250	-.112	.006				
	.9750	-.024	.006				
Aileron							
Upper							
Upper	.8050	-.189	-.178	-.178	-.071		
	.8150	-.201	-.166	-.201	-.036		
	.8350	-.130	-.172	-.172	-.148		
	.8750	-.065	-.130	-.130	-.201		
	.9250	-.024	-.047	-.047	-.124		
	.9750	.041	.036	.036	.083		
Lower							
Lower	.8100	.083	.071	.071	-.107		
	.8340	.183	.130	.130	-.148		
	.9250	-.006	.024	.024	-.118		
	.9750	.047	.047	.047	-.071		

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(d) \quad \alpha_f = 2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing					Wing		
Upper	.8350 .8750 .9250 .9750			.0000 -.077 -.053 -.006 .012	-.811 -3.663 -3.911 -3.420 -2.124 -1.154 -.645 -.201	-.195 -3.024 -3.509 -3.201 -1.775 -.970 -.503 -.225	.172 -2.059 -2.213 -2.036 -1.320 -.793 -.391 -.254
				.0500 .1000 .2250 .4500 .7500	.746 .367 .053 .077	.680 .284 .065 .047	.450 .130 -.065 -.148
Lower	.8530 .9250 .9750			-.000 -.036 -.053			
Aileron							
Upper	.8050 .8150 .8350 .8750 .9250 .9750			-.178 -.166 -.130 -.101 -.036 .006	-.142 -.148 -.142 -.089 -.024 .012	-.189 -.142 -.219 -.308 -.237 -.189	
Lower	.8100 .8340 .9250 .9750			.112 .148 -.024 .006	.071 .166 .030 .036	-.160 -.041 -.178 -.142	

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(e) \alpha_f = 6.7^0$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-2.136	-1.947	Upper	.0000	-2.047	-.964	-.402
	.0100	-5.142	-5.686		.0100	-5.189	-4.225	-2.905
	.0250	-4.876	-5.574		.0250	-5.154	-4.479	-2.864
	.0500	-3.988	-4.556		.0500	-4.349	-4.231	-2.533
	.1000	-2.325	-2.462		.1000	-2.515	-2.130	-1.515
	.2250	-1.284	-1.260		.2250	-1.260	-1.107	-.846
	.4500	-.550	-.609		.4500	-.615	-.544	-.414
	.7500	-.172	-.107		.7500	-.148	-.219	-.420
	.8350	-.041	-.077					
	.8750	-.018	-.065					
	.9250	.000	-.036					
	.9750	.030	-.018					
Lower	.0500	.976	.953	Lower	.0500	.911	.840	.651
	.1500	.657	.598		.1500	.574	.467	.249
	.4000	.278	.266		.4000	.219	.154	-.024
	.7000	.178	.166		.7000	.148	.083	-.201
	.8530	.142	.059					
	.9250	-.089	.018					
	.9750	-.024	-.053					
Aileron								
Upper	.8050	-.112	-.160	.8050	-.112	-.160	-.284	
	.8150	-.107	-.148	.8150	-.107	-.148	.124	
	.8350	-.101	-.142	.8350	-.101	-.142	-.302	
	.8750	-.065	-.107	.8750	-.065	-.107	-.462	
	.9250	-.030	-.059	.9250	-.030	-.059	-.367	
	.9750	-.024	.012	.9750	-.024	.012	-.249	
Lower	.8100	.160	.089	.8100	.160	.089	-.314	
	.8340	.249	.189	.8340	.249	.189	.041	
	.9250	-.012	-.018	.9250	-.012	-.018	-.237	
	.9750	.012	.012	.9750	.012	.012	-.172	

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(f) $\alpha_f = 7.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.391	-1.604	Upper	.0000	-2.077
	.0100	-4.000	-5.065		.0100	-5.112
	.0250	-3.899	-5.053		.0250	-5.213
	.0500	-3.266	-4.160		.0500	-4.249
	.1000	-1.846	-2.249		.1000	-2.485
	.2250	-1.047	-1.148		.2250	-1.266
	.4500	-.615	-.604		.4500	-.663
	.7500	-.485	-.284		.7500	-.189
	.8350	-.426	-.178			
	.8750	-.391	-.142			
	.9250	-.367	-.112			
	.9750	-.308	-.077			
Lower	.0500	.893	.911		.0500	.876
	.1500	.515	.527		.1500	.544
	.4000	.166	.213		.4000	.172
	.7000	.041	.077		.7000	.118
	.8530	.036	-.065			
	.9250	-.373	-.107			
	.9750	-.290	-.148			
Aileron						
Upper	.8050	-.178	-.142	Upper	.8050	-.178
	.8150	-.154	-.130		.8150	-.154
	.8350	-.142	-.112		.8350	-.142
	.8750	-.095	-.071		.8750	-.095
	.9250	-.071	-.030		.9250	-.071
	.9750	-.059	.018		.9750	-.059
Lower	.8100	.142	.118	Lower	.8100	.142
	.8340	.154	.189		.8340	.154
	.9250	-.089	-.012		.9250	-.089
	.9750	-.024	.012		.9750	-.024

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 8.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.414	-1.757	.0000	-2.343	-1.367
	.0100	-3.959	-5.213	.0100	-5.485	-4.751
	.0250	-3.787	-5.154	.0250	-5.556	-4.888
	.0500	-3.172	-4.260	.0500	-4.509	-4.633
	.1000	-1.757	-2.296	.1000	-2.604	-2.201
	.2250	-1.024	-1.154	.2250	-1.284	-1.112
	.4500	-.657	-.604	.4500	-.633	-.491
	.7500	-.491	-.379	.7500	-.160	-.160
	.8350	-.485	-.260			
	.8750	-.420	-.178			
	.9250	-.426	-.166			
	.9750	-.402	-.130			
Lower	.0500	.905	.923	.0500	.899	.935
	.1500	.544	.544	.1500	.580	.574
	.4000	.201	.225	.4000	.189	.195
	.7000	.018	.095	.7000	.101	.118
	.8530	-.059	-.071			
	.9250	-.420	-.118			
	.9750	-.343	-.178			
Aileron						
Upper	.8050	-.189	-.112	.8050	-.189	-.112
	.8150	-.183	-.101	.8150	-.183	-.101
	.8350	-.154	-.089	.8350	-.154	-.089
	.8750	-.130	-.059	.8750	-.130	-.059
	.9250	-.071	-.006	.9250	-.071	-.006
	.9750	-.065	.030	.9750	-.065	.030
Lower	.8100	.112	.136	.8100	.112	.136
	.8340	.178	.207	.8340	.178	.207
	.9250	-.059	.006	.9250	-.059	.006
	.9750	-.047	.018	.9750	-.047	.018

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(h) \quad \alpha_f = 9.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:
		0.1621	0.2895	
Wing				
Upper	.0000	-1.633	-1.953	
	.0100	-4.225	-5.485	
	.0250	-4.030	-5.314	
	.0500	-3.278	-4.343	
	.1000	-1.822	-2.308	
	.2250	-.988	-1.160	
	.4500	-.651	-.633	
	.7500	-.544	-.414	
	.8350	-.533	-.243	
	.8750	-.497	-.213	
	.9250	-.473	-.178	
	.9750	-.438	-.154	
Lower	.0500	.876	.911	
	.1500	.586	.568	
	.4000	.201	.213	
	.7000	.047	.071	
	.8530	-.130	-.065	
	.9250	-.462	-.112	
	.9750	-.367	-.189	
Aileron				
Upper	.8050	-.195	-.154	-.343
	.8150	-.213	-.136	.148
	.8350	-.166	-.124	-.343
	.8750	-.124	-.101	-.491
	.9250	-.118	-.047	-.385
	.9750	-.101	.012	-.231
Lower	.8100	.118	.101	-.361
	.8340	.154	.201	.018
	.9250	-.071	.000	-.278
	.9750	-.083	.000	-.219

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(i) $\alpha_f = 10.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.846	-2.260	Upper	.0000	-2.970	-1.704	-.846
	.0100	-4.402	-5.769		.0100	-6.012	-5.130	-3.426
	.0250	-4.178	-5.562		.0250	-5.870	-5.160	-3.178
	.0500	-3.349	-4.462		.0500	-4.728	-4.899	-2.740
	.1000	-1.799	-2.325		.1000	-2.675	-2.314	-1.633
	.2250	-0.947	-1.124		.2250	-1.308	-1.213	-.893
	.4500	-0.627	-.604		.4500	-.609	-.538	-.444
	.7500	-0.533	-.379		.7500	-.207	-.189	-.550
	.8350	-0.544	-.237	Wing				
	.8750	-0.521	-.237	Aileron				
	.9250	-0.467	-.213	Aileron				
	.9750	-0.456	-.172	Aileron				
Lower	.0500	.935	.899	Upper	.8050	-.189	-.142	-.373
	.1500	.604	.621		.8150	-.166	-.148	.136
	.4000	.243	.266		.8350	-.166	-.124	-.355
	.7000	.053	.083		.8750	-.148	-.101	-.497
	.8530	-.089	-.077		.9250	-.124	-.053	-.385
	.9250	-.473	-.118		.9750	-.124	-.030	-.266
	.9750	-.367	-.178	Aileron				
Lower	.8100		.112	Lower	.8100	.112	.101	-.355
	.8340		.178		.8340	.178	.213	-.012
	.9250		-.089		.9250	-.089	-.018	-.249
	.9750		-.118		.9750	-.118	.000	-.213

TABLE VII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(j) \quad \alpha_f = 11.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.314	-1.408	.805	.160	-.237
	.0100	-3.882	-4.414	-2.491	-2.166	-2.485
	.0250	-3.704	-4.314	-2.142	-2.254	-2.444
	.0500	-2.994	-3.479	-1.473	-1.994	-2.166
	.1000	-1.639	-1.905	.698	.740	-1.254
	.2250	-.728	-.947	-.627	-.663	-.651
	.4500	-.568	-.556	-.746	-.716	-.379
	.7500	-.633	-.627	-.787	-.633	-.467
	.8350	-.373	-.379			
	.8750	-.426	-.396			
Lower	.9250	-.414	-.314			
	.9750	-.343	-.278			
Aileron						
Upper	.0500	.793	.905	-.509	-.544	-.420
	.1500	.444	.533	-.568	-.521	-.272
	.4000	.201	.142	-.609	-.574	-.414
	.7000	.036	.065	-.615	-.538	-.479
	.8530	-.059	-.065	-.521	-.491	-.391
	.9250	-.491	-.231	-.533	-.462	-.290
Lower	.9750	-.402	-.302			

TABLE VIII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

$$(a) \alpha_f = -8.5^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper	.0000	1.021	.987	Upper	.0000	.970	
	.0100	-.174	.009		.0100	.213	
	.0250	-.817	-.706		.0250	-.477	
	.0500	-.804	-.706		.0500	-.502	
	.1000	-.766	-.762		.1000	-.723	
	.2250	-.587	-.549		.2250	-.511	
	.4500	-.362	-.400		.4500	-.460	
	.7500	-.281	-.562		.7500	-.247	
	.8350	-.055	-.085				
	.8750	-.089	-.055				
Lower	.9250	-.089	-.043				
	.9750	-.085	.038				
	.0500	-.481	-.366	Lower	.0500	-.553	
	.1500	-.366	-.443		.1500	-.485	
	.4000	-.298	-.315		.4000	-.374	
	.7000	-.170	-.106		.7000	-.094	
Upper	.8530	-.085	-.068	Aileron			
	.9250	-.077	-.034				
	.9750	-.047	.030				
	.8050	-.247	-.191	Upper	.8050	-.247	
	.8150	-.243	-.191		.8150	-.243	
	.8350	-.162	-.217		.8350	-.162	
Lower	.8750	-.132	-.200		.8750	-.132	
	.9250	-.055	-.081		.9250	-.055	
	.9750	.009	.013		.9750	.009	
	.8100	-.043	.000		.8100	-.043	
	.8340	.089	.047		.8340	.089	
	.9250	-.051	-.004		.9250	-.051	
	.9750	.034	.047		.9750	.034	

TABLE VIII.- Continued
CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(b) \quad \alpha_f = -4.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:				C_p for values of $y/b/2$ of:		
		0.1621	0.2895			0.5000	0.6800	0.8920
Wing								
Upper	.0000	.702	.715			.770	.885	.923
	.0100	-1.191	-1.209			-.996	-.660	-.349
	.0250	-1.745	-1.881			-1.681	-1.438	-.830
	.0500	-1.600	-1.719			-1.481	-1.336	-.830
	.1000	-1.187	-1.281			-1.285	-.945	-.736
	.2250	-.800	-.757			-.736	-.638	-.536
	.4500	-.451	-.498			-.536	-.396	-.302
	.7500	-.289	-.494			-.238	-.264	-.085
	.8350	-.072	-.081					
	.8750	-.060	-.051					
	.9250	-.051	-.017					
	.9750	.009	.060					
Lower	.0500	.068	.221			.094	-.026	-.149
	.1500	-.038	-.106			-.119	-.174	-.213
	.4000	-.157	-.132			-.200	-.200	-.162
	.7000	-.068	-.030			-.021	-.064	-.072
	.8530	-.098	-.026					
	.9250	-.106	-.017					
	.9750	-.017	.004					
Aileron								
Upper	.8050		-.209			-.204	.000	
	.8150		-.204			-.187	-.017	
	.8350		-.145			-.183	-.128	
	.8750		-.128			-.170	-.162	
	.9250		-.021			-.064	-.077	
	.9750		.030			.030	-.017	
Lower	.8100		.034			.077	-.034	
	.8340		.085			.098	-.051	
	.9250		-.030			-.017	-.106	
	.9750		.030			.055	-.047	

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(c) \quad \alpha_f = -1.0^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000	-.102	.009	
	.0100	-2.613	-2.868	
	.0250	-2.906	-3.319	
	.0500	-2.566	-2.970	
	.1000	-1.664	-1.779	
	.2250	-1.055	-1.000	
	.4500	-.528	-.604	
	.7500	-.277	-.434	
	.8350	-.068	-.081	
	.8750	-.034	-.034	
	.9250	-.004	.030	
	.9750	.068	.077	
Lower	.0500	.591	.655	
	.1500	.272	.179	
	.4000	.017	.034	
	.7000	.026	.047	
	.8530	-.004	.009	
	.9250	-.111	-.017	
Wing				
Upper	.0000	.009	.447	.626
	.0100	-2.536	-1.864	-1.213
	.0250	-2.974	-2.591	-1.545
	.0500	-2.634	-2.345	-1.485
	.1000	-1.821	-1.464	-1.068
	.2250	-1.013	-.838	-.677
	.4500	-.638	-.460	-.362
Lower	.7500	-.217	-.243	-.140
	.0500	.562	.413	.213
	.1500	.213	.115	-.021
	.4000	-.034	-.068	-.102
Aileron				
Upper	.8050	-.191	-.200	-.111
	.8150	-.187	-.166	-.064
	.8350	-.145	-.166	-.174
	.8750	-.094	-.140	-.226
	.9250	-.030	-.030	-.115
	.9750	.034	.038	-.072
Lower	.8100	.081	.089	-.068
	.8340	.123	.149	-.128
	.9250	-.034	.017	-.170
	.9750	.004	.064	-.077

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(d) \quad \alpha_f = 2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.370	-1.230	Upper	.0000	-1.315
	.0100	-4.306	-4.864		.0100	-4.468
	.0250	-4.187	-4.949		.0250	-4.523
	.0500	-3.583	-4.153		.0500	-3.949
	.1000	-2.179	-2.281		.1000	-2.383
	.2250	-1.260	-1.187		.2250	-1.209
	.4500	-.604	-.651		.4500	-.643
	.7500	-.238	-.272		.7500	-.162
	.8350	-.077	-.077			
	.8750	-.034	-.030			
	.9250	-.004	.009			
	.9750	.051	.047			
Lower	.0500	.894	.949	Lower	.0500	.881
	.1500	.523	.494		.1500	.489
	.4000	.204	.221		.4000	.153
	.7000	.128	.119		.7000	.136
	.8530	.094	.047			
	.9250	-.102	.004			
	.9750	-.017	-.034			
Aileron						
Upper	.8050	-.136	-.166	Upper	.8050	-.136
	.8150	-.119	-.140		.8150	-.119
	.8350	-.102	-.123		.8350	-.102
	.8750	-.072	-.072		.8750	-.072
	.9250	-.009	-.004		.9250	-.009
	.9750	.009	.051		.9750	.009
Lower	.8100	.136	.106	Lower	.8100	.136
	.8340	.157	.200		.8340	.157
	.9250	-.034	.017		.9250	-.034
	.9750	.009	.055		.9750	.009

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(e) \quad \alpha_f = 4.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.149	-1.302	Upper	.0000	-1.655	-.762	-.268
	.0100	-3.881	-4.889		.0100	-4.817	-4.026	-2.753
	.0250	-3.902	-4.945		.0250	-5.017	-4.413	-2.698
	.0500	-3.285	-4.132		.0500	-4.166	-4.111	-2.434
	.1000	-1.928	-2.268		.1000	-2.498	-2.111	-1.502
	.2250	-1.170	-1.166		.2250	-1.234	-1.111	-.851
	.4500	-.587	-.638		.4500	-.668	-.523	-.426
	.7500	-.357	-.289		.7500	-.191	-.213	-.357
	.8350	-.294	-.149					
	.8750	-.268	-.102					
	.9250	-.226	-.072					
	.9750	-.187	-.038					
Lower	.0500	.834	.911					
	.1500	.485	.460					
	.4000	.174	.209					
	.7000	.051	.081					
	.8530	.026	-.026					
	.9250	-.281	-.072					
	.9750	-.187	-.094					
Aileron								
Upper	.8050	-.187	-.170	-.170	-.272			
	.8150	-.170	-.149	-.149	.009			
	.8350	-.123	-.132	-.132	-.255			
	.8750	-.102	-.081	-.081	-.383			
	.9250	-.060	-.026	-.026	-.306			
	.9750	-.030	.030	.030	-.204			
Lower	.8100	.106	.111	.111	-.243			
	.8340	.111	.174	.174	.047			
	.9250	-.051	.000	.000	-.255			
	.9750	-.038	.034	.034	-.153			

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(f) $\alpha_f = 5.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.272	-1.481	Upper	.0000	-2.021	-1.017	-.477
	.0100	-3.949	-5.153		.0100	-5.204	-4.404	-3.034
	.0250	-3.966	-5.140		.0250	-5.328	-4.698	-2.881
	.0500	-3.268	-4.264		.0500	-4.370	-4.379	-2.519
	.1000	-1.940	-2.302		.1000	-2.604	-2.191	-1.545
	.2250	-1.149	-1.187		.2250	-1.268	-1.145	-.851
	.4500	-.621	-.660		.4500	-.664	-.523	-.430
	.7500	-.434	-.315		.7500	-.196	-.183	-.426
	.8350	-.366	-.187	Wing				
	.8750	-.328	-.145	Aileron				
	.9250	-.302	-.102	Aileron				
	.9750	-.277	-.081	Aileron				
Lower	.0500	.860	.911	Upper	.0500	.889	.877	.651
	.1500	.511	.494		.1500	.540	.464	.260
	.4000	.191	.221		.4000	.170	.140	-.034
	.7000	.017	.072		.7000	.111	.072	-.238
	.8530	-.026	-.064		Wing			
	.9250	-.328	-.094		Aileron			
	.9750	-.234	-.132		Aileron			
					Aileron			
Lower	.8100		.102	Upper	.8050	-.162	-.145	-.302
	.8340		.128		.8150	-.162	-.153	.060
	.9250		-.060		.8350	-.145	-.136	-.302
	.9750		-.043		.8750	-.111	-.089	-.430
					.9250	-.068	-.021	-.340
					.9750	-.060	.013	-.221
				Wing				
				Aileron				
				Aileron				

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 6.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000	-1.196	-1.430	
	.0100	-3.762	-4.962	
	.0250	-3.745	-4.962	
	.0500	-3.068	-4.102	
	.1000	-1.774	-2.187	
	.2250	-1.000	-1.098	
	.4500	-.596	-.604	
	.7500	-.460	-.345	
	.8350	-.409	-.179	
	.8750	-.379	-.149	
	.9250	-.362	-.123	
	.9750	-.328	-.094	
Lower	.0500	.851	.932	
	.1500	.511	.540	
	.4000	.209	.230	
	.7000	.043	.094	
	.8530	-.051	-.060	
	.9250	-.362	-.077	
	.9750	-.272	-.111	
Wing				
Upper	.0000	-1.940	-1.034	-.523
	.0100	-5.055	-4.336	-3.043
	.0250	-5.191	-4.647	-2.902
	.0500	-4.255	-4.349	-2.502
	.1000	-2.523	-2.166	-1.523
	.2250	-1.243	-1.115	-.847
	.4500	-.634	-.523	-.417
Lower	.7500	-.200	-.217	-.421
	.0500	.898	.864	.647
	.1500	.557	.455	.243
	.4000	.179	.140	-.043
Aileron	.7000	.115	.060	-.255
	.8050	-.153	-.166	-.302
	.8150	-.157	-.162	.106
	.8350	-.136	-.145	-.281
	.8750	-.106	-.106	-.421
	.9250	-.060	-.055	-.319
	.9750	-.047	-.004	-.213
	.8100	.098	.064	-.311
	.8340	.132	.183	.081
	.9250	-.043	.004	-.243
	.9750	-.030	.013	-.170

TABLE VIII.- Continued
 CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON
 $\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

(h) $\alpha_f = 7.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.298	-1.719	.0000	-2.494	-1.447
	.0100	-3.940	-5.387	.0100	-5.698	-4.966
	.0250	-3.906	-5.328	.0250	-5.672	-5.132
	.0500	-3.174	-4.340	.0500	-4.626	-4.757
	.1000	-1.809	-2.319	.1000	-2.719	-2.285
	.2250	-1.026	-1.136	.2250	-1.302	-1.140
	.4500	-.591	-.634	.4500	-.634	-.489
	.7500	-.481	-.391	.7500	-.187	-.170
	.8350	-.426	-.183			
	.8750	-.438	-.170			
	.9250	-.409	-.136			
	.9750	-.391	-.115			
Lower	.0500	.902	.970	.0500	.928	.936
	.1500	.579	.553	.1500	.621	.587
	.4000	.221	.264	.4000	.209	.204
	.7000	.068	.098	.7000	.136	.119
	.8530	-.094	-.060			
	.9250	-.400	-.089			
	.9750	-.306	-.136			
Aileron						
Upper	.8050	-.149	-.102	.8050	-.149	-.102
	.8150	-.149	-.102	.8150	-.136	-.094
	.8350	-.136	-.064	.8350	-.115	-.064
	.8750	-.077	-.034	.8750	-.055	-.004
	.9250			.9250		
	.9750			.9750		
Lower	.8100	.140	.115	.8100	.140	.115
	.8340	.140	.196	.8340	.047	.004
	.9250	-.047	.030	.9250	-.047	-.247
	.9750	-.047	.030	.9750	-.047	-.204

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(i) \quad \alpha_f = 8.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.506	-1.877	Upper	.0000	-2.574
	.0100	-4.038	-5.498		.0100	-5.791
	.0250	-3.928	-5.336		.0250	-5.719
	.0500	-3.166	-4.340		.0500	-4.638
	.1000	-1.817	-2.315		.1000	-2.715
	.2250	-0.987	-1.149		.2250	-1.294
	.4500	-0.630	-0.634		.4500	-0.626
	.7500	-0.540	-0.438		.7500	-0.179
	.8350	-0.502	-0.243			
	.8750	-0.485	-0.217			
Lower	.9250	-0.477	-0.179			
	.9750	-0.464	-0.128			
	.0500	.855	.932	Lower	.0500	.936
	.1500	.570	.557		.1500	.634
	.4000	.243	.264		.4000	.230
	.7000	.051	.089		.7000	.123
	.8530	-0.157	-0.072			
	.9250	-0.451	-0.115			
	.9750	-0.366	-0.162			
Aileron						
Upper	.8050	-0.166	-0.157	Upper	.8050	-0.166
	.8150	-0.166	-0.153		.8150	-0.166
	.8350	-0.145	-0.140		.8350	-0.145
	.8750	-0.119	-0.102		.8750	-0.119
	.9250	-0.098	-0.060		.9250	-0.098
	.9750	-0.089	-0.021		.9750	-0.089
Lower	.8100	.098	.077	Lower	.8100	.098
	.8340	.119	.166		.8340	.119
	.9250	-0.068	-0.021		.9250	-0.068
	.9750	-0.047	.000		.9750	-0.047

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(j) \quad \alpha_f = 9.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.000	-.617	Upper	.0000	-.681	-.638	-.434
	.0100	-3.340	-2.698		.0100	-2.421	-3.574	-2.872
	.0250	-3.285	-2.187		.0250	-1.906	-3.915	-2.749
	.0500	-2.694	-1.451		.0500	-1.285	-3.536	-2.387
	.1000	-1.494	-1.068		.1000	-.885	-1.711	-1.443
	.2250	-.851	-.953		.2250	-.902	-.911	-.791
	.4500	-.638	-.843		.4500	-.813	-.574	-.379
	.7500	-.540	-.570		.7500	-.549	-.438	-.374
	.8350	-.438	-.489					
	.8750	-.434	-.477					
	.9250	-.421	-.443					
	.9750	-.409	-.409					
Lower	.0500	.817	.864	Lower	.0500	.809	.766	.583
	.1500	.494	.438		.1500	.455	.417	.217
	.4000	.157	.162		.4000	.106	.077	-.072
	.7000	-.004	.017		.7000	.021	.009	-.226
	.8530	-.102	-.123					
	.9250	-.404	-.213					
	.9750	-.315	-.357					
Aileron								
Upper	.8050	-.404	-.349	Upper	.8050	-.404	-.349	-.272
	.8150	-.404	-.349		.8150	-.464	-.374	-.060
	.8350	-.464	-.374		.8350	-.396	-.349	-.238
	.8750	-.374	-.302		.8750	-.374	-.302	-.345
	.9250	-.353	-.264		.9250	-.353	-.264	-.247
	.9750	-.353	-.264		.9750	-.353	-.264	-.153
Lower	.8100	-.026	.060	Lower	.8100	-.026	.060	-.230
	.8340	.017	.102		.8340	.017	.102	.072
	.9250	-.226	-.136		.9250	-.226	-.136	-.196
	.9750	-.268	-.149		.9750	-.268	-.149	-.157

TABLE VIII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(k) \quad \alpha_f = 10.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.136	-.638	Upper	.0000	-.613	-.566	-.502
	.0100	-3.515	-2.689		.0100	-2.187	-3.443	-2.945
	.0250	-3.345	-2.145		.0250	-1.540	-3.813	-2.787
	.0500	-2.732	-1.430		.0500	-1.047	-3.455	-2.438
	.1000	-1.485	-1.017		.1000	-.779	-1.702	-1.464
	.2250	-.821	-.940		.2250	-.834	-.919	-.787
	.4500	-.634	-.847		.4500	-.813	-.638	-.379
	.7500	-.511	-.587		.7500	-.600	-.489	-.413
	.8350	-.455	-.506					
	.8750	-.451	-.502					
	.9250	-.434	-.460					
	.9750	-.400	-.413					
Lower	.0500	.872	.885	Lower	.0500	.804	.770	.583
	.1500	.511	.472		.1500	.464	.417	.226
	.4000	.174	.191		.4000	.115	.094	-.085
	.7000	.021	.021		.7000	.034	.004	-.251
	.8530	-.145	-.128					
	.9250	-.417	-.226					
	.9750	-.336	-.379					
Aileron								
Upper	.8050	-.417	-.391	-.294				
	.8150	-.455	-.391	-.055				
	.8350	-.485	-.430	-.277				
	.8750	-.426	-.396	-.374				
	.9250	-.396	-.332	-.277				
	.9750	-.349	-.281	-.183				
Lower	.8100	-.017	.030	-.281				
	.8340	.000	.089	.055				
	.9250	-.226	-.140	-.247				
	.9750	-.272	-.132	-.204				

TABLE IX

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

$$(a) \quad \alpha_f = -6.5^{\circ}$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
		Wing				
Upper	.0000		.968			
	.0100		-.192			
	.0250		-.915			
	.0500		-.822			
	.1000		-.964			
	.2250		-.623			
	.4500		-.445			
	.7500		-.612			
	.8350		-.114			
	.8750		-.071			
Lower	.9250		-.032			
	.9750		.050			
	.0500		-.310			
	.1500		-.480			
	.4000		-.292			
	.7000		-.139			
	.8530		-.064			
Upper	.9250		-.032			
	.9750		.039			
Lower	.0500		-.516			
	.1500		-.484			
	.4000		-.349			
	.7000		-.082			
	.0500		-.762			
	.1500		-.548			
	.4000		-.324			
	.7000		.032			
	.0500		-.769			
	.1500		-.470			
Aileron						
Upper	.8050		-.313			
	.8150		-.299			
	.8350		-.160			
	.8750		-.121			
	.9250		-.025			
	.9750		.011			
	.8050		-.413			
Lower	.8150		-.423			
	.8350		-.370			
	.8750		-.292			
	.9250		-.103			
	.9750		.036			
Lower	.8100		.050			
	.8340		-.103			
	.9250		-.093			
	.9750		.028			
	.8100		.142			
Lower	.8340		.199			
	.9250		.068			
	.9750		.071			
	.8100		.125			
Lower	.8340		.100			
	.9250		.064			
	.9750		.057			

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(b) \quad \alpha_f = -4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.875	.883	Upper	.0000	.890
	.0100	-.762	-.623		.0100	-.217
	.0250	-1.402	-1.359		.0250	-.986
	.0500	-1.260	-1.224		.0500	-.936
	.1000	-1.064	-1.093		.1000	-.986
	.2250	-.765	-.705		.2250	-.623
	.4500	-.459	-.512		.4500	-.552
	.7500	-.224	-.498		.7500	-.302
	.8350	-.053	-.117			
	.8750	-.053	-.082			
	.9250	-.053	-.057			
	.9750	-.053	.028			
Lower	.0500	-.185	-.046	Lower	.0500	-.210
	.1500	-.246	-.267		.1500	-.324
	.4000	-.181	-.246		.4000	-.267
	.7000	-.125	-.089		.7000	-.032
	.8530	-.057	-.043			
	.9250	-.110	-.057			
	.9750	-.000	-.018			
Aileron						
Upper	.8050	-.317	-.416	Upper	.8050	-.317
	.8150	-.302	-.416		.8150	-.302
	.8350	-.181	-.367		.8350	-.181
	.8750	-.117	-.256		.8750	-.117
	.9250	-.039	-.096		.9250	-.039
	.9750	.021	.018		.9750	.021
Lower	.8100	.075	.153	Lower	.8100	.075
	.8340	.157	.221		.8340	.157
	.9250	-.039	.057		.9250	-.039
	.9750	.025	.060		.9750	.025

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(c) \quad \alpha_f = -2.8^0$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.648	.637	Upper	.722	.936
	.0100	-1.370	-1.416		-1.028	-.580
	.0250	-1.972	-2.085		-1.719	-.740
	.0500	-1.754	-1.875		-1.552	-.776
	.1000	-1.267	-1.381		-1.310	-.751
	.2250	-.897	-.833		-.776	-.555
	.4500	-.473	-.584		-.569	-.338
	.7500	-.285	-.516		-.274	-.327
	.8350	-.014	-.057			
	.8750	-.018	-.032			
	.9250	-.028	.014			
	.9750	.043	.071			
Lower	.0500	.238	.320	Lower	.160	-.064
	.1500	.064	-.057		-.075	-.217
	.4000	-.100	-.121		-.164	-.238
	.7000	-.014	-.036		-.000	-.132
	.8530	-.135	-.014			-.011
	.9250	-.085	-.004			
	.9750	-.000	-.004			
Aileron						
Upper	.8050	-.313	-.406	Upper	-.313	-.313
	.8150	-.263	-.388		-.281	-.281
	.8350	-.192	-.345		-.466	-.466
	.8750	-.117	-.228		-.299	-.299
	.9250	-.025	-.071		-.153	-.153
	.9750	.025	.032		-.050	-.050
Lower	.8100	.075	.189	Lower	.164	
	.8340	.149	.274		.128	
	.9250	-.050	.078		-.028	
	.9750	.032	.060		-.028	

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(d) \quad \alpha_f = -1.0^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	.185	.263	Upper	.0000	.356	.769	.875
	.0100	-2.192	-2.299		.0100	-1.836	-1.199	-0.683
	.0250	-2.616	-2.868		.0250	-2.452	-1.975	-1.128
	.0500	-2.320	-2.569		.0500	-2.171	-1.829	-1.114
	.1000	-1.548	-1.687		.1000	-1.637	-1.224	-0.907
	.2250	-1.021	-0.986		.2250	-0.918	-0.751	-0.626
	.4500	-.651	-.651		.4500	-.655	-.448	-.363
	.7500	-.281	-.512		.7500	-.288	-.367	-.324
	.8350	-.053	-.082					
	.8750	-.046	-.025					
	.9250	-.078	-.007					
	.9750	.050	.060					
Lower	.0500	.388	.527					
	.1500	.174	.060					
	.4000	-.004	-.060					
	.7000	-.078	-.014					
	.8530	-.100	-.021					
	.9250	-.128	-.032					
	.9750	-.018	-.028					
Aileron								
Upper	.8050	-.302	-.356	.8050	-.302	-.356	-.285	-.295
	.8150	-.249	-.342	.8150	-.249	-.342	-.459	-.459
	.8350	-.189	-.285	.8350	-.189	-.285	-.327	-.327
	.8750	-.110	-.185	.8750	-.110	-.185	-.174	-.174
	.9250	-.039	-.053	.9250	-.039	-.053	-.064	-.064
	.9750	.004	.039	.9750	.004	.039		
Lower	.8100	.125	.210	.8100	.125	.210	.157	.125
	.8340	.157	.288	.8340	.157	.288	.025	.025
	.9250	-.057	.082	.9250	-.057	.082		
	.9750	.007	.064	.9750	.007	.064	-.064	-.064

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(e) \quad \alpha_f = 0.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper				.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-.256 -2.843 -3.402 -2.929 -1.964 -1.046 -.719 -.178	.459 -1.993 -2.744 -2.509 -1.541 -.904 -.527 -.363	.648 -1.267 -1.616 -1.537 -1.146 -.726 -.402 -.374
Lower				.0500 .1500 .4000 .7000	.598 .249 .000 .032	.441 .125 -.039 .096	.235 .000 -.082 -.064
Aileron							
Upper				.8050 .8150 .8350 .8750 .9250 .9750	-.285 -.206 -.157 -.085 -.021 -.028	-.342 -.335 -.278 -.164 -.060 .028	-.335 -.324 -.548 -.406 -.238 -.117
Lower				.8100 .8340 .9250 .9750	.064 .114 -.028 -.046	.199 .292 .068 .050	.121 .078 -.132 -.135

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(f) $\alpha_f = 1.4^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-.459	-.356	.313	.530	
	.0100	-3.117	-3.484	-2.246	-1.434	
	.0250	-3.452	-3.851	-2.972	-1.754	
	.0500	-2.911	-3.381	-2.982	-1.648	
	.1000	-1.815	-1.982	-1.996	-1.185	
	.2250	-1.096	-1.075	-1.050	-.943	-.747
	.4500	-.552	-.665	-.662	-.527	-.391
	.7500	-.221	-.406	-.221	-.345	-.384
	.8350	.021	-.046			
	.8750	.021	-.014			
	.9250	.068	.036			
	.9750	.117	.093			
Lower	.0500	.836	.836	.523	.302	
	.1500	.423	.313	.189	.036	
	.4000	.217	.103	.032	-.007	-.057
	.7000	.132	.060	.082	.103	-.085
	.8530	-.050	.036			
	.9250	-.146	.000			
	.9750	-.014	-.036			
Aileron						
Upper	.8050	-.242	-.306	-.338		
	.8150	-.181	-.302	-.313		
	.8350	-.146	-.238	-.541		
	.8750	-.064	-.149	-.413		
	.9250	-.032	-.046	-.242		
	.9750	.007	.025	-.110		
Lower	.8100	.135	.203	.142		
	.8340	.153	.295	.107		
	.9250	-.043	.064	-.125		
	.9750	.014	.039	-.117		

TABLE IX..- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 2.3^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.943			
	.0100		-4.548			
	.0250		-4.829			
	.0500		-3.904			
	.1000		-2.263			
	.2250		-1.149			
	.4500		-.690			
	.7500		-.342			
	.8350		-.036			
	.8750		.014			
	.9250		.060			
	.9750		.103			
Lower	.0500		.961			
	.1500		.488			
	.4000		.256			
	.7000		.146			
	.8530		.093			
	.9250		.078			
	.9750		.014			
Aileron						
Upper	.8050		-.174			
	.8150		-.128			
	.8350		-.093			
	.8750		-.057			
	.9250		.007			
	.9750		.014			
Lower	.8100		.078			
	.8340		.117			
	.9250		-.064			
	.9750		.039			

TABLE IX.—Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

$$(h) \quad \alpha_f = 2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:					
		0.1621	0.2895	0.5000	0.6800	0.8920			
		Wing							
Upper	.0000	-1.189	-1.075	Upper	.0000	-1.224	-.199	.153	
	.0100	-4.064	-4.648		.0100	-4.242	-3.242	-2.114	
	.0250	-4.228	-4.772		.0250	-4.491	-3.836	-2.281	
	.0500	-3.402	-4.011		.0500	-3.744	-3.509	-2.085	
	.1000	-2.107	-2.249		.1000	-2.327	-1.936	-1.388	
	.2250	-1.242	-1.196		.2250	-1.185	-1.075	-.826	
	.4500	-.587	-.669		.4500	-.673	-.559	-.441	
	.7500	-.199	-.342		.7500	-.210	-.295	-.420	
	.8350	-.018	-.064	Lower					
	.8750	-.036	-.032	.0500	.822	.726	.477		
	.9250	-.007	.007	.1500	.434	.363	.142		
	.9750	.100	.050	.4000	.128	.075	-.057		
Lower	.0500	.911	.833	.7000	.107	.125	-.142		
	.1500	.559	.470	Aileron					
	.4000	.288	.217						
	.7000	.085	.125						
	.8530	.032	.100						
	.9250	-.128	.007						
	.9750	.025	-.032						
Upper	.8050	-.149	-.242	Upper	.8100	.128	.228	-.399	
	.8150	-.157	-.228		.8340	.128	.420	-.367	
	.8350	-.117	-.185		.9250	-.028	.064	-.555	
	.8750	-.060	-.110		.9750	.000	.032	-.491	
	.9250	-.007	-.036		Lower				
	.9750	.000	.032						
Lower	.8100	.128	.228	Lower	.8340	.128	.420	-.036	
	.8340	.128	.420		.9250	-.028	.064	-.053	
	.9250	-.028	.064		.9750	.000	.032	-.160	
	.9750	.000	.032						

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(i) \quad \alpha_f = 3.3^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.459	-1.299			
	.0100	-4.484	-5.064			
	.0250	-4.552	-5.028			
	.0500	-3.559	-4.178			
	.1000	-2.285	-2.352			
	.2250	-1.370	-1.267			
	.4500	-0.765	-0.712			
	.7500	-0.231	-0.320			
	.8350	-0.014	-0.110			
	.8750	-0.050	-0.075			
	.9250	-0.046	-0.036			
	.9750	0.046	0.014			
Lower	.0500	.441	.865			
	.1500	.544	.452			
	.4000	.199	.206			
	.7000	.096	.160			
	.8530	-.043	.025			
	.9250	-.164	-.018			
	.9750	-.075	-.071			
Aileron						
Upper	.8050	-.181	-.238			
	.8150	-.146	-.224			
	.8350	-.114	-.178			
	.8750	-.075	-.117			
	.9250	-.036	-.039			
	.9750	.011	.018			
Lower	.8100	.110	.221			
	.8340	.082	.409			
	.9250	-.043	.046			
	.9750	-.000	.036			

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(j) \quad \alpha_f = 3.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		
		0.1621	0.2895	
Wing				
Upper	.0000	-1.794	-1.769	
	.0100	-4.758	-5.626	
	.0250	-4.786	-5.505	
	.0500	-3.815	-4.544	
	.1000	-2.441	-2.512	
	.2250	-1.537	-1.388	
	.4500	-0.779	-0.769	
	.7500	-0.374	-0.278	
	.8350	-0.196	-0.121	
	.8750	-0.114	-0.075	
	.9250	-0.071	-0.036	
	.9750	-0.011	-0.050	
Lower	.0500	.808	.879	
	.1500	.495	.448	
	.4000	.270	.174	
	.7000	.021	.014	
	.8530	.021	-0.078	
	.9250	-0.192	-0.114	
	.9750	-0.149	-0.043	
Aileron				
Upper	.8050	-.121	-.185	-.395
	.8150	-.110	-.178	-.381
	.8350	-.057	-.142	-.555
	.8750	-.021	-.089	-.548
	.9250	.004	-.021	-.399
	.9750	.103	.014	-.221
Lower	.8100	.071	.206	-.018
	.8340	-.053	.384	.004
	.9250	-.018	.032	-.210
	.9750	.021	.021	-.192

TABLE IX.- Continued
CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(k) $\alpha_f = 4.2^0$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.893	-1.922	.968	-.441	
	.0100	-4.890	-5.886	-4.548	-3.053	
	.0250	-5.025	-5.786	-5.673	-2.922	
	.0500	-3.922	-4.726	-5.644	-2.544	
	.1000	-2.391	-2.630	-4.601	-1.584	
	.2250	-1.388	-1.292	-2.715	-.890	
	.4500	-.523	-.687	-1.363	-.434	
	.7500	-.164	-.199	-.690	-.402	
	.8350	-.142	-.107	-.164		
	.8750	-.075	-.071			
	.9250	-.025	-.043			
	.9750	.046	-.021			
Lower	.0500	.957	.936	.943	.648	
	.1500	.690	.544	.562	.278	
	.4000	.313	.274	.214	-.014	
	.7000	.125	.132	.125	-.171	
	.8530	.050	.057			
	.9250	-.135	.011			
	.9750	-.018	-.082			
Aileron						
Upper	.8050	-.135	-.132	-.132	-.377	
	.8150	-.121	-.125	-.125	-.352	
	.8350	-.093	-.107	-.107	-.498	
	.8750	-.075	-.057	-.057	-.527	
	.9250	-.025	-.000	-.000	-.370	
	.9750	-.021	.039	.039	-.196	
Lower	.8100	.078	.203	.203	-.028	
	.8340	.107	.391	.391	.060	
	.9250	-.025	.053	.053	-.189	
	.9750	-.011	.036	.036	-.196	

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(l) \quad \alpha_f = 4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:				C_p for values of y/b of:
		0.1621	0.2895			
Wing						
Upper	.0000	-2.057	-2.089	Upper	.0000	-2.730
	.0100	-5.135	-6.128		.0100	-5.982
	.0250	-5.064	-5.879		.0250	-5.900
	.0500	-4.025	-4.762		.0500	-4.808
	.1000	-2.438	-2.658		.1000	-2.826
	.2250	-1.406	-1.384		.2250	-1.434
	.4500	-.665	-.680		.4500	-.701
	.7500	-.295	-.224		.7500	-.181
	.8350	-.174	-.139			
	.8750	-.157	-.139			
	.9250	-.114	-.107			
	.9750	-.089	-.071			
Lower	.0500	.900	.918	Lower	.0500	.886
	.1500	.605	.562		.1500	.573
	.4000	.242	.267		.4000	.203
	.7000	.100	.096		.7000	.082
	.8530	.060	.011			
	.9250	-.189	-.053			
	.9750	-.117	-.132			
Aileron						
Upper	.8050	-.146	-.167	Upper	.8050	-.146
	.8150	-.149	-.149		.8150	-.149
	.8350	-.139	-.132		.8350	-.139
	.8750	-.110	-.096		.8750	-.110
	.9250	-.075	-.050		.9250	-.075
	.9750	-.039	-.007		.9750	-.039
Lower	.8100	.043	.167	Lower	.8100	.043
	.8340	.057	.352		.8340	.057
	.9250	-.085	-.004		.9250	-.085
	.9750	-.053	.004		.9750	-.053

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(m) \quad \alpha_f = 5.2^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.552	-1.797	Upper	-2.591	-1.359
	.0100	-4.480	-5.577		-5.698	-4.968
	.0250	-4.480	-5.345		-5.722	-5.174
	.0500	-3.584	-4.281		-4.598	-4.701
	.1000	-2.110	-2.452		-2.609	-2.338
	.2250	-1.167	-1.335		-1.367	-1.235
	.4500	-.559	-.676		-.680	-.562
	.7500	-.302	-.238		-.174	-.192
	.8350	-.263	-.160			
	.8750	-.246	-.132			
	.9250	-.164	-.093			
	.9750	-.103	-.071			
Lower	.0500	.950	.972	Lower	.893	.843
	.1500	.623	.559		.559	.502
	.4000	.320	.253		.203	.160
	.7000	.121	.135		.093	.078
	.8530	.039	.007			
	.9250	-.242	-.046			
	.9750	-.139	-.114			
Aileron						
Upper	.8050	-.153	-.160	Upper	-.153	-.160
	.8150	-.146	-.142		-.146	-.142
	.8350	-.135	-.121		-.135	-.121
	.8750	-.107	-.100		-.107	-.100
	.9250	-.068	-.057		-.068	-.057
	.9750	-.043	-.018		-.043	-.018
Lower	.8100	.060	.164	Lower	.060	.164
	.8340	.078	.345		.078	.345
	.9250	-.071	.000		-.071	.000
	.9750	-.039	.000		-.039	.000

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(n) \quad \alpha_f = 5.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.630			
	.0100		-5.388			
	.0250		-5.292			
	.0500		-4.345			
	.1000		-2.406			
	.2250		-1.171			
	.4500		-.562			
	.7500		-.270			
	.8350		-.132			
	.8750		-.089			
	.9250		-.117			
	.9750		-.100			
Lower	.0500		.918			
	.1500		.616			
	.4000		.263			
	.7000		.135			
	.8530		.000			
	.9250		-.032			
Aileron	.0500		-.171			
	.8150		-.135			
	.8350		-.100			
	.8750		-.082			
	.9250		-.064			
	.9750		-.007			
	.8100		.050			
	.8340		.046			
	.9250		-.032			
	.9750		-.043			
				.153		
				.349		

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^{\circ}$; cable configuration 2-5-8; $q_{\infty} = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(o) \quad \alpha_f = 6.4^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing					Wing		
Upper	.0000	-1.577	-1.950	.0000	-2.836	-1.477	-.872
	.0100	-4.431	-5.762	.0100	-5.975	-5.199	-3.566
	.0250	-4.441	-5.559	.0250	-5.861	-5.356	-3.228
	.0500	-3.470	-4.463	.0500	-4.715	-4.836	-2.794
	.1000	-2.110	-2.544	.1000	-2.747	-2.388	-1.651
	.2250	-1.206	-1.388	.2250	-1.402	-1.214	-.886
	.4500	-.765	-.708	.4500	-.694	-.562	-.423
	.7500	-.491	-.349	.7500	-.231	-.178	-.416
	.8350	-.470	-.164				
	.8750	-.409	-.178				
Lower	.9250	-.399	-.174				
	.9750	-.327	-.135				
Upper	.0500	.747	.854	.0500	.868	.861	.683
	.1500	.520	.619	.1500	.552	.537	.299
	.4000	.238	.128	.4000	.181	.171	-.007
	.7000	.007	-.011	.7000	.028	.068	-.196
	.8530	.018	-.075				
	.9250	-.359	-.125				
	.9750	-.256	-.199				
Aileron							
Upper	.8050		-.196	.8050	-.196	-.142	-.391
	.8150		-.189	.8150	-.189	-.114	-.260
	.8350		-.160	.8350	-.160	-.096	-.459
	.8750		-.121	.8750	-.121	-.068	-.516
	.9250		-.107	.9250	-.107	-.032	-.370
	.9750		-.053	.9750	-.053	-.000	-.210
Lower	.8100		.007	.8100	.007	.164	-.078
	.8340		.046	.8340	.046	.363	.004
	.9250		-.110	.9250	-.110	-.007	-.249
	.9750		-.078	.9750	-.078	.007	-.224

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(p) \quad \alpha_f = 6.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper						
	.0000	-2.865	-1.302	-0.680		
	.0100	-6.053	-4.986	-3.352		
	.0250	-5.932	-5.235	-3.093		
	.0500	-4.836	-4.794	-2.715		
	.1000	-2.769	-2.370	-1.633		
	.2250	-1.416	-1.203	-0.893		
	.4500	-0.765	-0.544	-0.448		
Lower	.7500	-0.253	-0.178	-0.402		
	.0500	.893	.883	.665		
	.1500	.619	.544	.281		
	.4000	.238	.185	-.025		
Lower	.7000	.028	.103	-.203		
Aileron						
Upper	.8050	-.078	-.139	-.391		
	.8150	-.142	-.125	-.288		
	.8350	-.128	-.103	-.463		
	.8750	-.117	-.068	-.512		
	.9250	-.093	-.032	-.363		
	.9750	-.093	.011	-.210		
Lower						
	.8100	-.004	.149	-.093		
	.8340	.078	.377	-.039		
	.9250	-.093	.021	-.267		
	.9750	-.139	.021	-.221		

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

$$(q) \quad \alpha_f = 7.3^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-2.018		-1.694	-1.082
	.0100		-5.815		-5.413	-3.819
	.0250		-5.544		-5.530	-3.370
	.0500		-4.477		-4.819	-2.922
	.1000		-2.448		-2.783	-1.683
	.2250		-1.181		-1.352	-0.868
	.4500		-.619		-.580	-0.409
	.7500		-.406		-.228	-0.452
	.8350		-.157			
	.8750		-.121			
	.9250		-.174			
	.9750		-.053			
Lower	.0500		.975		.865	.712
	.1500		.662		.559	.317
	.4000		.270		.199	-0.028
	.7000		.214		.075	-0.253
	.8530		.028			
	.9250		-.110			
	.9750		-.149			
Aileron						
Upper	.8050		-.157		-.139	-0.431
	.8150		-.149		-.121	-0.210
	.8350		-.135		-.110	-0.420
	.8750		-.103		-.089	-0.473
	.9250		-.068		-.071	-0.327
	.9750		-.050		-.043	-0.206
Lower	.8100		.057		.132	-0.142
	.8340		.110		.324	-0.032
	.9250		-.078		-.025	-0.256
	.9750		-.028		-.007	-0.206

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(r) \quad \alpha_f = 7.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.270	-1.313	Upper	.0000	-1.769	-.925	-.559
	.0100	-3.897	-5.530		.0100	-4.338	-4.174	-3.078
	.0250	-3.865	-5.217		.0250	-5.039	-4.544	-2.858
	.0500	-3.050	-4.100		.0500	-2.957	-4.181	-2.505
	.1000	-1.797	-2.224		.1000	-1.769	-2.025	-1.498
	.2250	-1.046	-1.078		.2250	-1.043	-1.011	-.815
	.4500	-.562	-.580		.4500	-.594	-.456	-.374
	.7500	-.409	-.306		.7500	-.256	-.238	-.391
	.8350	-.342	-.189					
	.8750	-.260	-.231					
	.9250	-.274	-.196					
	.9750	-.285	-.128					
Lower	.0500	.900	1.014	Lower	.0500	.943	.872	.630
	.1500	.609	.683		.1500	.623	.534	.285
	.4000	.238	.295		.4000	.263	.189	-.011
	.7000	.025	.174		.7000	.174	.121	-.160
	.8530	.068	.050					
	.9250	-.253	-.004					
	.9750	-.185	-.146					
Aileron								
Upper	.8050	-.210	-.189	-.384	Upper	.8100	.203	.203
	.8150	-.139	-.164	-.292		.8340	.203	.370
	.8350	-.192	-.164	-.356		.9250	-.025	.014
	.8750	-.082	-.142	-.381		.9750	-.053	-.082
	.9250	-.039	-.096	-.249				
	.9750	-.053	-.082	-.110				
Lower	.8100	.203	.203	.021	Lower	.8340	.203	.114
	.8340	.203	.370	-.117		.9250	-.025	-.014
	.9250	-.025	.014	-.103		.9750	.025	-.014
	.9750	.025	-.014					

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(s) $\alpha_f = 8.4^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.000	-1.349			
	.0100	-3.021	-4.167			
	.0250	-3.018	-4.068			
	.0500	-2.441	-3.295			
	.1000	-1.388	-1.851			
	.2250	-.815	-.911			
	.4500	-.520	-.498			
	.7500	-.399	-.335			
	.8350	-.370	-.203			
	.8750	-.349	-.189			
	.9250	-.338	-.142			
	.9750	-.327	-.107			
Lower	.0500	.676	.779			
	.1500	.423	.427			
	.4000	.167	.196			
	.7000	.011	.071			
	.8530	.043	-.057			
	.9250	-.356	-.103			
	.9750	-.274	-.146			
Aileron						
Upper	.8050	-.121	-.142	-.409		
	.8150	-.114	-.132	-.288		
	.8350	-.103	-.114	-.370		
	.8750	-.082	-.082	-.377		
	.9250	-.057	-.050	-.242		
	.9750	-.039	-.032	-.117		
Lower	.8100	.057	.096	-.028		
	.8340	.071	.224	.028		
	.9250	-.064	.000	-.132		
	.9750	-.028	-.014	-.135		

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(t) $\alpha_f = 8.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.737			
	.0100		-5.253			
	.0250		-4.804			
	.0500		-3.836			
	.1000		-2.210			
	.2250		-1.274			
	.4500		-0.790			
	.7500		-0.598			
	.8350		-0.356			
	.8750		-0.306			
	.9250		-0.306			
	.9750		-0.278			
Lower	.0500		.890			
	.1500		.516			
	.4000		.206			
	.7000		-.057			
	.8530		-.107			
	.9250		-.174			
	.9750		-.274			
Wing						
Upper	.0000		-2.548			
	.0100		-5.295			
	.0250		-5.046			
	.0500		-3.943			
	.1000		-2.214			
	.2250		-1.352			
	.4500		-0.783			
Lower	.7500		-.399			
	.0500		.836			
	.1500		.505			
	.4000		.128			
Aileron	.7000		-.014			
	.8050		-.320			
	.8150		-.317			
	.8350		-.320			
	.8750		-.310			
	.9250		-.292			
	.9750		-.246			
Upper	.8100		-.043			
	.8340		.011			
	.9250		-.210			
	.9750		-.217			
Lower	.8100		.096			
	.8340		.295			
	.9250		-.028			
	.9750		-.028			

TABLE IX.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 16.9 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(u) \quad \alpha_f = 9.4^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper		.0000	-2.217	-1.370	-.833	
		.0100	-6.149	-4.915	-3.498	
		.0250	-5.922	-5.064	-3.171	
		.0500	-3.544	-4.512	-2.744	
		.1000	-2.117	-2.253	-1.616	
		.2250	-1.335	-1.149	-.879	
		.4500	-.797	-.580	-.420	
		.7500	-.548	-.299	-.431	
Lower		.0500	.968	.836	.676	
		.1500	.541	.495	.281	
		.4000	.249	.174	-.018	
		.7000	.046	.085	-.224	
Aileron						
Upper		.8050	-.249	-.231	-.423	
		.8150	-.342	-.224	-.281	
		.8350	-.292	-.217	-.505	
		.8750	-.228	-.196	-.523	
		.9250	-.246	-.142	-.388	
		.9750	-.299	-.142	-.270	
Lower		.8100	.014	.064	-.153	
		.8340	-.046	.274	-.021	
		.9250	-.164	-.021	-.267	
		.9750	-.203	-.039	-.263	

TABLE X

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(a) \quad \alpha_f = -6.6^0$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
		Wing					
Upper				.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.881 .215 .478 .409 .716 .472 .525 .313	.907 .475 .203 .269 .379 .328 .290 .415	.878 .504 .012 .090 .340 .358 .293 .266
Lower				.0500 .1500 .4000 .7000	-.615 -.510 -.370 -.113	-.842 -.594 -.328 .036	-.845 -.501 -.221 .039
Aileron							
Upper				.8050 .8150 .8350 .8750 .9250 .9750	-.310 -.281 -.197 -.152 -.054 -.000	-.382 -.388 -.346 -.269 -.096 .033	-.197 -.173 -.364 -.218 -.054 .042
Lower				.8100 .8340 .9250 .9750	.024 .084 -.054 .048	.146 .227 .072 .018	.096 .110 .087 .069

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.887	.901	Upper	.916	.997
	.0100	-.818	-.675		-.346	.030
	.0250	-1.451	-1.427		-1.030	-.704
	.0500	-1.325	-1.284		-.958	-.728
	.1000	-1.048	-1.069		-1.039	-.639
	.2250	-.687	-.639		-.630	-.457
	.4500	-.445	-.454		-.528	-.352
	.7500	-.299	-.531		-.284	-.394
	.8350	-.060	-.039			
	.8750	-.066	-.048			
	.9250	-.104	.006			
	.9750	-.087	.093			
Lower	.0500	-.101	.048	Lower	-.203	-.436
	.1500	-.158	-.176		-.307	-.394
	.4000	-.096	-.269		-.281	-.254
	.7000	-.042	-.054		-.036	-.048
	.8530	-.087	-.000			
	.9250	-.122	-.033			
	.9750	-.018	.006			
Aileron						
Upper	.8050	-.287	-.376	Upper	-.376	-.191
	.8150	-.269	-.364		-.364	-.191
	.8350	-.173	-.325		-.325	-.349
	.8750	-.101	-.248		-.248	-.209
	.9250	-.036	-.078		-.078	-.066
	.9750	.015	.042		.042	.012
Lower	.8100	.051	.176	Lower	.176	.155
	.8340	.054	.299		.299	.054
	.9250	-.039	.063		.063	.084
	.9750	.012	-.024		-.024	.027

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^{\circ}$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(c) \quad \alpha_f = -2.8^0$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						Wing
Upper				.0000	.782	.925
				.0100	-.922	-.537
				.0250	-1.660	-1.337
				.0500	-1.555	-1.287
				.1000	-1.313	-.931
				.2250	-.681	-.621
				.4500	-.490	-.397
				.7500	-.197	-.415
Lower				.0500	.125	-.075
				.1500	-.033	-.194
				.4000	-.149	-.176
				.7000	-.012	.072
						-.000
Aileron						
Upper				.8050	-.200	-.349
				.8150	-.191	-.361
				.8350	-.146	-.328
				.8750	-.018	-.248
				.9250	.000	-.069
				.9750	.060	.039
Lower				.8100	.119	.194
				.8340	.119	.319
				.9250	.015	.051
				.9750	.096	-.069

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(d) \quad \alpha_f = -1.0^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:			C_p for values of y/b of:
		0.1621	0.2895		
Wing					
Upper	.0000		.236		.830
	.0100		-2.397		-.728
	.0250		-2.964		-1.158
	.0500		-2.657		-1.158
	.1000		-1.704		-1.949
	.2250		-.931		-.651
	.4500		-.603		-.382
	.7500		-.316		-.322
	.8350		.030		
	.8750		.030		
	.9250		.039		
	.9750		.054		
Lower	.0500		.525		.024
	.1500		.146		-.110
	.4000		.081		-.116
	.7000		.057		
	.8530		.009		
	.9250		-.033		
	.9750		-.009		
Aileron					
Upper	.8050		-.290		-.239
	.8150		-.275		-.287
	.8350		-.179		-.448
	.8750		-.104		-.310
	.9250		-.057		-.167
	.9750		.027		-.078
Lower	.8100		.087		.140
	.8340		.021		.069
	.9250		-.075		-.030
	.9750		.009		-.075

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(e) \quad \alpha_f = -0.1^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-.257	-.039	Upper	.0000	.018	.564	.728
	.0100	-2.722	-2.958		.0100	-2.481	-1.693	-1.042
	.0250	-3.006	-3.424		.0250	-2.976	-2.501	-1.442
	.0500	-2.666	-3.015		.0500	-2.564	-2.284	-1.394
	.1000	-1.734	-1.803		.1000	-1.881	-1.433	-1.060
	.2250	-1.107	-.976		.2250	-.964	-.827	-.696
	.4500	-.582	-.627		.4500	-.645	-.490	-.397
	.7500	-.281	-.406		.7500	-.242	-.373	-.361
	.8350	-.057	-.066					
	.8750	-.033	-.036					
	.9250	.006	.009					
	.9750	.024	.081					
Lower	.0500	.573	.701					
	.1500	.290	.218					
	.4000	.045	.078					
	.7000	.048	.060					
	.8530	-.066	.012					
	.9250	-.152	-.006					
	.9750	-.042	-.018					
Aileron								
Upper	.8050	-.206	-.328	Upper	.8050	-.206	-.328	-.290
	.8150	-.212	-.322		.8150	-.212	-.322	-.313
	.8350	-.161	-.284		.8350	-.161	-.284	-.475
	.8750	-.075	-.176		.8750	-.075	-.176	-.367
	.9250	-.012	-.066		.9250	-.012	-.066	-.212
	.9750	.024	.006		.9750	.024	.006	-.093
Lower	.8100	.134	.200	Lower	.8100	.134	.200	.119
	.8340	.107	.322		.8340	.107	.322	.045
	.9250	-.012	.048		.9250	-.012	.048	-.057
	.9750	-.006	-.084		.9750	-.006	-.084	-.131

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(f) $\alpha_f = 0.4^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-.472	-.355	Upper	.0000	-2.866	.394	.603
	.0100	-3.096	-3.531		.0100	-3.310	-1.991	-1.203
	.0250	-3.296	-3.412		.0250	-2.833	-2.728	-1.558
	.0500	-2.878	-2.006		.0500	-1.973	-2.484	-1.493
	.1000	-1.767	-1.075		.1000	-1.036	-1.504	-1.099
	.2250	-1.110	-.690		.2250	-.657	-.881	-.710
	.4500	-.603	-.433		.4500	-.248	-.510	-.394
	.7500	-.245	.690		.7500	.606	-.364	-.388
	.8350	-.084	-.033	Lower	.0500	.236	.436	.200
	.8750	-.069	.006		.1500	-.021	.119	-.006
	.9250	.006	.045		.4000	.069	-.048	-.087
	.9750	.015	-.003		.7000	-.218	.084	-.033
Lower	.0500	.367	.275	Aileron				
	.1500	.284	.084	Upper	.8050	-.215	-.322	-.287
	.4000	.143	.039		.8150	-.173	-.319	-.299
	.7000	.027	-.039		.8350	-.093	-.269	-.478
	.8530	-.075	-.021		.8750	-.039	-.179	-.388
	.9250	-.122	-.042		.9250	-.000	-.075	-.239
	.9750	-.063	.012		.9750	.122	.003	-.113
Lower	.8100		.099	Lower	.8100	.099	.173	.090
	.8340		-.024		.8340	-.024	.290	.042
	.9250		-.018		.9250	-.018	.042	-.048
	.9750		-.009		.9750	-.009	-.093	-.110

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 0.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
		Wing				
Upper	.0000		-.319			
	.0100		-3.609			
	.0250		-3.940			
	.0500		-3.388			
	.1000		-1.976			
	.2250		-1.063			
	.4500		-.687			
	.7500		-.385			
	.8350		-.030			
	.8750		.018			
Lower	.9250		.057			
	.9750		.042			
	.0500		.776			
	.1500		.358			
	.4000		.185			
	.7000		.116			
	.8530		.033			
	.9250		-.006			
	.9750		-.009			
Aileron						
Upper	.8050		-.251			
	.8150		-.203			
	.8350		-.149			
	.8750		-.119			
	.9250		-.042			
	.9750		.030			
Lower	.8100		.090			
	.8340		.036			
	.9250		-.042			
	.9750		-.045			

TABLE X..- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(h) \quad \alpha_f = 1.3^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000	-.907	-.731	
	.0100	-3.704	-4.230	
	.0250	-3.824	-4.436	
	.0500	-3.236	-3.749	
	.1000	-2.030	-2.125	
	.2250	-1.218	-1.146	
	.4500	-.639	-.704	
	.7500	-.284	-.346	
	.8350	-.057	-.090	
	.8750	-.042	-.060	
Lower	.9250	.018	-.003	
	.9750	.024	.048	
	.0500	-.084	.842	
	.1500	.409	.352	
	.4000	.128	.161	
	.7000	.096	.078	
	.8530	-.107	.042	
	.9250	-.158	-.018	
	.9750	-.072	-.060	
Aileron				
Upper	.8050	-.203	-.290	-.382
	.8150	-.203	-.293	-.331
	.8350	-.158	-.221	-.573
	.8750	-.104	-.128	-.463
	.9250	-.060	-.045	-.296
	.9750	-.030	-.027	-.143
Lower	.8100	.140	.242	.045
	.8340	.096	.367	.048
	.9750	-.045	.075	-.057
			-.051	-.143

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $a_{\infty} = 20.1 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(i) \quad \alpha_f = 1.8^0$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper	.0000	-1.125	-.967	Upper	.0000	-1.116	
	.0100	-4.033	-4.645		.0100	-4.149	
	.0250	-3.964	-4.716		.0250	-4.427	
	.0500	-3.310	-3.979		.0500	-3.669	
	.1000	-2.128	-2.245		.1000	-2.373	
	.2250	-1.203	-1.290		.2250	-1.170	
	.4500	-.558	-.696		.4500	-.672	
	.7500	-.179	-.301		.7500	-.185	
	.8350	.024	-.090				
	.8750	.027	.000				
	.9250	.018	.036				
	.9750	.042	.078				
Lower	.0500	.806	.910	Lower	.0500	.824	
	.1500	.499	.484		.1500	.412	
	.4000	.215	.209		.4000	.096	
	.7000	.122	.176		.7000	.137	
	.8530	.030	.027				
	.9250	-.101	.087				
	.9750	-.042	.000				
Aileron							
Upper	.8050	-.182	Upper	.8050	-.182	Upper	-.182
	.8150	-.149		.8150	-.149		-.149
	.8350	-.101		.8350	-.101		-.101
	.8750	-.060		.8750	-.060		-.060
	.9250	-.030		.9250	-.030		-.030
	.9750	-.003		.9750	-.003		-.003
Lower	.8100	.167	Lower	.8100	.167	Lower	.167
	.8340	.107		.8340	.107		.107
	.9250	-.000		.9250	-.000		-.000
	.9750	-.009		.9750	-.009		-.009

TABLE XI

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(a) \alpha_f = -8.5^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper				.945 .255 -.455 -.491 -.497 -.430 -.321 -.333		.939 .345 -.188 -.248 -.418 -.406 -.297 -.230
Lower				.0500 .1500 .4000 .7000		-.624 -.485 -.321 -.006
Aileron						
Upper				.8050 .8150 .8350 .8750 .9250 .9750		-.370 -.309 -.273 -.236 -.085 .024
Lower				.8100 .8340 .9250 .9750		.091 .212 .042 .067
						.103 .079 .042 .048

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(b) \quad \alpha_f = -4.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000	.636		
	.0100	-1.170		
	.0250	-1.752		
	.0500	-1.752		
	.1000	-1.364		
	.2250	-0.800		
	.4500	-0.661		
	.7500	-0.309		
	.8350	-0.127		
	.8750	-0.115		
	.9250	-0.139		
	.9750	-0.109		
Lower	.0500	.158		
	.1500	-0.067		
	.4000	-0.133		
	.7000	-0.164		
	.8530	-0.079		
	.9250	-0.085		
	.9750	-0.061		
Aileron				
Upper	.8050	-0.206	-0.321	-0.176
	.8150	-0.182	-0.291	-0.200
	.8350	-0.115	-0.273	-0.339
	.8750	-0.170	-0.218	-0.242
	.9250	-0.042	-0.085	-0.115
	.9750	0.067	0.061	-0.024
Lower	.8100	0.067	0.109	0.109
	.8340	0.085	0.255	0.091
	.9250	-0.067	0.055	-0.030
	.9750	0.030	0.085	-0.012

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(c) \quad \alpha_f = -1.0^0$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		.085			
	.0100		-2.339			
	.0250		-3.127			
	.0500		-2.430			
	.1000		-1.588			
	.2250		-.867			
	.4500		-.503			
	.7500		-.333			
	.8350		.012			
	.8750		.006			
	.9250		.079			
	.9750		.115			
Lower	.0500		.564			
	.1500		-.006			
	.4000		.067			
	.7000		.109			
	.8530		.091			
	.9250		-.018			
	.9750		.055			
Aileron						
Upper	.8050		-.188			
	.8150		-.139			
	.8350		-.079			
	.8750		-.042			
	.9250		.030			
	.9750		-.024			
Lower	.8100		.200			
	.8340		.194			
	.9250		-.006			
	.9750		-.018			

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(d) \quad \alpha_f = 4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper	.0000	-1.624	-1.648	Upper	.0000	-1.988	-.927
	.0100	-4.545	-5.145		.0100	-4.988	-4.170
	.0250	-4.418	-5.188		.0250	-5.097	-4.467
	.0500	-3.788	-4.333		.0500	-4.200	-4.188
	.1000	-2.230	-2.430		.1000	-2.485	-2.121
	.2250	-1.406	-1.224		.2250	-1.248	-1.158
	.4500	-.642	-.679		.4500	-.630	-.576
	.7500	-.242	-.206		.7500	-.164	-.309
	.8350	-.048	-.085				
	.8750	-.018	-.079				
	.9250	-.024	-.061				
	.9750	-.012	-.012				
Lower	.0500	.879	.927	Lower	.0500	.891	.800
	.1500	.594	.545		.1500	.539	
	.4000	.194	.248		.4000	.200	
	.7000	.127	.097		.7000	.145	
	.8530	-.012	-.073				
	.9250	-.127	-.012				
	.9750	-.042	-.061				
Aileron							
Upper	.8050		-.121	Upper	.8050	-.121	
	.8150		-.133		.8150	-.133	
	.8350		-.103		.8350	-.103	
	.8750		-.048		.8750	-.048	
	.9250		-.024		.9250	-.024	
	.9750		-.012		.9750	-.012	
Lower	.8100		.139	Lower	.8100	.139	
	.8340		.236		.8340	.236	
	.9250		-.012		.9250	-.012	
	.9750		.030		.9750	.030	

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(e) \quad \alpha_f = 6.9^0$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:									
		0.1621	0.2895								
Wing											
Upper	.0000			-1.733							
	.0100			-5.236							
	.0250			-5.339							
	.0500			-4.358							
	.1000			-2.370							
	.2250			-1.152							
	.4500			-0.515							
	.7500			-0.242							
	.8350			.006							
	.8750			-0.115							
Lower	.9250			-0.085							
	.9750			-0.085							
Upper	.0500			1.000							
	.1500			.497							
	.4000			.261							
	.7000			.152							
	.8530			-.091							
	.9250			-.103							
	.9750			-.152							
Lower	.8100			.115							
	.8340			.067							
	.9250			-.055							
	.9750			-.127							
Aileron											
Upper	.8050			-.224							
	.8150			-.309							
	.8350			-.085							
	.8750			-.073							
	.9250			.030							
	.9750			-.061							
Lower	.8100			.121							
	.8340			.285							
	.9250			.055							
	.9750			.048							

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(f) \quad \alpha_f = 7.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:							
		0.1621	0.2895						
Wing									
Upper	.0000	-1.509	-2.048	Upper	.0000	-2.776	-1.503		
	.0100	-4.212	-5.521		.0100	-5.788	-4.885		
	.0250	-4.285	-5.630		.0250	-5.800	-5.091		
	.0500	-2.921	-4.467		.0500	-4.709	-4.812		
	.1000	-1.988	-2.588		.1000	-2.661	-2.364		
	.2250	-.709	-1.267		.2250	-1.412	-1.176		
	.7500	-.212	-.339		.4500	-.703	-.642		
	.8350	-.515	-.121		.7500	-.212	-.303		
	.8750	-.479	-.188						
	.9250	-.479	-.230						
	.9750	-.515	-.242						
Lower	.0500	.642	.897	Lower	.0500	.818	.836		
	.1500	.139	.600		.1500	.533	.521		
	.4000	.212	.206		.4000	.176	.073		
	.7000	.115	.073		.7000	.085	.061		
	.8530	-.230	-.042						
	.9250	-.533	-.030						
	.9750	-.491	-.188						
Aileron									
Upper	.8050		-.273	Upper			-.309		
	.8150		-.139						
	.8350		-.176						
	.8750		-.176						
	.9250		-.158						
	.9750		-.103						
Lower	.8100		.024	Lower					
	.8340		.127						
	.9250		-.097						
	.9750		-.024						

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(g) $\alpha_f = 8.8^0$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.952			
	.0100		-5.339			
	.0250		-5.309			
	.0500		-4.261			
	.1000		-2.303			
	.2250		-1.212			
	.4500		-.642			
	.7500		-.309			
	.8350		-.194			
	.8750		-.224			
	.9250		-.279			
	.9750		-.267			
Lower	.0500		.909			
	.1500		.648			
	.4000		.127			
	.7000		.036			
	.8530		-.042			
	.9250		-.103			
	.9750		-.097			
Aileron						
Upper	.8050		-.176			
	.8150		-.121			
	.8350		-.133			
	.8750		-.164			
	.9250		-.121			
	.9750		-.067			
Lower	.8100		.073			
	.8340		.152			
	.9250		-.091			
	.9750		-.061			

TABLE XI.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 9.9 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(h) \quad \alpha_f = 9.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-2.145			
	.0100		-5.564			
	.0250		-5.455			
	.0500		-4.485			
	.1000		-2.497			
	.2250		-1.073			
	.4500		-.667			
	.7500		-.206			
	.8350		-.188			
	.8750		-.315			
Lower	.9250		-.436			
	.9750		-.218			
	.0500		.861			
	.1500		-.564			
	.4000		-.121			
	.7000		-.073			
Aileron	.8530		-.085			
	.9250		-.133			
	.9750		-.200			
	.8050		-.164			
	.8150		-.170			
	.8350		-.133			
	.8750		-.121			
Upper	.9250		-.024			
	.9750		-.079			
	.8100		-.073			
	.8340		-.164			
Lower	.9250		-.109			
	.9750		-.012			
	.8100		.055			
	.8340		-.230			
Aileron	.9250		.000			
	.9750		-.000			

TABLE XII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(a) \alpha_f = -8.6^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000		.948	
	.0100		-.104	
	.0250		-.860	
	.0500		-.762	
	.1000		-.819	
	.2250		-.617	
	.4500		-.352	
	.7500		-.425	
	.8350		-.026	
	.8750		-.166	
	.9250		-.010	
	.9750		.073	
Lower	.0500		-.280	
	.1500		-.321	
	.4000		-.311	
	.7000		-.088	
	.8530		-.067	
	.9250		-.073	
			-.005	
Wing				
Upper	.0000		.938	
	.0100		.031	
	.0250		-.611	
	.0500		-.570	
	.1000		-.746	
	.2250		-.497	
	.4500		-.497	
	.7500		-.280	
Lower	.0500		-.534	
	.1500		-.368	
	.4000		-.290	
	.7000		-.031	
Aileron				
Upper	.8050		-.347	
	.8150		-.259	
	.8350		-.124	
	.8750		-.047	
	.9250		.010	
	.9750		.016	
Lower	.8100		-.021	
	.8340		.073	
	.9250		.016	
	.9750		.104	
			.057	
			.197	
			.036	
			.088	
			.104	
			.078	
			.036	
			.062	

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000		.699	
	.0100		-1.150	
	.0250		-1.917	
	.0500		-1.756	
	.1000		-1.332	
	.2250		-0.772	
	.4500		-0.503	
	.7500		-0.430	
	.8350		-0.062	
	.8750		-0.057	
	.9250		-0.021	
	.9750		.104	
Lower	.0500		.171	
	.1500		-0.083	
	.4000		-0.119	
	.7000		-0.067	
	.8530		.041	
	.9250		.010	
	.9750		-0.005	
Aileron				
Upper	.8050		-0.249	-0.363
	.8150		-0.249	-0.326
	.8350		-0.171	-0.301
	.8750		-0.109	-0.233
	.9250		.010	-0.083
	.9750		.026	.031
Lower	.8100		.078	.098
	.8340		.078	.254
	.9250		-0.021	.036
	.9750		.041	.062

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(c) \quad \alpha_f = -2.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		.285			
	.0100		-2.093			
	.0250		-2.803			
	.0500		-2.368			
	.1000		-1.570			
	.2250		-.886			
	.4500		-.513			
	.7500		-.233			
	.8350		-.093			
	.8750		-.016			
	.9250		.036			
	.9750		.088			
Lower	.0500		.601			
	.1500		.207			
	.4000		.104			
	.7000		.041			
	.8530		-.036			
	.9250		-.005			
	.9750		-.005			
Aileron						
Upper	.8050		-.181			
	.8150		-.192			
	.8350		-.150			
	.8750		-.098			
	.9250		-.036			
	.9750		.062			
Lower	.8100		.062			
	.8340		.057			
	.9250		.057			
	.9750		.057			
				.083	.093	
				.306	.067	
				.047	-.078	
				.047	-.052	

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(d) \quad \alpha_f = -1.0^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:
		0.1621	0.2895	
Wing				
Upper	.0000	-.171	-.109	
	.0100	-2.736	-3.098	
	.0250	-3.109	-3.642	
	.0500	-2.720	-3.223	
	.1000	-1.736	-1.938	
	.2250	-1.176	-1.036	
	.4500	-.663	-.741	
	.7500	-.301	-.440	
	.8350	-.088	-.171	
	.8750	-.073	-.031	
	.9250	-.078	-.031	
	.9750	-.161	.098	
Lower	.0500	.446	.684	
	.1500	.264	.228	
	.4000	.171	.078	
	.7000	.041	.067	
	.8530	-.073	.041	
	.9250	-.114	-.047	
	.9750	-.062	-.062	
Aileron				
Upper	.8050	-.207	-.264	-.264
	.8150	-.187	-.238	-.269
	.8350	-.135	-.228	-.399
	.8750	-.073	-.176	-.332
	.9250	-.005	-.057	-.207
	.9750	.026	.047	-.098
Lower	.8100	.114	.114	.062
	.8340	.181	.280	.052
	.9250	-.021	.041	-.130
	.9750	.041	.057	-.093

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(e) \quad \alpha_f = -0.1^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.580			
	.0100		-3.674			
	.0250		-4.155			
	.0500		-3.637			
	.1000		-2.135			
	.2250		-1.223			
	.4500		-.819			
	.7500		-.549			
	.8350		-.228			
	.8750		-.233			
	.9250		-.135			
	.9750		-.098			
Lower	.0500		.627			
	.1500		.181			
	.4000		.047			
	.7000		-.041			
	.8530		-.088			
	.9250		-.114			
	.9750		-.114			
Aileron						
Upper	.8050		-.187			
	.8150		-.223			
	.8350		-.161			
	.8750		-.073			
	.9250		-.036			
	.9750		.041			
Lower	.8100		.130			
	.8340		.119			
	.9250		-.021			
	.9750		.067			

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $a_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened

$$(f) \quad \alpha_f = 0.8^0$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-.575	-.658	.0000	-.839	.233
	.0100	-3.342	-3.933	.0100	-3.668	-1.979
	.0250	-3.617	-4.218	.0250	-4.010	-2.166
	.0500	-3.259	-3.648	.0500	-3.394	-2.000
	.1000	-1.984	-2.176	.1000	-2.161	-1.316
	.2250	-1.259	-1.124	.2250	-1.155	-1.021
	.4500	-.699	-.653	.4500	-.689	-.565
	.7500	-.383	-.244	.7500	-.259	-.311
	.8350	-.041	-.078			
	.8750	-.041	-.016			
Lower	.9250	-.047	-.031			
	.9750	-.016	.062			
	.0500	.731	.876	.0500	.793	.461
	.1500	.451	.399	.1500	.394	.104
	.4000	.223	.135	.4000	.104	-.057
	.7000	.067	.119	.7000	.088	-.135
Aileron						
Upper	.8050		-.181	.8050	-.238	-.295
	.8150		-.171	.8150	-.218	-.285
	.8350		-.140	.8350	-.202	-.415
	.8750		-.083	.8750	-.150	-.394
	.9250		-.041	.9250	-.041	-.249
	.9750		-.010	.9750	.010	-.124
Lower	.8100		.078	.8100	.104	.016
	.8340		.150	.8340	.311	.031
	.9250		-.041	.9250	.041	-.166
	.9750		-.021	.9750	.041	-.119

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(g) \quad \alpha_f = 1.3^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.031	-1.202	Upper	.0000	-1.363	-.383	-.067
	.0100	-4.031	-4.699		.0100	-4.352	-3.435	-2.451
	.0250	-4.155	-4.829		.0250	-4.575	-4.047	-2.544
	.0500	-3.715	-4.109		.0500	-3.907	-3.710	-2.311
	.1000	-2.389	-2.358		.1000	-2.363	-2.021	-1.477
	.2250	-1.373	-1.259		.2250	-1.244	-1.093	-.876
	.4500	-.699	-.710		.4500	-.725	-.580	-.466
	.7500	-.295	-.192		.7500	-.187	-.275	-.389
	.8350	-.109	-.088	Wing				
	.8750	-.052	-.062	Aileron				
	.9250	-.067	.005	Aileron				
	.9750	-.031	.031	Upper	.8050	-.166	-.244	-.342
Lower	.0500	.850	.860		.8150	-.145	-.238	-.342
	.1500	.425	.508		.8350	-.114	-.228	-.394
	.4000	.212	.098		.8750	-.067	-.171	-.435
	.7000	.083	.062		.9250	-.031	-.067	-.342
	.8530	-.021	.067		.9750	.021	-.016	-.171
	.9250	-.145	.021	Lower	.8100	.098	.078	-.057
	.9750	-.104	-.047		.8340	.161	.275	.005
					.9250	-.021	-.021	-.233
					.9750	-.000	-.010	-.176

TABLE XII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 11.6 \text{ lb/sq ft}$; belly plate off;
 $p = 4.8 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(h) \quad \alpha_f = 1.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.373	-1.534	Wing	.0000	-2.161	-.876	-.342
	.0100	-4.580	-5.218		.0100	-5.212	-4.301	-2.860
	.0250	-4.420	-5.275		.0250	-5.358	-4.705	-2.834
	.0500	-3.902	-4.404		.0500	-4.316	-4.383	-2.528
	.1000	-2.404	-2.477		.1000	-2.601	-2.202	-1.539
	.2250	-1.352	-1.337		.2250	-1.337	-1.150	-.881
	.4500	-.689	-.627		.4500	-.674	-.549	-.435
	.7500	-.280	-.166		.7500	-.181	-.212	-.332
	.8350	-.093	-.130					
	.8750	-.057	-.093					
	.9250	-.031	-.031					
	.9750	-.010	-.031					
Lower	.0500	.741	.896	Aileron	.0500	.886	.855	.637
	.1500	.596	.565		.1500	.549	.482	.228
	.4000	.218	.228		.4000	.187	.135	-.021
	.7000	.166	.057		.7000	.062	.073	-.202
	.8530	.016	.083					
	.9250	-.166	.010					
	.9750	-.135	-.031					
Upper	.8050	-.166	-.145	Aileron	.8050	-.166	-.145	-.259
	.8150	-.155	-.114		.8150	-.155	-.114	-.264
	.8350	-.114	-.104		.8350	-.114	-.104	-.358
	.8750	-.062	-.073		.8750	-.062	-.073	-.420
	.9250	-.005	-.021		.9250	-.005	-.021	-.342
	.9750	.016	.021		.9750	.016	.021	-.218
Lower	.8100	.052	.078	Aileron	.8100	.052	.078	-.083
	.8340	.114	.269		.8340	.114	.269	-.036
	.9250	-.052	.005		.9250	-.052	.005	-.228
	.9750	-.026	.021		.9750	-.026	.021	-.161

TABLE XIII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^0$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate off;
 $p = 2.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(a) \alpha_f = -8.6^0$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		.868			
	.0100		-1.000			
	.0250		-1.912			
	.0500		-1.807			
	.1000		-1.368			
	.2250		-0.877			
	.4500		-0.526			
	.7500		-0.175			
	.8350		-0.088			
	.8750		-0.123			
	.9250		-0.123			
	.9750		-0.044			
Lower	.0500		.132			
	.1500		-0.246			
	.4000		-0.193			
	.7000		-0.114			
	.8530		-0.053			
	.9250		-0.053			
	.9750		-0.053			
Aileron						
Upper	.8050		-0.228			
	.8150		-0.123			
	.8350		-0.079			
	.8750		-0.035			
	.9250		-0.000			
	.9750		-0.123			
Lower	.8100		-0.289			
	.8340		-0.149			
	.9250		0.009			
	.9750		-0.018			

TABLE XIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate off;
 $p = 2.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000		-.009	
	.0100		-2.298	
	.0250		-3.044	
	.0500		-2.640	
	.1000		-1.860	
	.2250		-.912	
	.4500		-.702	
	.7500		-.518	
	.8350		-.158	
	.8750		-.132	
	.9250		-.035	
	.9750		.009	
Lower	.0500		.395	
	.1500		.026	
	.4000		-.114	
	.7000		-.105	
	.8530		-.167	
	.9250		-.096	
	.9750		-.114	
Wing				
Upper	.0000		.123	
	.0100		-2.202	
	.0250		-2.789	
	.0500		-2.395	
	.1000		-1.658	
	.2250		-1.044	
	.4500		-.649	
	.7500		-.219	
Lower	.0500		.465	
	.1500		.009	
	.4000		-.184	
	.7000		-.184	
Aileron				
Upper	.8050		-.237	
	.8150		-.254	
	.8350		-.132	
	.8750		-.105	
	.9250		-.061	
	.9750		-.035	
Lower	.8100		-.237	
	.8340		-.123	
	.9250		-.026	
	.9750		-.009	

TABLE XIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate off;
 $p = 2.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(c) \quad \alpha_f = -3.0^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-.123	-.263	Upper	.0000	-.570	.061	.360
	.0100	-2.439	-3.026		.0100	-2.956	-2.535	-1.640
	.0250	-2.807	-3.570		.0250	-3.482	-3.149	-1.886
	.0500	-2.553	-3.175		.0500	-2.947	-2.868	-1.728
	.1000	-1.772	-1.781		.1000	-1.939	-1.623	-1.184
	.2250	-1.351	-.956		.2250	-1.061	-.851	-.746
	.4500	-.614	-.640		.4500	-.518	-.421	-.333
	.7500	-.123	-.456		.7500	-.149	-.079	-.053
	.8350	-.061	-.061					
	.8750	-.035	.026					
	.9250	-.132	.070					
	.9750	-.026	.026					
Lower	.0500	.377	.632	Lower	.0500	.675	.561	.351
	.1500	.211	.325		.1500	.237	.254	.070
	.4000	.149	.096		.4000	.009	.035	-.088
	.7000	.044	.053		.7000	-.018	-.114	-.228
	.8530	-.175	-.018					
	.9250	-.219	-.009					
	.9750	-.167	-.009					
Aileron								
Upper	.8050	-.026	-.009	.026				
	.8150	-.070	.018	.096				
	.8350	.044	.026	.053				
	.8750	.044	-.018	-.061				
	.9250	.053	.026	-.044				
	.9750	.088	.061	-.053				
Lower	.8100	-.193	-.140	.219				
	.8340	.018	.035	-.289				
	.9250	.018	-.061	-.184				
	.9750	.026	.053	-.096				

TABLE XIII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 6.9 \text{ lb/sq ft}$; belly plate off;
 $p = 2.0 \text{ lb/sq in.}$; forward guy cables, lightly tightened]

$$(d) \quad \alpha_f = -1.1^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing					Wing		
Upper				.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-.1.316 -.4.289 -.4.561 -.3.675 -.2.482 -.1.114 -.561 -.132	-.719 -3.991 -4.342 -4.070 -2.044 -1.009 -.465 -.061	-.333 -2.772 -2.711 -2.500 -1.465 -.825 -.377 -.061
				.0500 .1500 .4000 .7000	.895 .377 .228 .132	.833 .447 .140 -.070	.640 .219 -.035 -.254
Lower							
Aileron							
Upper				.8050 .8150 .8350 .8750 .9250 .9750	.018 -.088 -.044 .026 -.009 .061	.026 .035 .044 .053 .044 .114	.026 .009 .061 -.140 -.193 -.184
Lower				.8100 .8340 .9250 .9750	.000 -.088 -.044 .053	-.140 .088 -.044 .079	-.325 -.105 -.254 -.123

TABLE XIV

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(a) $\alpha_f = -4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.692	.769	Upper	.0000	.769
	.0100	-.889	-.855		.0100	-.838
	.0250	-1.556	-1.581		.0250	-1.513
	.0500	-1.487	-1.504		.0500	-1.308
	.1000	-1.393	-1.231		.1000	-1.197
	.2250	-.821	-.795		.2250	-.752
	.4500	-.479	-.427		.4500	-.538
	.7500	-.368	-.581		.7500	-.231
	.8350	-.171	-.077			
	.8750	-.043	-.060			
	.9250	-.043	-.085			
	.9750	-.094	.034			
Lower	.0500	-.043	.009	Lower	.0500	-.009
	.1500	-.291	-.214		.1500	-.214
	.4000	-.214	-.316		.4000	-.256
	.7000	-.103	-.137		.7000	-.128
	.8530	-.248	-.068			
	.9250	-.137	-.034			
	.9750	-.145	.017			
Aileron						
Upper	.8050	-.162	-.094	Upper	.8050	-.162
	.8150	-.154	-.085		.8150	-.154
	.8350	-.085	-.103		.8350	-.085
	.8750	-.085	-.094		.8750	-.085
	.9250	-.000	-.034		.9250	-.026
	.9750	-.009	.051		.9750	-.026
Lower	.8100	-.137	-.034	Lower	.8100	-.137
	.8340	-.026	.248		.8340	-.026
	.9250	.051	.017		.9250	-.043
	.9750	.043	.077		.9750	.043

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	.547	.402	Upper	.0000	.607	.915	.915
	.0100	-1.094	-1.222		.0100	-.940	-.632	-.376
	.0250	-1.641	-1.966		.0250	-1.641	-1.350	-.846
	.0500	-1.436	-1.761		.0500	-1.410	-1.316	-.829
	.1000	-1.154	-1.308		.1000	-1.214	-.863	-.692
	.2250	-.769	-.821		.2250	-.675	-.547	-.470
	.4500	-.530	-.556		.4500	-.530	-.274	-.205
	.7500	-.359	-.556		.7500	-.009	.145	.197
	.8350	-.188	-.188					
	.8750	-.197	-.103					
	.9250	-.239	-.103					
	.9750	-.103	-.026					
Lower	.0500	-.034	.068	Lower	.0500	-.000	-.026	-.120
	.1500	.034	-.308		.1500	-.214	-.179	-.248
	.4000	-.060	-.308		.4000	-.274	-.248	-.162
	.7000	-.085	-.325		.7000	-.291	-.299	-.222
	.8530	-.299	-.094					
	.9250	-.325	-.162					
	.9750	-.179	-.137					
Aileron								
Upper	.8050	-.282	.145	Upper	.8050	-.282	.145	.248
	.8150	.017	.179		.8150	.017	.179	.282
	.8350	.162	.197		.8350	.162	.197	.316
	.8750	.094	.111		.8750	.094	.111	.162
	.9250	-.034	.068		.9250	-.034	.068	.094
	.9750	-.171	-.017		.9750	-.171	-.017	.051
Lower	.8100	-.470	-.410	Lower	.8100	-.470	-.410	-.385
	.8340	-.162	-.222		.8340	-.162	-.222	-.274
	.9250	-.171	-.103		.9250	-.171	-.103	-.085
	.9750	-.077	-.034		.9750	-.077	-.034	.043

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(c) \quad \alpha_f = -4.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	.581	.667	Upper	.0000	.735	.932	.906
	.0100	-.966	-.829		.0100	-.761	-.444	-.137
	.0250	-1.709	-1.692		.0250	-1.444	-1.154	-.675
	.0500	-1.547	-1.513		.0500	-1.342	-1.137	-.709
	.1000	-1.154	-1.248		.1000	-1.179	-.855	-.658
	.2250	-.692	-.863		.2250	-.769	-.598	-.530
	.4500	-.504	-.556		.4500	-.615	-.427	-.342
	.7500	-.376	-.462		.7500	-.419	-.453	-.427
	.8350	-.154	-.179	Wing				
	.8750	-.179	-.026					
	.9250	-.034	-.051					
	.9750	-.291	-.017					
Lower	.0500	-.009	.154	Lower	.0500	.009	-.120	-.256
	.1500	-.197	-.214		.1500	-.179	-.248	-.248
	.4000	-.308	-.239		.4000	-.128	-.120	-.120
	.7000	-.308	-.068		.7000	.077	.154	.068
	.8530	-.282	-.051	Aileron				
	.9250	-.256	-.051					
	.9750	-.137	-.145					
Upper	.8050	-.427	-.479	.8050	-.427	-.419	-.427	
	.8150	-.376	-.299	.8150	-.376	-.291	-.462	
	.8350	-.299	-.214	.8350	-.299	-.188	-.256	
	.8750	-.214	-.162	.8750	-.214	-.154	-.188	
	.9250	-.162	-.111	.9250	-.162	-.171	-.188	
	.9750	-.111	-.043	.9750	-.111	-.034	-.060	
Lower	.8100	.162	.308	.8100	.162	.308	.274	
	.8340	.188	.308	.8340	.188	.308	.239	
	.9250	-.043	.111	.9250	-.043	.111	.085	
	.9750	-.043	-.034	.9750	-.043	-.034	-.060	

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(d) \quad \alpha_f = -0.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.333	.376	Upper	.239	.573
	.0100	-1.932	-2.120		-2.026	-1.470
	.0250	-2.308	-2.735		-2.410	-2.162
	.0500	-2.085	-2.299		-2.239	-1.983
	.1000	-1.436	-1.624		-1.581	-1.188
	.2250	-.991	-.889		-.897	-.709
	.4500	-.598	-.504		-.538	-.385
	.7500	-.325	-.359		-.239	-.111
	.8350	-.051	-.068			
	.8750	-.060	-.051			
	.9250	-.060	-.051			
	.9750	-.051	.103			
Lower	.0500	.359	.427	Lower	.376	.333
	.1500	.111	-.009		.085	.179
	.4000	-.026	-.103		-.077	-.043
	.7000	.068	-.154		-.103	-.103
	.8530	-.205	-.009			
	.9250	-.145	-.000			
	.9750	.017	-.017			
Aileron						
Upper	.8050	-.145	-.077	Upper	-.077	-.034
	.8150	-.154	-.077		-.077	-.043
	.8350	-.068	-.094		-.094	-.085
	.8750	-.060	-.077		-.077	-.128
	.9250	-.017	.000		.000	-.051
	.9750	-.017	.051		.051	-.009
Lower	.8100	-.103	-.009	Lower	-.009	-.043
	.8340	.000	.222		.222	-.000
	.9250	-.060	.000		.000	-.085
	.9750	.017	.103		.103	-.017

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(e) \alpha_f = -0.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.188	.111	Upper	.205	.444
	.0100	-2.026	-2.274		-2.581	-1.709
	.0250	-2.479	-2.795		-2.342	-2.316
	.0500	-2.103	-2.547		-1.658	-2.120
	.1000	-1.573	-1.564		-0.880	-1.248
	.2250	-1.068	-0.940		-0.470	-0.709
	.4500	-0.556	-0.462		-0.043	-0.291
	.7500	-0.248	-0.342		.453	.137
	.8350	-0.145	-0.043			
	.8750	-0.017	-0.017			
	.9250	.060	.000			
	.9750	.000	.085			
Lower	.0500	.291	.547	Lower	.085	.427
	.1500	.034	.111		-0.154	.103
	.4000	-0.154	-0.051		-0.171	-0.103
	.7000	-0.043	-0.051		-0.248	-0.282
	.8530	-0.000	-0.026			
	.9250	-0.137	-0.017			
	.9750	-0.068	.000			
Aileron						
Upper	.8050	-0.017	.154	Upper	.214	.205
	.8150	.077	.188		.239	.188
	.8350	.043	.197		.111	.034
	.8750	-0.017	.137		.051	
	.9250	-0.085	.094			
	.9750	-0.291				
Lower	.8100	-0.120	-0.419	Lower	-0.393	-0.282
	.8340	-0.085	-0.205		-0.094	-0.017
	.9250	-0.034	-0.060			
	.9750	-0.009	-0.009			

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^{\circ}$; cable configuration 2-5-8; $q_{\infty} = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

(f) $\alpha_f = -1.0^{\circ}$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						Wing
Upper	.0000	.291	.051	.128	.530	.709
	.0100	-1.880	-2.145	-2.120	-1.547	-.966
	.0250	-2.368	-2.786	-2.487	-2.265	-1.385
	.0500	-2.077	-2.444	-2.359	-2.043	-1.325
	.1000	-1.504	-1.607	-1.615	-1.308	-.966
	.2250	-.957	-.991	-1.000	-.769	-.675
	.4500	-.513	-.624	-.675	-.453	-.376
	.7500	-.274	-.427	-.436	-.385	-.487
	.8350	-.000	-.034			
	.8750	-.085	.009			
Lower	.9250	-.026	.009			
	.9750	-.034	.017			
Upper	.0500	.274	.496	.427	.342	.179
	.1500	.154	.051	.068	.051	-.043
	.4000	-.085	-.026	.017	-.034	-.077
	.7000	-.017	-.034	.120	.222	.043
	.8530	.000	-.026			
	.9250	-.103	.026			
	.9750	-.068	-.009			
Aileron						
Upper	.8050	-.402	-.350	-.453		
	.8150	-.350	-.333	-.436		
	.8350	-.333	-.231	-.444		
	.8750	-.274	-.162	-.299		
	.9250	-.239	-.137	-.214		
	.9750	-.137	-.137	-.239		
Lower	.8100	.205	.385	.188		
	.8340	.231	.333	.205		
	.9250	-.034	.145	-.026		
	.9750	-.034	.017	-.154		

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 2.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:				C_p for values of $y/b/2$ of:			
		0.1621	0.2895			0.5000	0.6800	0.8920	
Wing									
Upper	.0000		-.632			.0000	-.718	-.051	.256
	.0100		-3.880			.0100	-3.410	-2.692	-1.838
	.0250		-4.068			.0250	-3.812	-3.231	-2.034
	.0500		-3.556			.0500	-3.282	-2.906	-1.846
	.1000		-2.205			.1000	-2.077	-1.632	-1.248
	.2250		-1.239			.2250	-1.231	-.923	-.735
	.4500		-.667			.4500	-.667	-.453	-.359
	.7500		-.436			.7500	-.171	-.145	-.179
	.8350		-.034						
	.8750		.034						
	.9250		.111						
	.9750		.051						
Lower	.0500		.821			.0500	.624	.641	.419
	.1500		.564			.1500	.239	.239	.077
	.4000		.274			.4000	.060	.026	-.068
	.7000		.103			.7000	-.051	-.051	-.154
	.8530		-.017						
	.9250		-.068						
	.9750		-.026						
Aileron									
Upper	.8050		-.111			.8050	-.077	-.137	
	.8150		-.085			.8150	-.085	-.111	
	.8350		-.103			.8350	-.051	-.205	
	.8750		-.060			.8750	-.051	-.265	
	.9250		.000			.9250	-.017	-.205	
	.9750		.077			.9750	.043	-.094	
Lower	.8100		-.017			.8100	.026	-.111	
	.8340		-.009			.8340	.239	-.145	
	.9250		-.026			.9250	.034	-.137	
	.9750		.094			.9750	.077	-.111	

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(h) \quad \alpha_f = 2.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:			
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper							
Upper	.0000	-.778	-.735	Upper	-.838	-.248	
	.0100	-3.205	-3.632		-3.513	-2.991	
	.0250	-3.573	-4.017		-3.829	-3.419	
	.0500	-3.085	-3.615		-3.274	-3.060	
	.1000	-2.077	-1.983		-2.034	-1.641	
	.2250	-1.179	-1.026		-1.017	-.863	
	.4500	-.564	-.564		-.521	-.325	
	.7500	-.231	-.274		-.077	.103	
	.8350	-.060	-.085				
	.8750	-.145	-.009				
	.9250	-.171	.000				
	.9750	-.128	.043				
Lower							
Lower	.0500	.778	.829	Lower	.718	.513	
	.1500	.256	.385		.385	.137	
	.4000	.000	.145		.026	-.077	
	.7000	.085	.043		-.103	-.282	
	.8530	.009	.000				
	.9250	-.171	-.026				
	.9750	-.077	-.034				
Aileron							
Upper							
Upper	.8050	-.162	.120	Upper	.120	.162	
	.8150	-.009	.137		.137	.179	
	.8350	.034	.137		.060	.248	
	.8750	.060	.120		.094	.162	
	.9250	.060	.068		.068	.077	
	.9750	.034				-.026	
Lower							
Lower	.8100	-.256	-.479	Lower	-.479	-.462	
	.8340	-.034	-.162		-.162	-.188	
	.9250	-.034	-.077		-.077	-.145	
	.9750	.000	.017		.017	-.077	

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(i) \alpha_f = 2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.786			
	.0100		-3.684			
	.0250		-4.094			
	.0500		-3.521			
	.1000		-2.188			
	.2250		-1.470			
	.4500		-.838			
	.7500		-.658			
	.8350		-.197			
	.8750		-.188			
Lower	.9250		-.077			
	.9750		-.077			
	.0500		.564			
	.1500		.179			
	.4000		-.103			
	.7000		-.256			
	.8530		-.239			
Upper	.9250		-.051			
	.9750		.017			
Wing						
Upper	.0000		-.530			
	.0100		-3.350			
	.0250		-3.624			
	.0500		-3.128			
	.1000		-2.162			
	.2250		-1.197			
	.4500		-.624			
Lower	.7500		-.410			
	.0500		.709			
	.1500		.239			
	.4000		.043			
	.7000		.171			
Aileron						
Upper	.8050		-.350			
	.8150		-.265			
	.8350		-.265			
	.8750		-.205			
	.9250		-.128			
	.9750		-.051			
	.8100		.410			
Lower	.8340		.350			
	.9250		.137			
	.9750		.009			
	.8100		.385			
	.8340		.333			
	.9250		.154			
	.9750		.017			
	.8100		.154			

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(j) \quad \alpha_f = 7.2^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000		-1.915	
	.0100		-5.479	
	.0250		-5.299	
	.0500		-4.530	
	.1000		-2.650	
	.2250		-1.393	
	.4500		-.692	
	.7500		-.154	
	.8350		-.043	
	.8750		-.051	
	.9250		-.103	
	.9750		-.000	
Lower	.0500		.915	
	.1500		.615	
	.4000		.239	
	.7000		.051	
	.8530		.009	
	.9250		-.103	
	.9750		-.103	
Wing				
Upper	.0000		-2.103	
	.0100		-5.162	
	.0250		-5.248	
	.0500		-4.188	
	.1000		-2.504	
	.2250		-1.231	
	.4500		-.632	
Lower	.7500		-.120	
	.0500		.821	
	.1500		.496	
	.4000		.214	
Aileron	.7000		.026	
	.8100		-.026	
	.8340		-.026	
	.9250		-.145	
	.9750		-.034	
Aileron				
Upper	.8050		.068	
	.8150		.000	
	.8350		-.111	
	.8750		-.068	
	.9250		-.103	
	.9750		-.026	
Lower	.8100		.162	
	.8340		.436	
	.9250		.128	
	.9750		.154	

TABLE XIV., Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(k) \quad \alpha_f = 7.2^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-2.137	-1.932	Upper	.0000	-2.214	-1.316	-.692
	.0100	-5.060	-5.692		.0100	-5.376	-4.538	-3.222
	.0250	-4.855	-5.444		.0250	-5.419	-4.615	-3.060
	.0500	-4.103	-4.624		.0500	-4.470	-4.359	-2.701
	.1000	-2.342	-2.530		.1000	-2.641	-2.051	-1.496
	.2250	-1.299	-1.282		.2250	-1.299	-.983	-.752
	.4500	-.624	-.547		.4500	-.504	-.308	-.256
	.7500	-.188	-.145		.7500	-.145	.060	.137
	.8350	.017	-.103					
	.8750	.060	-.068					
	.9250	.034	.000					
	.9750	.034	.017					
Lower	.0500	.991	.983	Lower	.0500	.829	.923	.735
	.1500	.564	.632		.1500	.547	.547	.325
	.4000	.265	.316		.4000	.188	.171	-.009
	.7000	.256	.205		.7000	-.068	-.145	-.316
	.8530	.077	.034					
	.9250	-.085	.017					
	.9750	.103	-.085					
Aileron								
Upper	.8050	-.188	-.000	.8050	-.188	-.000	.231	
	.8150	-.085	.051	.8150	-.085	.051	.179	
	.8350	-.060	.051	.8350	-.060	.051	.402	
	.8750	.000	.068	.8750	.000	.068	.171	
	.9250	.017	.103	.9250	.017	.103	.051	
	.9750	-.017	.120	.9750	-.017	.120	-.068	
Lower	.8100	-.222	-.410	.8100	-.222	-.410	-.427	
	.8340	-.017	-.094	.8340	-.017	-.094	-.197	
	.9250	-.017	-.034	.9250	-.017	-.034	-.231	
	.9750	.017	.000	.9750	.017	.000	-.085	

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(l) \quad \alpha_f = 7.1^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-2.017	-1.675	Upper	-1.932	-.966
	.0100	-4.795	-5.137		-4.940	-4.171
	.0250	-4.658	-5.265		-5.068	-4.479
	.0500	-4.000	-4.615		-4.274	-4.162
	.1000	-2.316	-2.521		-2.504	-2.137
	.2250	-1.530	-1.402		-1.325	-1.162
	.4500	-.581	-.718		-.726	-.615
	.7500	-.316	-.333		-.325	-.402
	.8350	-.094	-.137			
	.8750	-.094	-.077			
	.9250	-.068	-.051			
	.9750	-.068	-.017			
Lower	.0500	1.017	.923	Lower	.812	.803
	.1500	.598	.513		.504	.479
	.4000	.188	.197		.239	.188
	.7000	.145	.085		.256	.299
	.8530	.094	.060			
	.9250	-.111	.000			
	.9750	-.017	-.068			
Aileron						
Upper	.8050	-.291	-.316	Upper	-.291	-.316
	.8150	-.214	-.274		-.214	-.274
	.8350	-.214	-.197		-.214	-.197
	.8750	-.171	-.171		-.171	-.171
	.9250	-.128	-.145		-.128	-.145
	.9750	-.111	-.171		-.111	-.171
Lower	.8100	.342	.376	Lower	.342	.376
	.8340	.402	.316		.402	.316
	.9250	.103	.154		.103	.154
	.9750	.026	.043		.026	.043

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(m) \quad \alpha_f = 10.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-1.769	-2.282	Upper	.0000	-2.769	-1.598	-.872
	.0100	-4.316	-5.547		.0100	-5.872	-4.915	-3.462
	.0250	-4.085	-5.376		.0250	-5.675	-4.957	-3.248
	.0500	-3.393	-4.496		.0500	-4.641	-4.769	-2.838
	.1000	-1.974	-2.513		.1000	-2.650	-2.239	-1.598
	.2250	-.966	-1.274		.2250	-1.350	-1.137	-.846
	.4500	-.709	-.675		.4500	-.658	-.462	-.402
	.7500	-.538	-.359		.7500	-.188	-.111	-.265
	.8350	-.462	-.248					
	.8750	-.470	-.299					
Lower	.9250	-.470	-.214					
	.9750	-.470	-.145					
Aileron								
Upper	.0500	.932	.897	Upper	.8050	-.145	-.068	-.197
	.1500	.521	.590		.8150	-.154	-.060	-.154
	.4000	.094	.299		.8350	-.145	-.060	-.205
	.7000	-.077	.077		.8750	-.094	-.043	-.368
	.8530	-.103	-.085		.9250	-.009	-.017	-.316
	.9250	-.513	-.154		.9750	-.000	-.026	-.248
Lower	.9750	-.487	-.214					

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(n) \quad \alpha_f = 10.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-1.530	-1.342	Upper	.0000	-.692
	.0100	-3.949	-4.333		.0100	-2.265
	.0250	-3.838	-4.265		.0250	-1.880
	.0500	-3.222	-3.778		.0500	-1.154
	.1000	-1.855	-1.983		.1000	-.692
	.2250	-.769	-1.026		.2250	-.650
	.4500	-.564	-.641		.4500	-.667
	.7500	-.470	-.632		.7500	-.632
	.8350	-.368	-.410			
	.8750	-.214	-.402			
	.9250	-.368	-.385			
	.9750	-.239	-.333			
Lower	.0500	.778	.855	Lower	.0500	.744
	.1500	.547	.385		.1500	.393
	.4000	.231	.111		.4000	.026
	.7000	.060	-.034		.7000	-.299
	.8530	.026	-.103			
	.9250	-.282	-.231			
	.9750	-.137	-.350			
Aileron						
Upper	.8050	-.641	-.530	Upper	.8100	-.359
	.8150	-.521	-.521		.8340	-.419
	.8350	-.521	-.590		.9250	-.513
	.8750	-.521	-.530		.9750	-.487
	.9250	-.521	-.470			
	.9750	-.487	-.376			
Lower	.8100	-.316	-.556	Lower	.8340	-.333
	.8340	-.333	-.368		.9250	-.376
	.9250	-.291	-.265		.9750	-.179
	.9750	-.291	-.179			

TABLE XIV.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 7.0 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(o) \quad \alpha_f = 10.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-2.077			
	.0100		-5.350			
	.0250		-5.410			
	.0500		-4.359			
	.1000		-2.222			
	.2250		-1.094			
	.4500		-.667			
	.7500		-.444			
	.8350		-.197			
	.8750		-.274			
	.9250		-.222			
	.9750		-.137			
Lower	.0500		.872			
	.1500		.658			
	.4000		.325			
	.7000		.145			
	.8530		-.060			
	.9250		-.103			
	.9750		-.111			
Wing						
Upper	.0000		-2.487			
	.0100		-5.598			
	.0250		-5.658			
	.0500		-4.624			
	.1000		-2.692			
	.2250		-1.308			
	.4500		-.692			
Lower	.7500		-.368			
	.0500		.855			
	.1500		.556			
	.4000		.256			
Aileron	.7000		.256			
	.8050		-.171			
	.8150		-.256			
	.8350		-.231			
	.8750		-.222			
	.9250		-.188			
Lower	.9750		-.222			
	.8100		.350			
	.8340		.368			
	.9250		.009			
	.9750		-.068			
Upper						
	.333					
	.222					
	.120					
	-.026					
Lower						
	-.077					
	-.068					
	-.179					
	-.274					

TABLE XV

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(a) \quad \alpha_f = -4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.669	-.947			
	.0100	-1.071	-1.680			
	.0250	-1.586	-1.645			
	.0500	-1.467	-1.195			
	.1000	-.994	-.852			
	.2250	-.746	-.509			
	.4500	-.396	-.355			
	.7500	-.237	-.178			
	.8350	-.077	-.112			
	.8750	-.107	-.107			
	.9250	-.077	.089			
	.9750	-.024	-.012			
Lower	.0500	-.000	-.142			
	.1500	-.089	-.183			
	.4000	-.095	-.118			
	.7000	-.041	.012			
	.8530	-.118	.024			
	.9250	-.077	.036			
	.9750	.053	.065			
Aileron						
Upper	.8050	-.095	-.083	-.006		
	.8150	-.012	-.083			
	.8350	-.024	-.089	-.107		
	.8750	.012	-.112	-.107		
	.9250	.036	-.047	-.041		
	.9750	-.095	-.012	-.000		
Lower	.8100	.024	-.059	-.059		
	.8340	.000	.142	-.041		
	.9250	.077	-.018	-.047		
	.9750	.012	.053	.018		

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.521	.598	Upper	.0000	.604
	.0100	-1.266	-1.320		.0100	-.870
	.0250	-1.817	-1.935		.0250	-1.592
	.0500	-1.627	-1.787		.0500	-1.485
	.1000	-1.207	-1.284		.1000	-.964
	.2250	-.811	-.828		.2250	-.586
	.4500	-.527	-.456		.4500	-.266
	.7500	-.302	-.396		.7500	.142
	.8350	-.036	-.101			
	.8750	-.065	-.047			
	.9250	-.047	-.053			
	.9750	-.077	.006			
Lower	.0500	.041	.183	Lower	.0500	.077
	.1500	-.077	-.160		.1500	-.101
	.4000	-.160	-.195		.4000	-.207
	.7000	-.095	-.136		.7000	-.302
	.8530	-.118	-.065			
	.9250	-.095	-.036			
	.9750	-.065	-.018			
Aileron						
Upper	.8050	-.201	.136	Upper	.8050	.201
	.8150	-.036	.207		.8150	.243
	.8350	.136	.183		.8350	.254
	.8750	.053	.107		.8750	.160
	.9250	.006	.053		.9250	.107
	.9750	-.006	.030		.9750	.041
Lower	.8100	-.254	-.414	Lower	.8100	-.385
	.8340	-.065	-.207		.8340	-.278
	.9250	-.041	-.077		.9250	-.089
	.9750	-.036	-.024		.9750	-.012

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(c) \quad \alpha_f = -4.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.645	.734	Upper	.0000	.899
	.0100	-.976	-.888		.0100	-.627
	.0250	-1.527	-1.680		.0250	-1.331
	.0500	-1.491	-1.485		.0500	-1.231
	.1000	-1.041	-1.059		.1000	-1.077
	.2250	-.769	-.751		.2250	-.710
	.4500	-.426	-.473		.4500	-.556
	.7500	-.201	-.485		.7500	-.361
	.8350	.018	-.024			
	.8750	.012	-.012			
	.9250	-.107	.000			
	.9750	-.036	.101			
Lower	.0500	.065	.189	Lower	.0500	.000
	.1500	-.166	-.124		.1500	-.178
	.4000	-.237	-.201		.4000	-.118
	.7000	-.059	-.047		.7000	.095
	.8530	-.095	-.012			
	.9250	-.101	.024			
	.9750	-.030	.018			
Aileron						
Upper	.8050	-.391	-.491	Upper	.8050	-.391
	.8150	-.355	-.432		.8150	-.355
	.8350	-.272	-.302		.8350	-.272
	.8750	-.189	-.195		.8750	-.189
	.9250	-.107	-.136		.9250	-.107
	.9750	-.095	-.112		.9750	-.095
Lower	.8100	.183	.320	Lower	.8100	.183
	.8340	.237	.314		.8340	.237
	.9250	-.036	.112		.9250	-.036
	.9750	-.012	-.000		.9750	-.012

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^{\circ}$; cable configuration 2-5-8; $q_{\infty} = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

$$(d) \quad \alpha_f = -0.9^{\circ}$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
		Wing				
Upper	.0000	.065	.172	Upper	.0000	.195
	.0100	-2.213	-2.260		.0100	-2.036
	.0250	-2.609	-2.905		.0250	-2.550
	.0500	-2.337	-2.521		.0500	-2.284
	.1000	-1.550	-1.651		.1000	-1.669
	.2250	-1.065	-0.970		.2250	-0.947
	.4500	-.527	-.651		.4500	-.580
	.7500	-.290	-.438		.7500	-.172
	.8350	-.160	-.107			
	.8750	-.065	-.083			
Lower	.9250	-.118	-.024	Lower	.0500	.450
	.9750	.030	.018		.1500	.118
					.4000	-.059
					.7000	-.065
Aileron						
Upper	.0500	.373	.521	Upper	.8050	-.101
	.1500	.107	.112		.8150	-.107
	.4000	-.036	-.101		.8350	-.036
	.7000	.036	-.006		.8750	-.006
	.8530	-.095	-.012		.9250	.024
	.9250	-.207	-.012		.9750	-.030
	.9750	-.148	-.012			
Lower				Lower	.8100	-.047
					.8340	-.012
					.9250	-.012
					.9750	.059

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(e) \alpha_f = -0.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		.006			
	.0100		-2.651			
	.0250		-3.089			
	.0500		-2.769			
	.1000		-1.757			
	.2250		-1.018			
	.4500		-0.580			
	.7500		-0.438			
	.8350		-0.047			
	.8750		.006			
	.9250		.024			
	.9750		-0.118			
Lower	.0500		.574			
	.1500		.142			
	.4000		-0.012			
	.7000		-0.118			
	.8530		-0.000			
	.9250		-0.036			
	.9750		-0.000			
Aileron						
Upper	.8050		.047			
	.8150		.107			
	.8350		.172			
	.8750		-0.053			
	.9250		-0.018			
	.9750		-0.142			
Lower	.8100		.006			
	.8340		.059			
	.9250		.041			
	.9750		.083			

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(f) \quad \alpha_f = -1.0^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.207	.272	Upper	.308	.598
	.0100	-2.018	-2.083		-1.858	-1.349
	.0250	-2.462	-2.763		-2.391	-2.047
	.0500	-2.166	-2.379		-2.142	-1.882
	.1000	-1.450	-1.568		-1.609	-1.231
	.2250	-888	-870		-888	-775
	.4500	-479	-586		-615	-485
	.7500	-213	-408		-325	-420
	.8350	-107	-112			
	.8750	-124	-041			
	.9250	-089	.012			
	.9750	.006	.059			
Lower	.0500	.426	.456	Lower	.438	.260
	.1500	.160	.107		.112	.024
	.4000	-107	-024		-0.012	.201
	.7000	-000	.018		.142	
	.8530	-142	-006			
	.9250	-148	-012			
	.9750	-101	-024			
Aileron						
Upper	.8050	-361	-396	Upper	-361	-467
	.8150	-296	-361		-296	-450
	.8350	-243	-249		-243	-456
	.8750	-195	-178		-195	-314
	.9250	-148	-142		-148	-225
	.9750	-095	-148		-095	-201
Lower	.8100	.219	.349	Lower	.219	.213
	.8340	.314	.331		.314	.225
	.9250	.018	.095		.018	-.059
	.9750	-.012	-.024		-.012	-.160

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(g) $\alpha_f = 2.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000		-.959	
	.0100		-3.988	
	.0250		-4.343	
	.0500		-3.870	
	.1000		-2.231	
	.2250		-1.243	
	.4500		-.787	
	.7500		-.379	
	.8350		-.148	
	.8750		-.041	
	.9250		-.041	
	.9750		-.030	
Lower	.0500		.746	
	.1500		.361	
	.4000		.136	
	.7000		-.018	
	.8530		-.018	
	.9250		-.041	
	.9750		-.036	
Wing				
Upper	.0000		-.941	
	.0100		-3.746	
	.0250		-4.041	
	.0500		-3.568	
	.1000		-2.195	
	.2250		-1.178	
	.4500		-.716	
	.7500		-.124	
Lower	.0500		.775	
	.1500		.290	
	.4000		.172	
	.7000		.024	
Aileron				
Upper	.8050		-.101	
	.8150		-.018	
	.8350		-.077	
	.8750		.018	
	.9250		-.018	
	.9750		.065	
			-.077	
Lower	.8100		-.071	
	.8340		.053	
	.9250		.041	
	.9750		.036	
			.089	

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(h) \quad \alpha_f = 2.9^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:							
		0.1621	0.2895						
Wing									
Upper	.0000		-1.136						
	.0100		-4.500						
	.0250		-4.818						
	.0500		-4.182						
	.1000		-2.273						
	.2250		-.909						
	.4500		-.636						
	.7500		-.409						
	.8350		-.136						
	.8750		.000						
	.9250		-.091						
	.9750		-.046						
Lower	.0500		.773						
	.1500		.500						
	.4000		.227						
	.7000		.000						
	.8530		-.046						
	.9250		-.091						
	.9750		-.091						
Aileron									
Upper	.8050		-.182						
	.8150		-.091						
	.8350		-.046						
	.8750		.000						
	.9250		.000						
	.9750		.000						
Lower	.8100		-.227						
	.8340		-.091						
	.9250		-.091						
	.9750		-.046						

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^0$; cable configuration 2-5-8; $a_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(i) \quad \alpha_f = 2.8^0$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:				
		0.1621	0.2895	0.5000	0.6800	0.8920		
Wing								
Upper	.0000	-.852	-.769	Upper	.0000	-.698	-.012	.266
	.0100	-3.515	-3.769		.0100	-3.396	-2.627	-1.781
	.0250	-3.663	-4.095		.0250	-3.722	-3.195	-2.012
	.0500	-3.195	-3.645		.0500	-3.296	-2.923	-1.852
	.1000	-2.047	-2.083		.1000	-2.142	-1.686	-1.225
	.2250	-1.302	-1.189		.2250	-1.172	-.964	-.781
	.4500	-.663	-.710		.4500	-.686	-.562	-.408
	.7500	-.296	-.462		.7500	-.367	-.391	-.574
	.8350	-.000	-.047					
	.8750	-.041	-.118					
	.9250	-.065	-.036					
	.9750	-.059	.006					
Lower	.0500	.621	.740	Lower	.0500	.775	.592	.408
	.1500	.373	.361		.1500	.355	.272	.095
	.4000	.101	.107		.4000	.112	.071	-.065
	.7000	.083	.077		.7000	.225	.231	-.041
	.8530	-.018	-.024					
	.9250	-.071	-.024					
	.9750	-.083	-.065					
Aileron								
Upper	.8050	-.331	-.325	Upper	.8050	-.331	-.325	-.598
	.8150	-.266	-.320		.8150	-.266	-.320	-.568
	.8350	-.225	-.219		.8350	-.225	-.219	-.669
	.8750	-.154	-.148		.8750	-.154	-.148	-.456
	.9250	-.124	-.142		.9250	-.124	-.142	-.337
	.9750	-.118	-.154		.9750	-.118	-.154	-.325
Lower	.8100	.213	.343	Lower	.8100	.213	.343	.118
	.8340	.296	.308		.8340	.296	.308	.130
	.9250	-.024	.112		.9250	-.024	.112	-.089
	.9750	-.012	.006		.9750	-.012	.006	-.183

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.2 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(j) \quad \alpha_f = 6.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-2.083			
	.0100		-5.633			
	.0250		-5.538			
	.0500		-4.450			
	.1000		-2.503			
	.2250		-1.172			
	.4500		-.521			
	.7500		-.065			
	.8350		-.024			
	.8750		-.036			
Lower	.9250		-.036			
	.9750		-.065			
	.0500		1.018			
	.1500		.734			
	.4000		.574			
	.7000		.414			
	.8530		.018			
Upper	.9250		.018			
	.9750		.018			
Aileron						
.8050		.000				
.8150		-.047				
.8350		-.006				
.8750		.036				
.9250		.024				
.9750		.095				
.8100		.065				
.8340		.107				
Lower	.9250		.053			
	.9750		.065			
	.8100		.047			
	.8340		.237			
Lower	.9250		-.006			
	.9750		.036			
	.8100		.118			
	.8340		-.083			

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 10.8 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(k) \quad \alpha_f = 6.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						Wing
Upper						
	.8350			.0000	-1.875	-1.100
	.8750			.0100	-4.017	-3.492
	.9250			.0250	-3.950	-3.517
	.9750			.0500	-3.175	-3.296
				.1000	-1.904	-1.521
				.2250	-.896	-.717
				.4500	-.350	-.238
				.7500	-.063	-.038
Lower						
	.0500			.0500	.608	.654
	.1500			.1500	.367	.392
	.4000			.4000	.150	.113
	.7000			.7000	-.067	-.133
Aileron						
Upper						
	.8050			.8050	-.150	-.042
	.8150			.8150	-.067	-.004
	.8350			.8350	-.025	.000
	.8750			.8750	-.025	.021
	.9250			.9250	-.021	.054
	.9750			.9750	-.021	.054
Lower						
	.8100			.8100	-.025	-.275
	.8340			.8340	-.033	-.088
	.9250			.9250	-.025	-.042
	.9750			.9750	-.017	-.004

TABLE XV.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 10.8 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(l) \quad \alpha_f = 6.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.308			
	.0100		-3.854			
	.0250		-3.896			
	.0500		-3.154			
	.1000		-1.754			
	.2250		-.896			
	.4500		-.429			
	.7500		-.075			
	.8350		-.021			
	.8750		.029			
	.9250		.025			
	.9750		.029			
Lower	.0500		.700			
	.1500		.479			
	.4000		.246			
	.7000		.146			
	.8530		.088			
	.9250		.046			
	.9750		-.038			
Aileron						
Upper	.8050		-.121			
	.8150		-.108			
	.8350		-.071			
	.8750		-.067			
	.9250		-.075			
	.9750		-.038			
Lower	.8100		.283			
	.8340		.283			
	.9250		.033			
	.9750		.021			

TABLE XVI

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(a) $\alpha_f = -4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		C_p for values of $y/b/2$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.636	.697	.727	.879	.879
	.0100	-1.273	-1.364	-1.242	-0.939	-0.545
	.0250	-1.848	-2.091	-1.909	-1.697	-1.030
	.0500	-1.697	-1.879	-1.727	-1.667	-1.030
	.1000	-1.182	-1.364	-1.394	-1.091	-0.879
	.2250	-0.758	-0.818	-0.788	-0.727	-0.606
	.4500	-0.485	-0.545	-0.545	-0.424	-0.303
	.7500	-0.242	-0.485	-0.121	-0.152	-0.061
	.8350	.000	.000			
	.8750	-0.030	.061			
	.9250	.030	.061			
	.9750	.061	.091			
Lower	.0500	.182	.212	.182	.030	-.091
	.1500	.000	-.121	-.030	-.182	-.212
	.4000	-.152	-.121	-.182	-.242	-.182
	.7000	.030	-.121	-.091	-.182	-.152
	.8530	.030	-.061			
	.9250	.030	-.030			
	.9750	.030	.000			
Aileron						
Upper	.8050		-.091	-.091	.000	
	.8150		-.121	-.091	.030	
	.8350		-.030	-.091	-.121	
	.8750		-.030	-.121	-.152	
	.9250		-.030	-.061	-.061	
	.9750		.000	.000	.000	
Lower	.8100		-.091	-.091	-.121	
	.8340		.000	.000	-.091	
	.9250		-.030	-.061	-.091	
	.9750		.030	.000	.000	

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.8^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		.455			
	.0100		-1.754			
	.0250		-2.500			
	.0500		-2.197			
	.1000		-1.434			
	.2250		-.881			
	.4500		-.664			
	.7500		-.475			
	.8350		-.131			
	.8750		-.053			
Lower	.9250		-.053			
	.9750		-.020			
	.0500		.344			
	.1500		-.029			
	.4000		-.135			
	.7000		-.143			
Upper	.8530		-.074			
	.9250		-.074			
	.9750		-.057			
	.8050		-.172			
	.8150		-.074			
	.8350		-.074			
	.8750		-.025			
Lower	.9250		-.041			
	.9750		-.049			
	.8100		-.012			
	.8340		-.012			
	.9250		-.066			
Aileron	.9750		-.029			
	.8050		.066			
	.8150		.082			
	.8350		.061			
	.8750		.033			
Upper	.9250		.033			
	.9750		.041			
	.8100		-.008			
	.8340		-.102			
	.9250		-.041			
Lower	.9750		.033			
	.8100		-.275			
	.8340		-.230			
	.9250		-.086			
	.9750		-.029			

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(c) \quad \alpha_f = -4.7^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/b/2$ of:		
		0.1621	0.2895	
Wing				
Upper	.0000		.402	
	.0100		-.742	
	.0250		-1.455	
	.0500		-1.361	
	.1000		-1.164	
	.2250		-.885	
	.4500		-.701	
	.7500		-.545	
	.8350		-.045	
	.8750		-.057	
	.9250		-.107	
	.9750		-.082	
Lower	.0500		-.066	
	.1500		-.369	
	.4000		-.307	
	.7000		-.156	
	.8530		-.070	
	.9250		-.045	
	.9750		-.008	
Wing				
Upper	.0000		.885	
	.0100		-.209	
	.0250		-.988	
	.0500		-.926	
	.1000		-1.086	
	.2250		-.668	
	.4500		-.553	
	.7500		-.447	
Lower	.0500		-.238	
	.1500		-.307	
	.4000		-.156	
	.7000		.111	
Aileron				
Upper	.8050		-.533	
	.8150		-.500	
	.8350		-.459	
	.8750		-.348	
	.9250		-.266	
	.9750		-.164	
Lower	.8100		.213	
	.8340		.451	
	.9250		.016	
	.9750		-.066	

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.4 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened

(d) $\alpha_f = -1.0^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						Wing
Upper	.0000	.042	.096	Upper	.0000	.075
	.0100	-2.217	-2.454		.0100	-2.321
	.0250	-2.467	-2.904		.0250	-2.775
	.0500	-2.208	-2.583		.0500	-2.442
	.1000	-1.500	-1.571		.1000	-1.750
	.2250	-.921	-.908		.2250	-.908
	.4500	-.517	-.546		.4500	-.638
	.7500	-.154	-.350		.7500	-.221
	.8350	-.071	-.071			
	.8750	-.017	.008			
	.9250	.063	.100			
	.9750	.054	.150			
Lower	.0500	.517	.621	Lower	.0500	.550
	.1500	.267	.163		.1500	.175
	.4000	.042	.017		.4000	-.046
	.7000	.042	.017		.7000	.046
	.8530	-.021	.038			
	.9250	-.088	.017			
	.9750	.033	.038			
Aileron						
Upper	.8050	-.221	-.204	Upper	.8050	-.221
	.8150	-.179	-.179		.8150	-.179
	.8350	-.138	-.154		.8350	-.138
	.8750	-.046	-.096		.8750	-.046
	.9250	.021	-.013		.9250	.021
	.9750	.033	.067		.9750	.033
Lower	.8100	.146	.096	Lower	.8100	.146
	.8340	.146	.192		.8340	.146
	.9250	-.029	.021		.9250	-.029
	.9750	.029	.058		.9750	.029

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(e) $\alpha_f = -1.0^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-.373	-.242	.094	.348	
	.0100	-2.803	-3.225	-2.488	-1.701	
	.0250	-3.041	-3.664	-3.053	-1.914	
	.0500	-2.697	-3.234	-2.779	-1.758	
	.1000	-1.627	-1.770	-1.553	-1.143	
	.2250	-.963	-.943	-.848	-.672	
	.4500	-.520	-.607	-.381	-.270	
	.7500		-.332	-.008	.066	
	.8350	.000	-.115			
	.8750	-.025	-.049			
	.9250	-.016	-.037			
	.9750	-.012	-.041			
Lower	.0500	.512	.680	.590	.414	
	.1500	.266	.270	.234	.111	
	.4000	.086	.045	-.004	-.070	
	.7000	-.016	-.016	-.160	-.225	
	.8530	-.082	-.037			
	.9250	-.131	-.086			
	.9750	-.033	-.066			
Aileron						
Upper	.8050	-.066	.045	.127		
	.8150	-.066	.053	.201		
	.8350	.000	.057	.148		
	.8750	.020	.029	.033		
	.9250	.016	.049	-.012		
	.9750	.045	.070	-.061		
Lower	.8100	.033	.053	-.299		
	.8340	.049	.020	-.172		
	.9250	-.029	-.029	-.176		
	.9750	.020	.049	-.090		

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(f) \quad \alpha_f = -1.0^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		.324			
	.0100		-2.160			
	.0250		-2.787			
	.0500		-2.357			
	.1000		-1.561			
	.2250		-.881			
	.4500		-.594			
	.7500		-.459			
	.8350		-.057			
	.8750		-.057			
	.9250		-.037			
	.9750		-.037			
Lower	.0500		.328			
	.1500		.070			
	.4000		.004			
	.7000		-.070			
	.8530		-.008			
	.9250		.025			
	.9750		-.020			
Aileron						
Upper	.8050		-.303			
	.8150		-.373			
	.8350		-.287			
	.8750		-.266			
	.9250		-.266			
	.9750		-.168			
Lower	.8100		.303			
	.8340		.516			
	.9250		.074			
	.9750		.020			

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$, belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(g) \quad \alpha_f = 2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-.182	-.061	.242	.455	
	.0100	-2.727	-3.061	-2.364	-1.606	
	.0250	-2.939	-3.576	-3.030	-1.939	
	.0500	-2.667	-3.152	-2.727	-1.818	
	.1000	-1.818	-2.242	-1.636	-1.242	
	.2250	-.848	-1.000	-.970	-.758	
	.4500	-.424	-.576	-.515	-.364	
	.7500	-.303	-.394	-.182	-.121	
	.8350	-.061	-.091			
	.8750	-.030	-.000			
	.9250	-.182	-.000			
	.9750	-.121	-.152			
Lower	.0500	.606	.788			
	.1500	.121	.303			
	.4000	.030	.152			
	.7000	.091	.121			
	.8530	-.091	.030			
	.9250	-.030	-.030			
	.9750	.061	-.030			
Aileron						
Upper	.8050	-.091	-.091			
	.8150	-.091	-.091			
	.8350	-.030	-.091			
	.8750	.000	-.091			
	.9250	.030	-.030			
	.9750	.061	.061			
Lower	.9100	-.081	.030			
	.8340	.030	.182			
	.9250	.000	.000			
	.9750	.061	.061			

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(h) \quad \alpha_f = 2.8^\circ$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.635			
	.0100		-5.373			
	.0250		-5.307			
	.0500		-4.410			
	.1000		-2.414			
	.2250		-1.246			
	.4500		-.734			
	.7500		-.213			
	.8350		-.115			
	.8750		-.098			
	.9250		-.004			
	.9750		-.025			
Lower	.0500		.951			
	.1500		.459			
	.4000		.148			
	.7000		.107			
	.8530		-.008			
	.9250		-.045			
	.9750		-.098			
Aileron						
Upper	.8050		-.078		.016	+.057
	.8150		-.078		.037	.114
	.8350		-.049		.045	.114
	.8750		-.025		.053	-.057
	.9250		.016		-.057	-.085
	.9750		.049		-.057	-.142
Lower	.8100		.086		.000	-.171
	.8340		.049		-.114	-.313
	.9250		-.029		-.171	-.256
	.9750		.045		-.085	

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^0$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(i) \quad \alpha_f = 2.8^0$$

Surface	$\frac{x}{c}$	C_p for values of y/b of:			
		0.1621	0.2895		
Wing					
Upper	.0000		-.750		
	.0100		-3.951		
	.0250		-4.266		
	.0500		-3.742		
	.1000		-2.225		
	.2250		-1.287		
	.4500		-.807		
	.7500		-.484		
	.8350		-.221		
	.8750		-.098		
	.9250		-.033		
	.9750		-.041		
Lower	.0500		.758		
	.1500		.205		
	.4000		.135		
	.7000		.016		
	.8530		-.041		
	.9250		-.029		
	.9750		-.037		
Wing					
Upper	.0000		-.533	.205	
	.0100		-3.389	-1.500	
	.0250		-3.832	-1.816	
	.0500		-3.193	-1.672	
	.1000		-2.156	-1.189	
	.2250		-1.193	-.775	
	.4500		-.775	-.406	
Lower	.7500		-.410	-.570	
	.0500		.553	.311	
	.1500		.307	.049	
	.4000		.082	-.074	
Aileron	.7000		.184	-.061	
	Aileron				
	.8050		-.398	-.320	-.545
	.8150		-.352	-.320	-.504
	.8350		-.324	-.225	-.602
Upper	.8750		-.303	-.172	-.393
	.9250		-.246	-.156	-.279
	.9750		-.180	-.168	-.279
	.8100		.291	.402	.143
Lower	.8340		.553	.434	.143
	.9250		.045	.139	-.131
	.9750		-.037	.008	-.242

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = -150^\circ$; cable configuration 2-5-8; $q_{\infty} = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

(j) $\alpha_f = 7.3^\circ$

Surface	$\frac{x}{c}$	C_p for values of y/b of:		C_p for values of y/b of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						Wing
Upper	.0000	-1.994	-1.610	Upper	.0000	-1.823
	.0100	-5.640	-5.799		.0100	-4.610
	.0250	-5.451	-5.933		.0250	-3.646
	.0500	-4.530	-5.591		.0500	-2.896
	.1000	-2.823	-2.329		.1000	-2.012
	.2250	-1.677	-1.659		.2250	-1.591
	.4500	-.945	-1.122		.4500	-1.085
	.7500	-.610	-.549		.7500	-.530
	.8350	-.384	-.494			
	.8750	-.573	-.494			
	.9250	-.537	-.421			
	.9750	-.500	-.311			
Lower	.0500	.933	1.311	Lower	.0500	1.177
	.1500	.390	.860		.1500	.768
	.4000	.159	.360		.4000	.165
	.7000	.067	.067		.7000	-.024
	.8530	-.177	-.098			
	.9250	-.457	-.220			
	.9750	-.274	-.445			
Aileron						
Upper	.8050	-.695	-.427	Upper	.8050	-.695
	.8150	-.524	-.311		.8150	-.524
	.8350	-.537	-.323		.8350	-.537
	.8750	-.506	-.238		.8750	-.506
	.9250	-.457	-.226		.9250	-.457
	.9750	-.415	-.226		.9750	-.415
Lower	.8100	-.280	-.018	Lower	.8100	-.280
	.8340	.165	-.152		.8340	.165
	.9250	-.232	-.183		.9250	-.232
	.9750	-.311	-.177		.9750	-.311

TABLE XVI.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

$\delta_a = 26^\circ$; cable configuration 2-5-8; $q_\infty = 14.7 \text{ lb/sq ft}$; belly plate on;
 $p = 7.0 \text{ lb/sq in.}$; forward guy cables, heavily tightened]

$$(k) \quad \alpha_f = 7.4^\circ$$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-1.246			
	.0100		-4.635			
	.0250		-4.746			
	.0500		-3.992			
	.1000		-2.193			
	.2250		-1.135			
	.4500		-.746			
	.7500		-.459			
	.8350		-.275			
	.8750		-.201			
	.9250		-.201			
	.9750		-.131			
Lower	.0500		.799			
	.1500		.389			
	.4000		.180			
	.7000		-.004			
	.8530		-.107			
	.9250		-.135			
	.9750		-.135			
Aileron						
Upper	.8050		-.348			
	.8150		-.324			
	.8350		-.270			
	.8750		-.217			
	.9250		-.217			
	.9750		-.168			
Lower	.8100		.369			
	.8340		.520			
	.9250		.025			
	.9750		-.078			

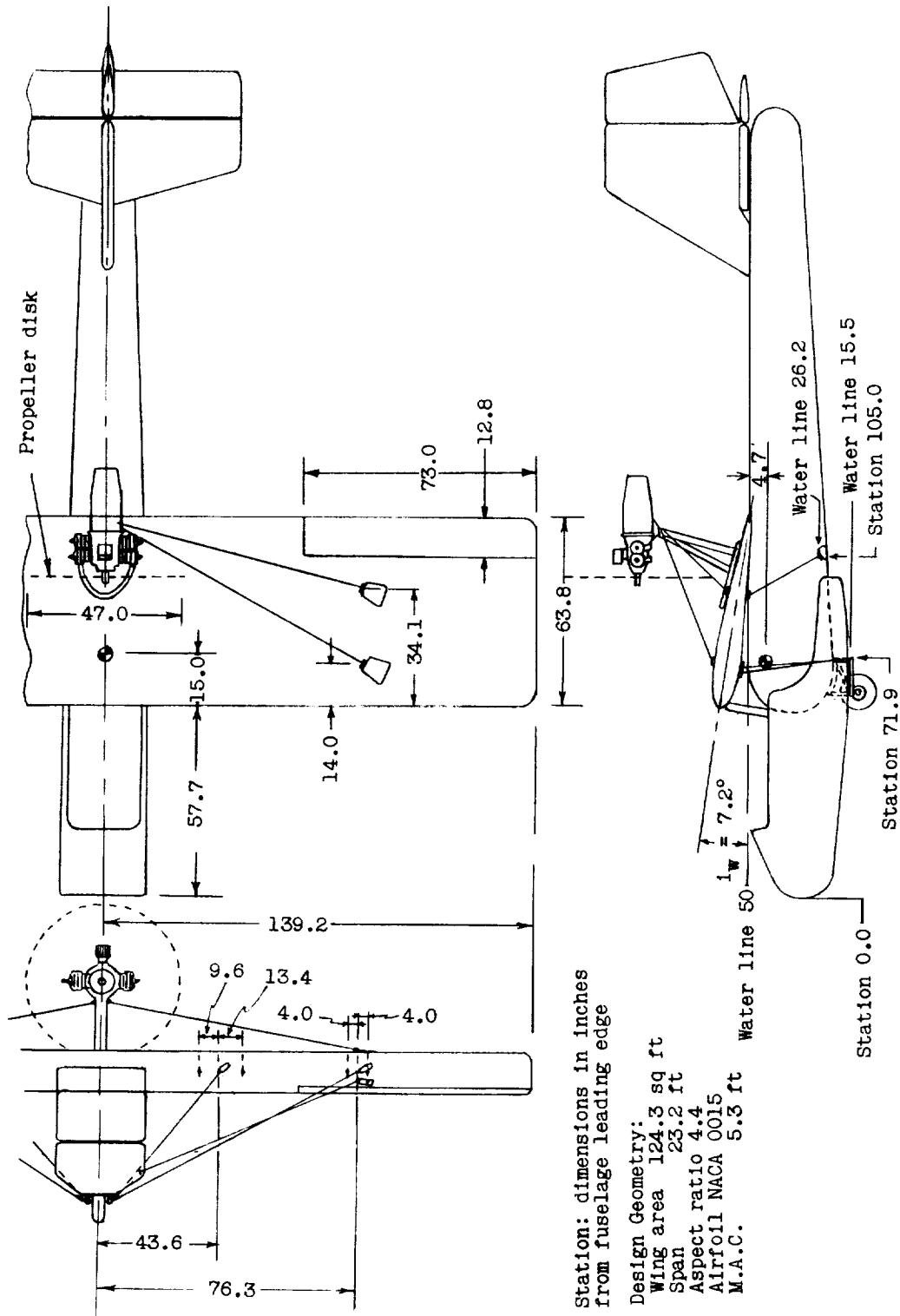


Figure 1.- Geometric characteristics of Inflatoplane II. All dimensions are in inches.



L-57-3414
Figure 2.- Inflatoplane I of reference 1 mounted for tests in the Langley full-scale tunnel.
(Airplane is similar to the one used for the present investigation.)

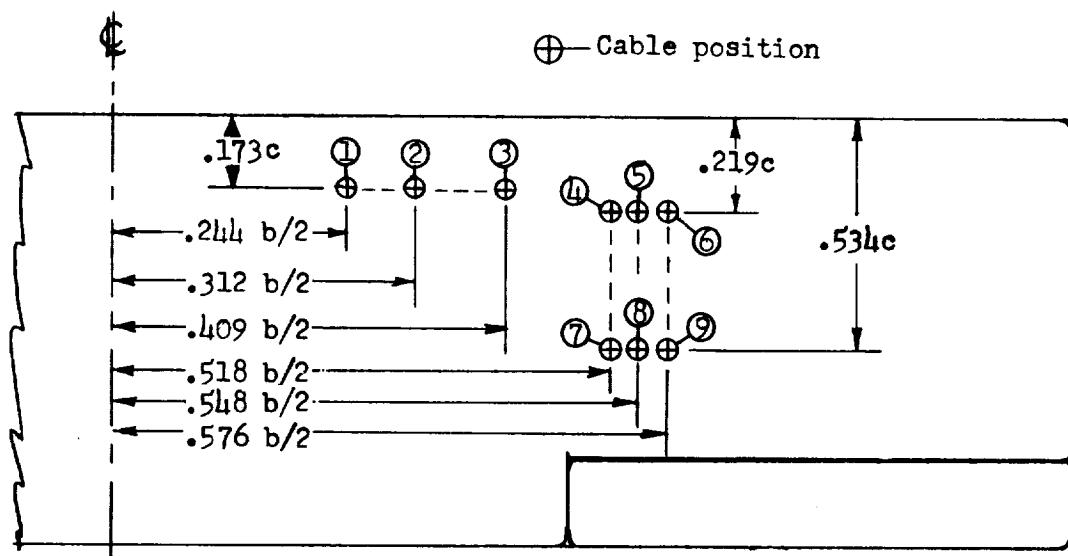


Figure 3.- Guy-cable attachment on wing lower surface. View is from above right-hand wing.

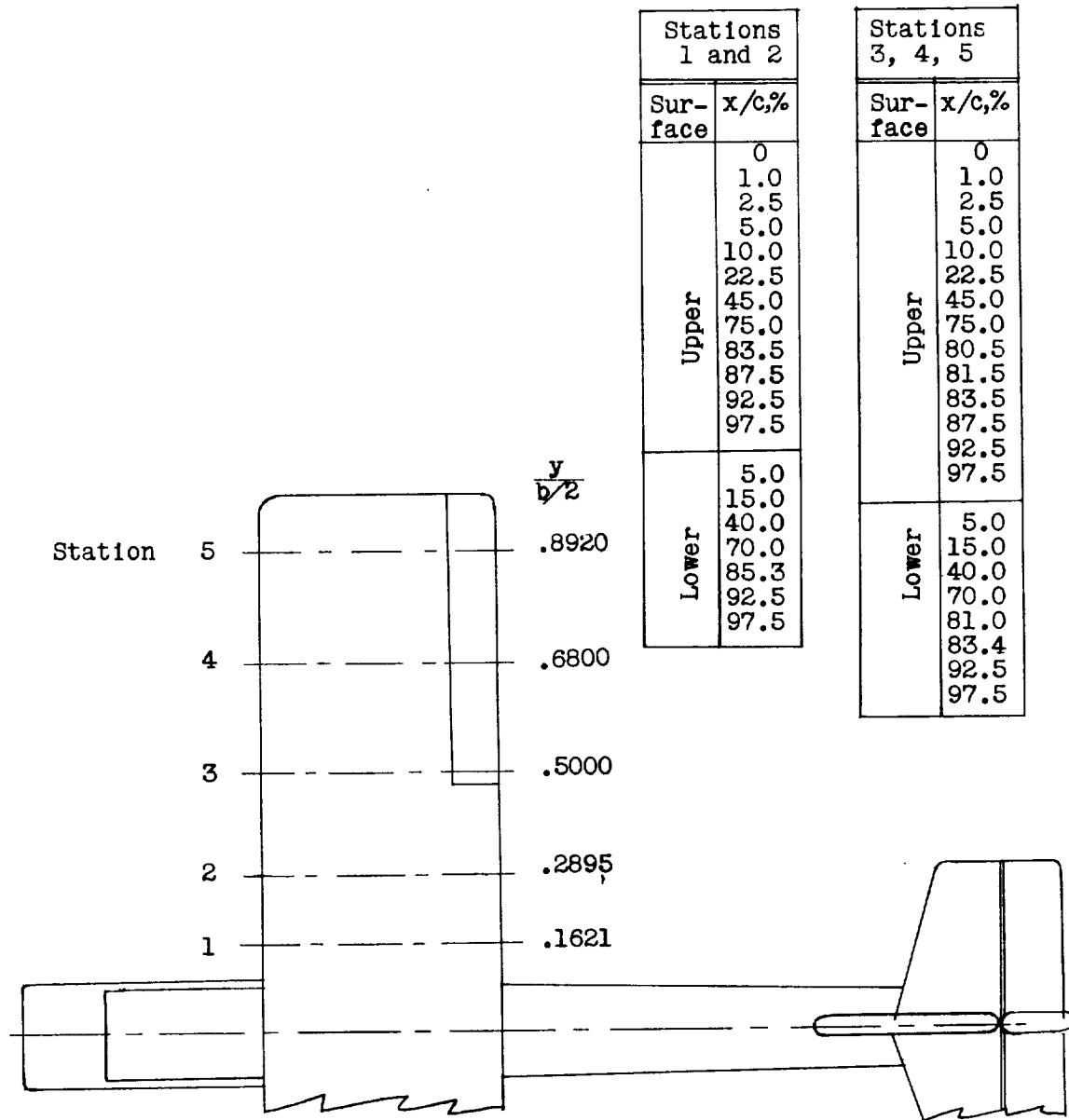


Figure 4.- Spanwise and chordwise locations of the surface-pressure orifices.

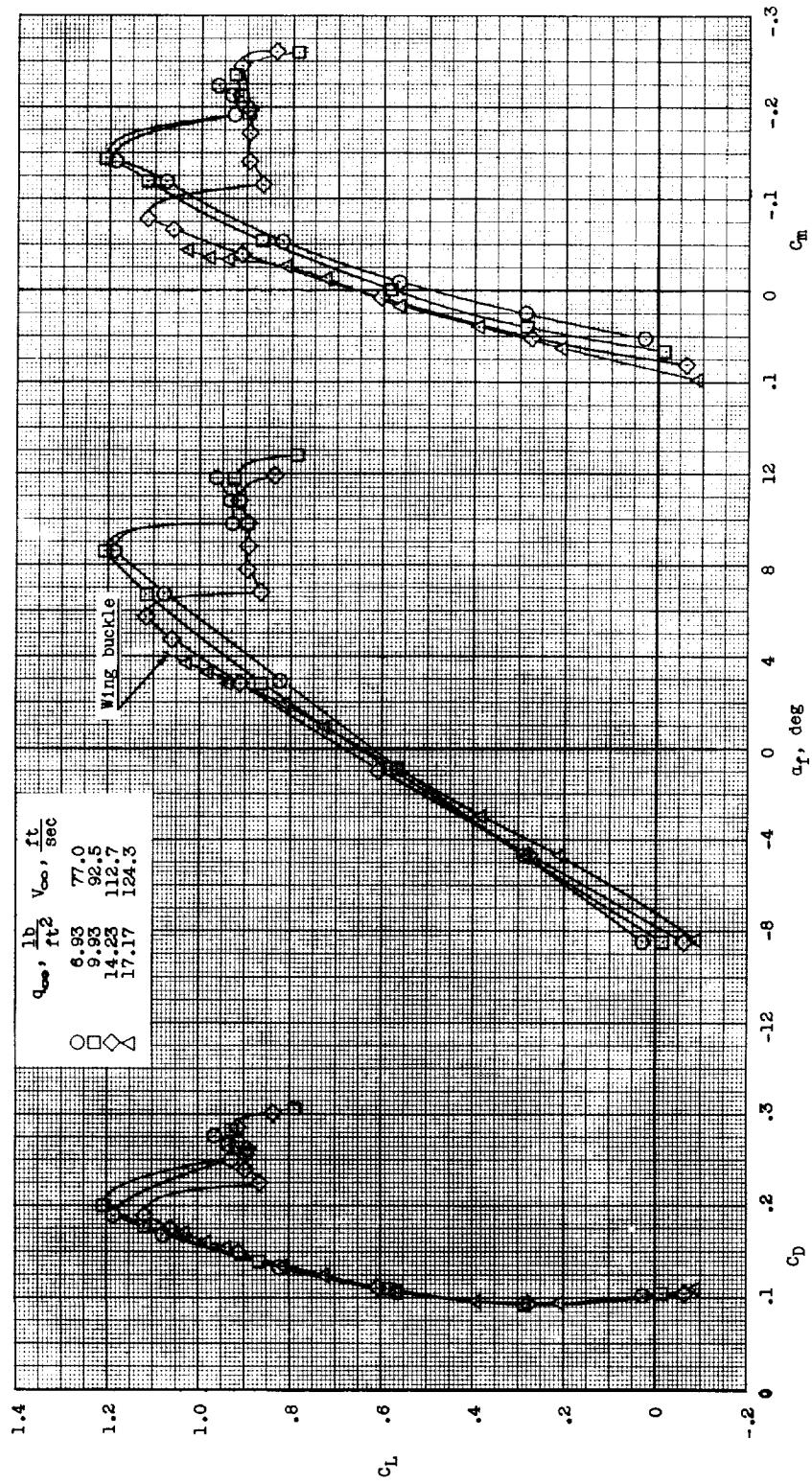


Figure 5.- Aerodynamic characteristics for several tunnel speeds. Cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened in static condition,
 $p = 7.0$ lb/sq in.

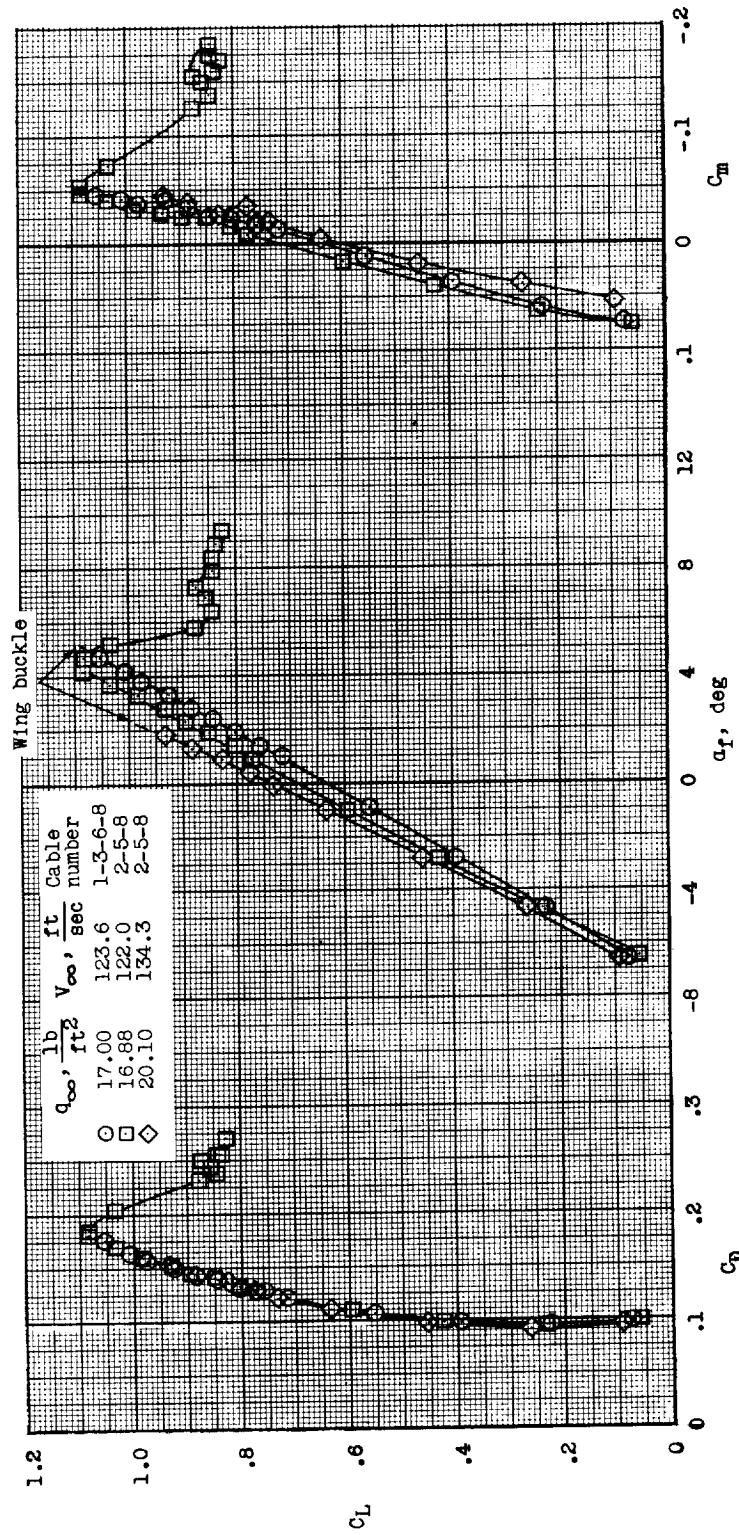


Figure 6.- Aerodynamic characteristics of the airplane for two cable configurations.
Belly plate installed; lower forward cables heavily tightened in static condition;
 $P = 7.0$ lb/sq in.



$$\begin{aligned}V_{\infty} &= 0 \frac{\text{ft}}{\text{sec}} & V_{\infty} &= 123.6 \frac{\text{ft}}{\text{sec}} \\ \alpha_f &= 3.5^\circ & \alpha_f &= 4.2^\circ \\ n &= 0 & n &= 3.98 \\ && n \text{ (just prior to buckle)} &= 4.22\end{aligned}$$

L-62-41
Figure 7.- Photographs of the airplane showing a typical wing deflection and buckle during tests. (Cross-pylon 1 is used for deflection reference.)

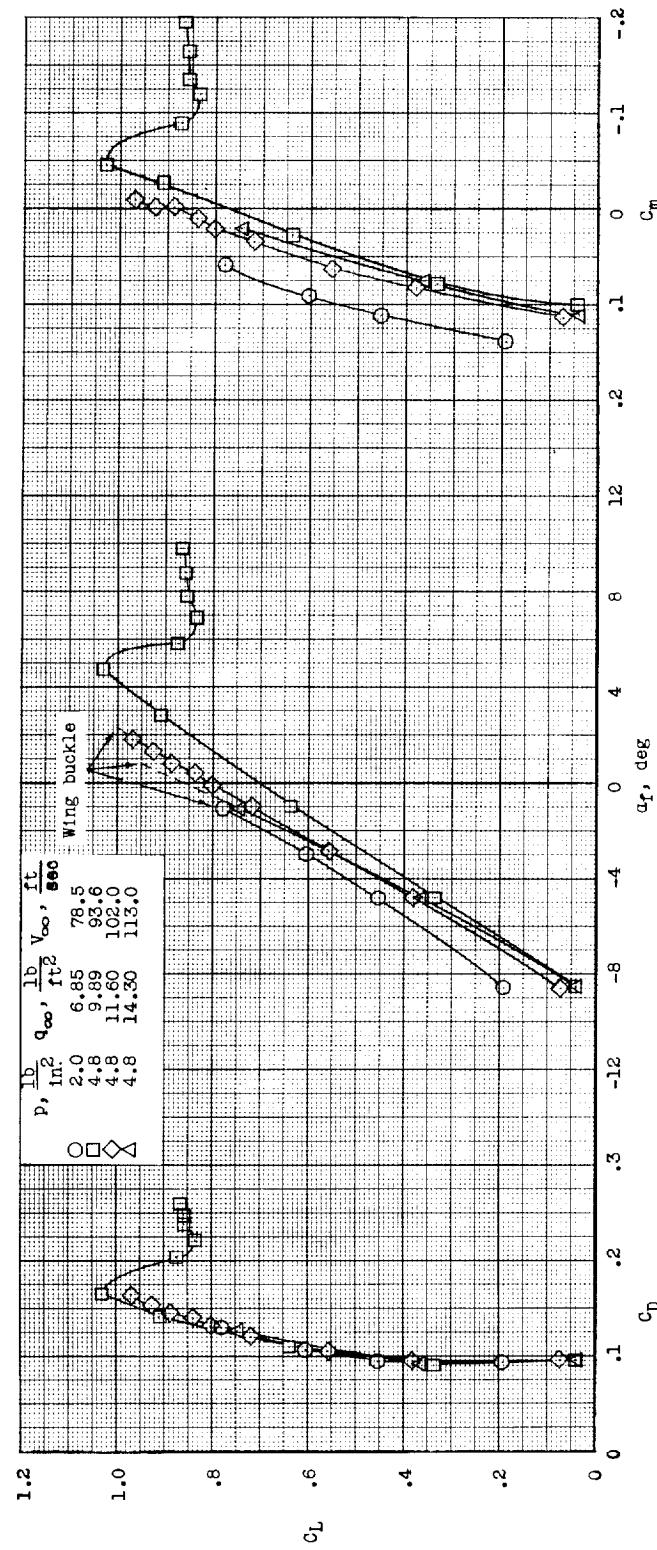


Figure 8.- Aerodynamic characteristics of the airplane at reduced inflation pressure. Cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened in static condition.

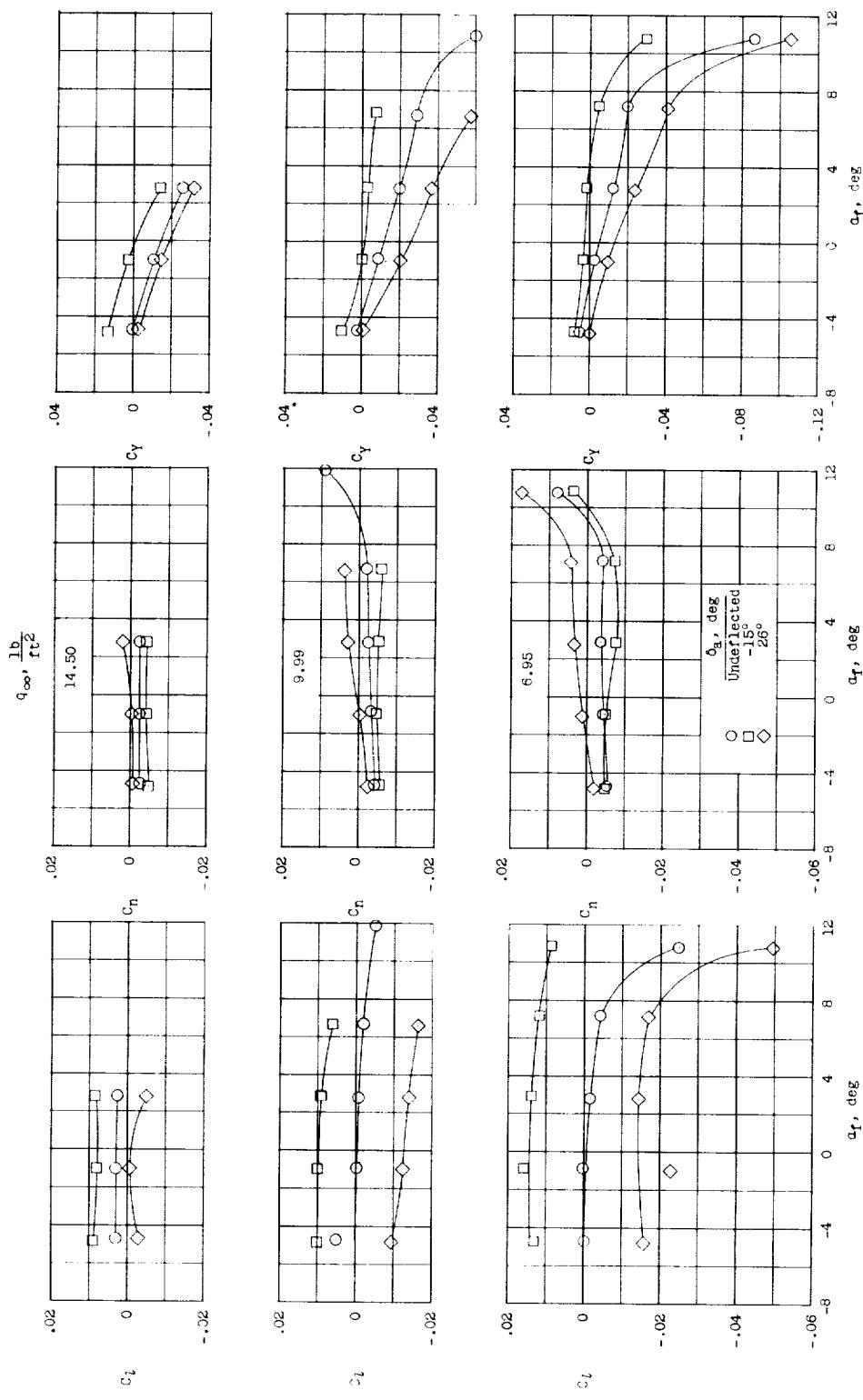


Figure 9.—Variation of lateral characteristics with angle of attack and right-hand-aileron deflection for three airspeed conditions. Cable configuration 2-5-8 with belly plate installed; lower forward cables heavily tightened; $p = 7.0 \text{ lb/sq in.}$

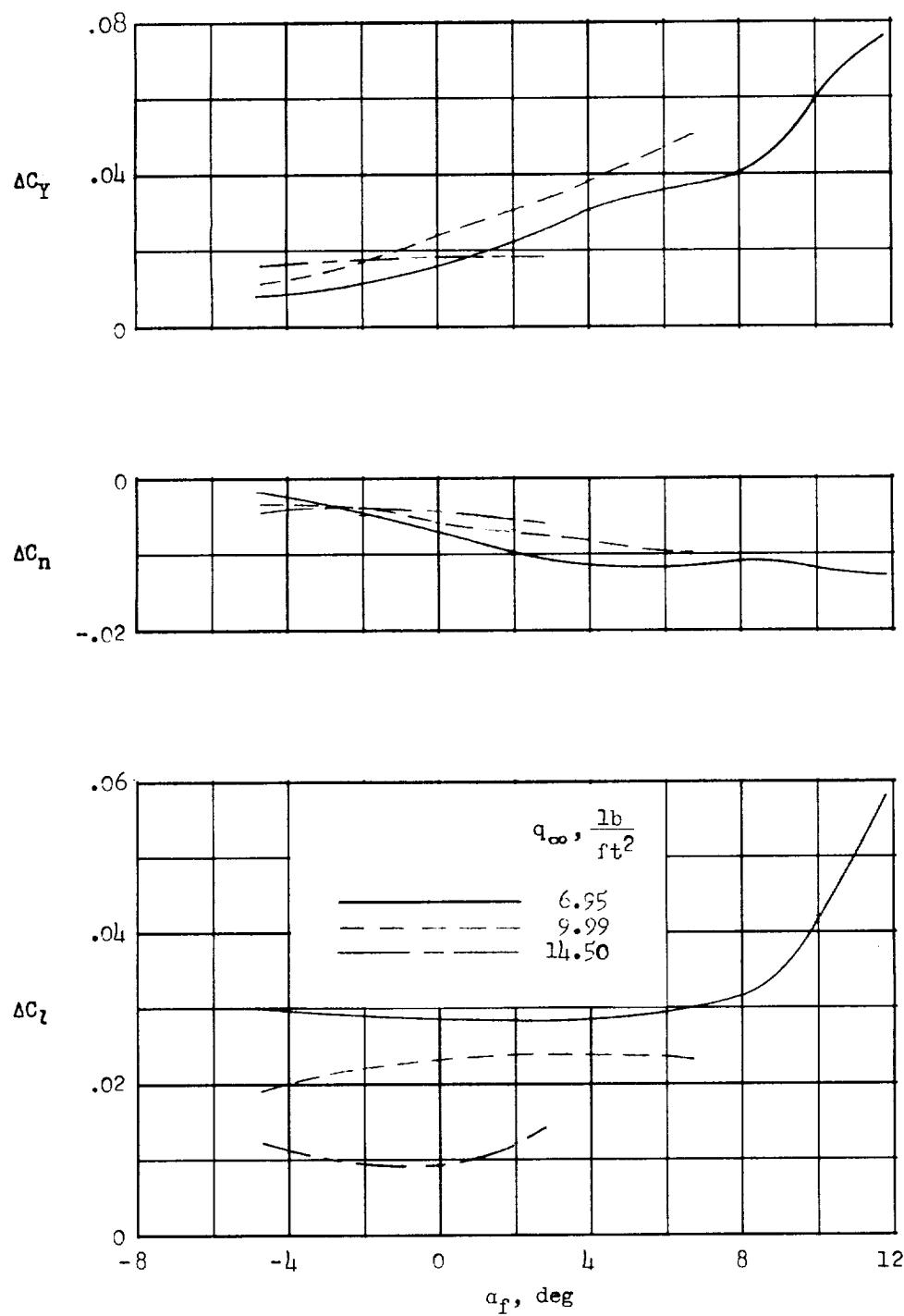


Figure 10.- Increment of C_l , C_n , and C_Y produced by deflecting the right-hand aileron up 15° and the left-hand aileron down 26° .

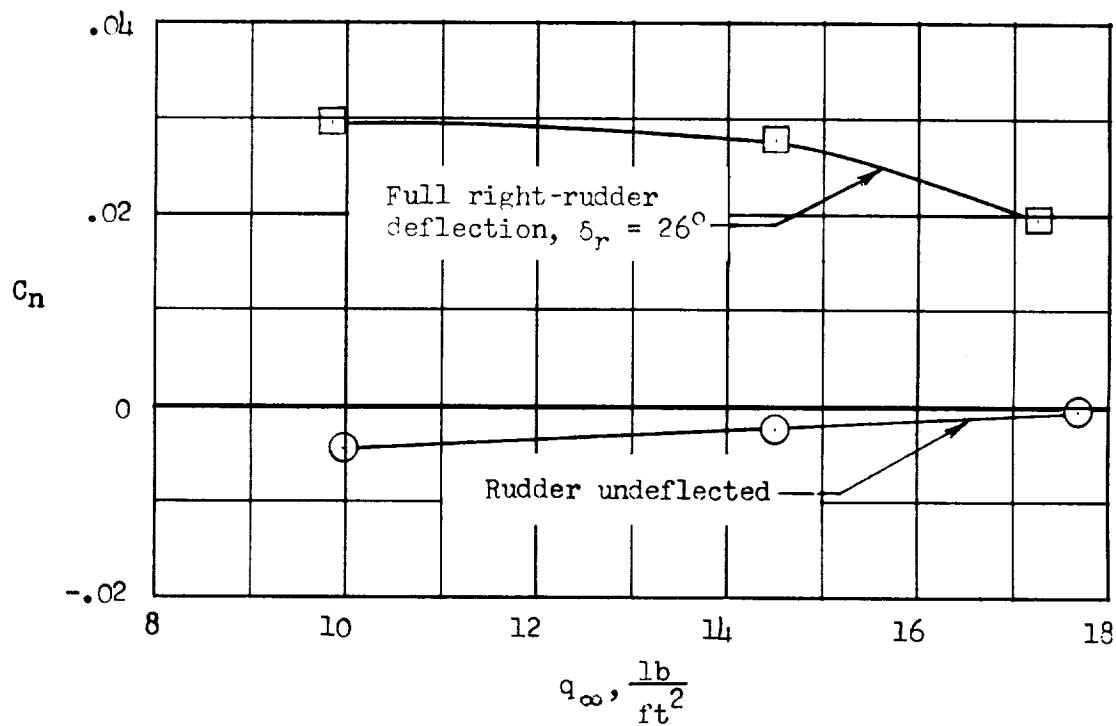


Figure 11.- Effect of full right-rudder deflection on the yawing-moment characteristics of the airplane at an angle of attack of the fuselage of about -7.5° . Cable configuration 2-5-8 with belly plate installed; lower forward cables heavily tightened; $p = 7.0 \text{ lb/sq in.}$

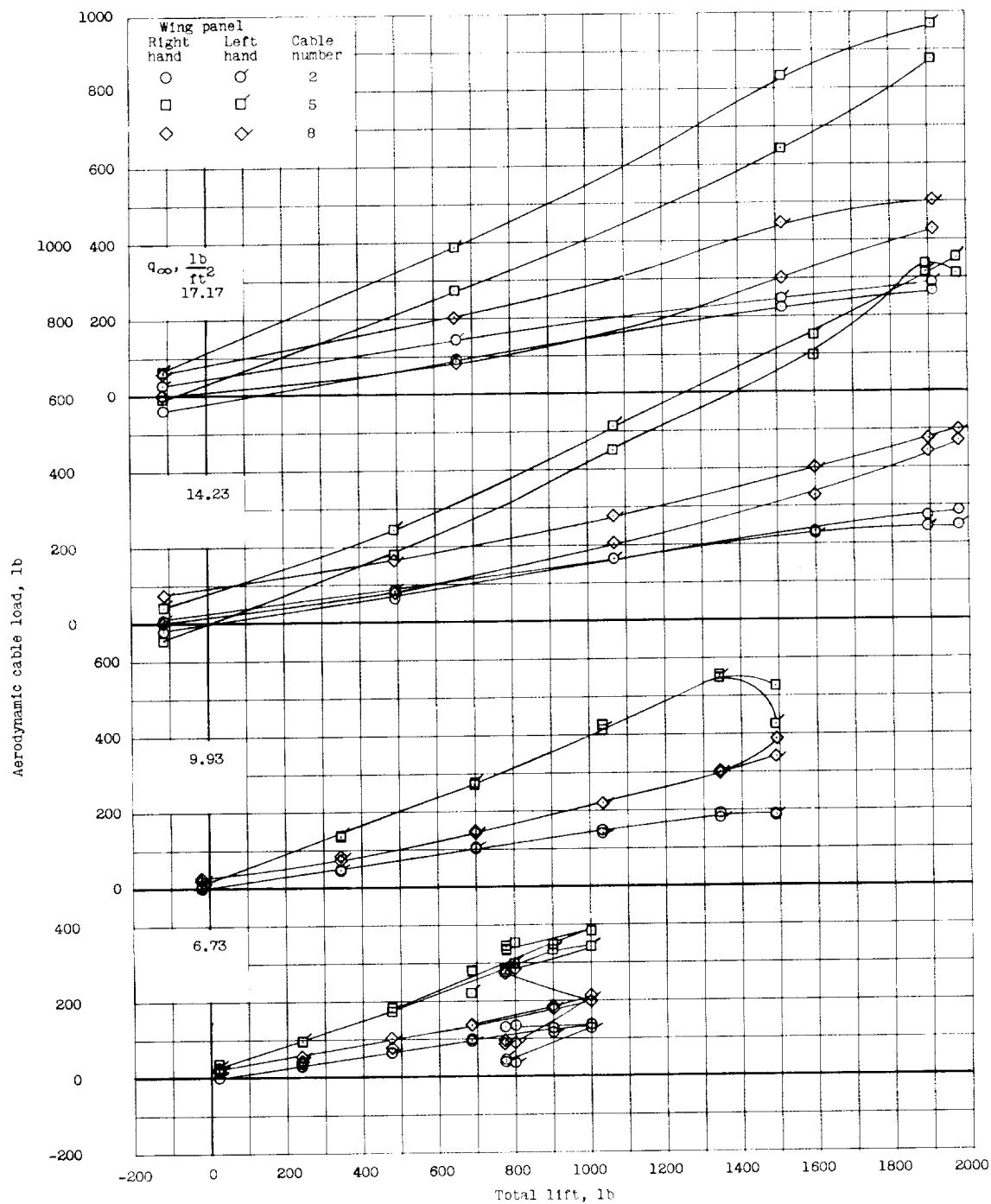


Figure 12.- Variation of the wing-guy-cable loads with total lift of the configuration for several airspeeds. Original cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened; $p = 7.0$ lb/sq in.

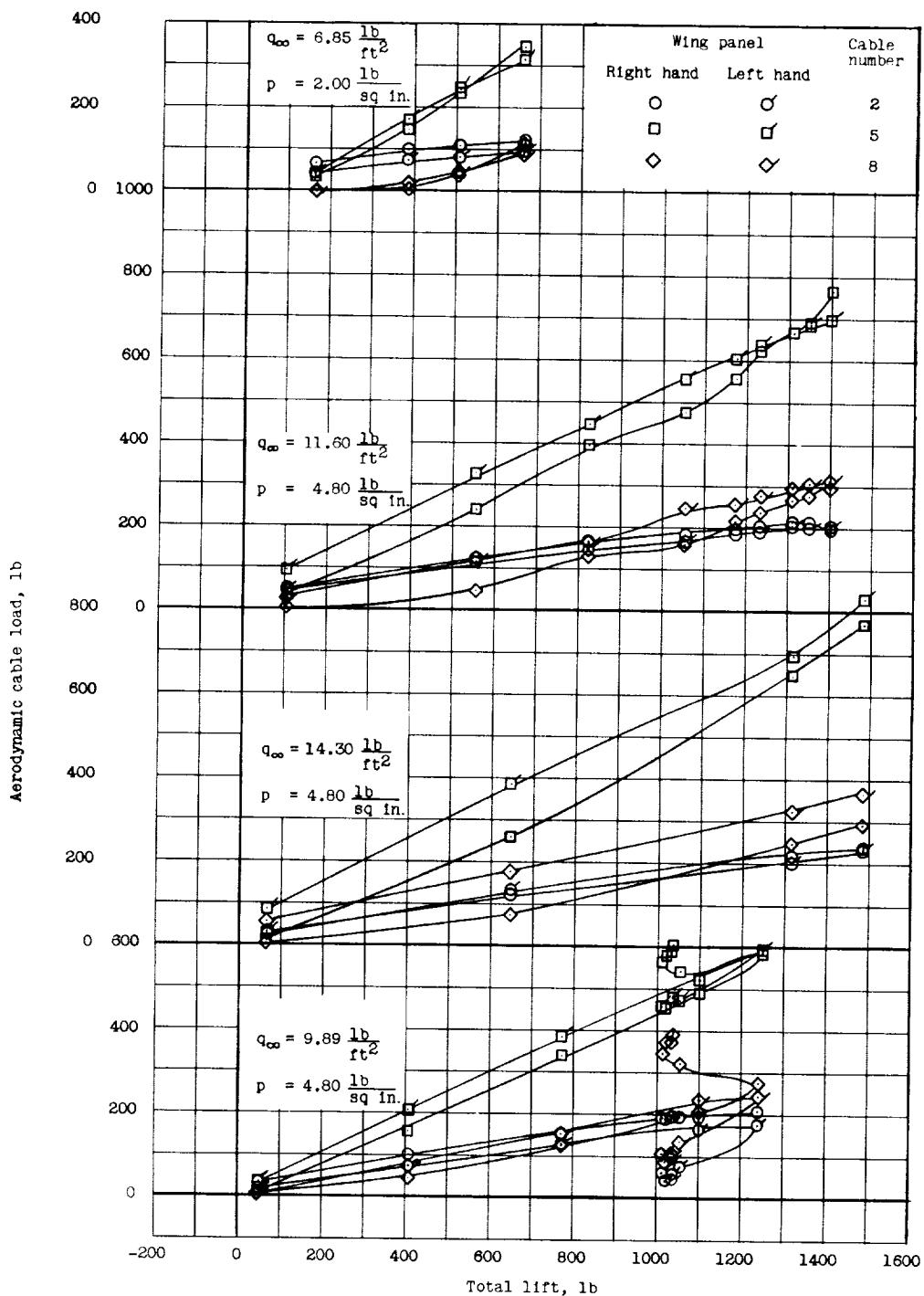


Figure 13.- Variation of the wing-guy-cable loads with total lift of the configuration for reduced inflation pressures. Original cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened.

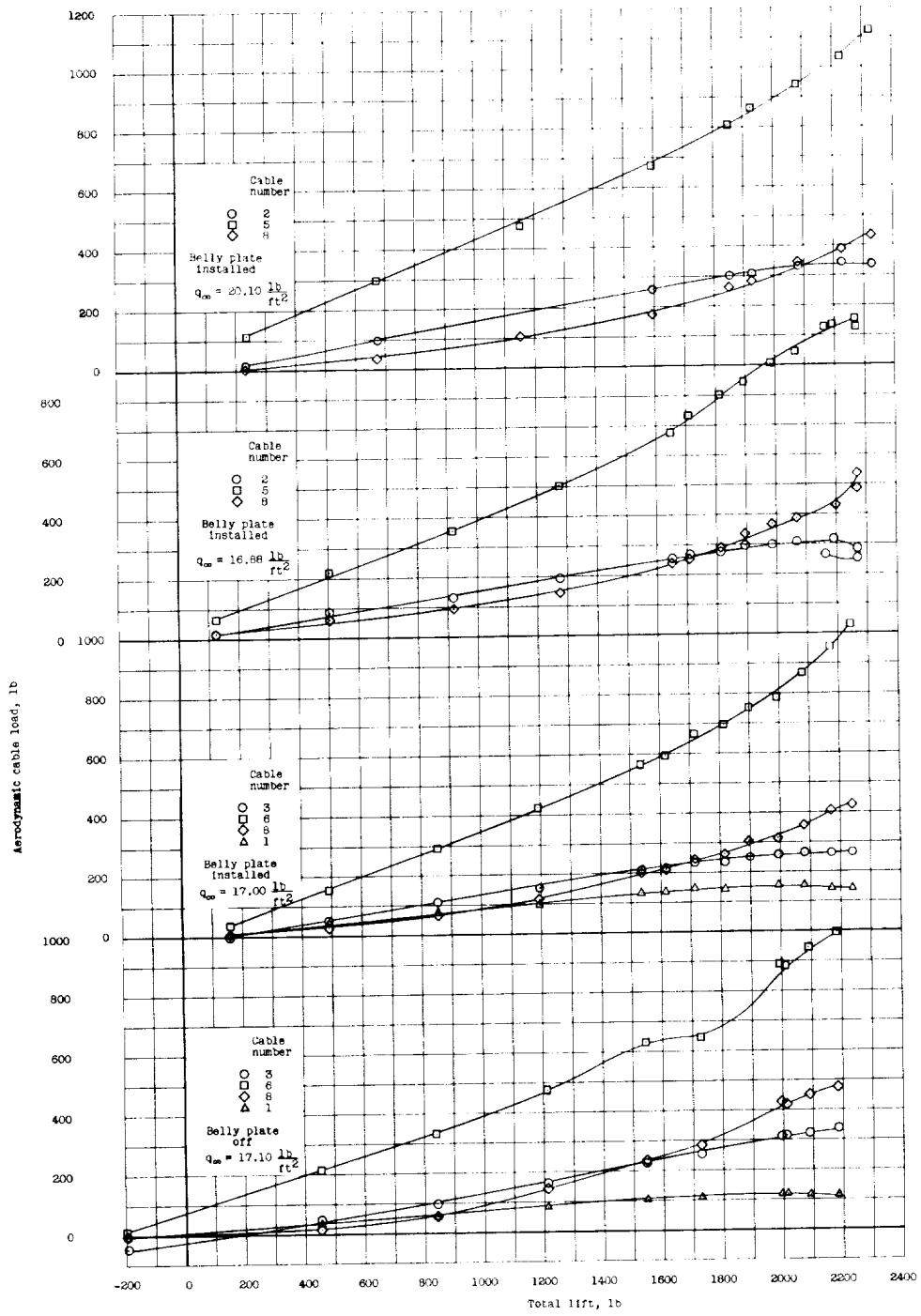


Figure 14.- Variation of the right-hand-wing guy-cable loads with total lift of the configuration. Lower forward cables heavily tightened; $p = 7.0 \text{ lb/sq in.}$

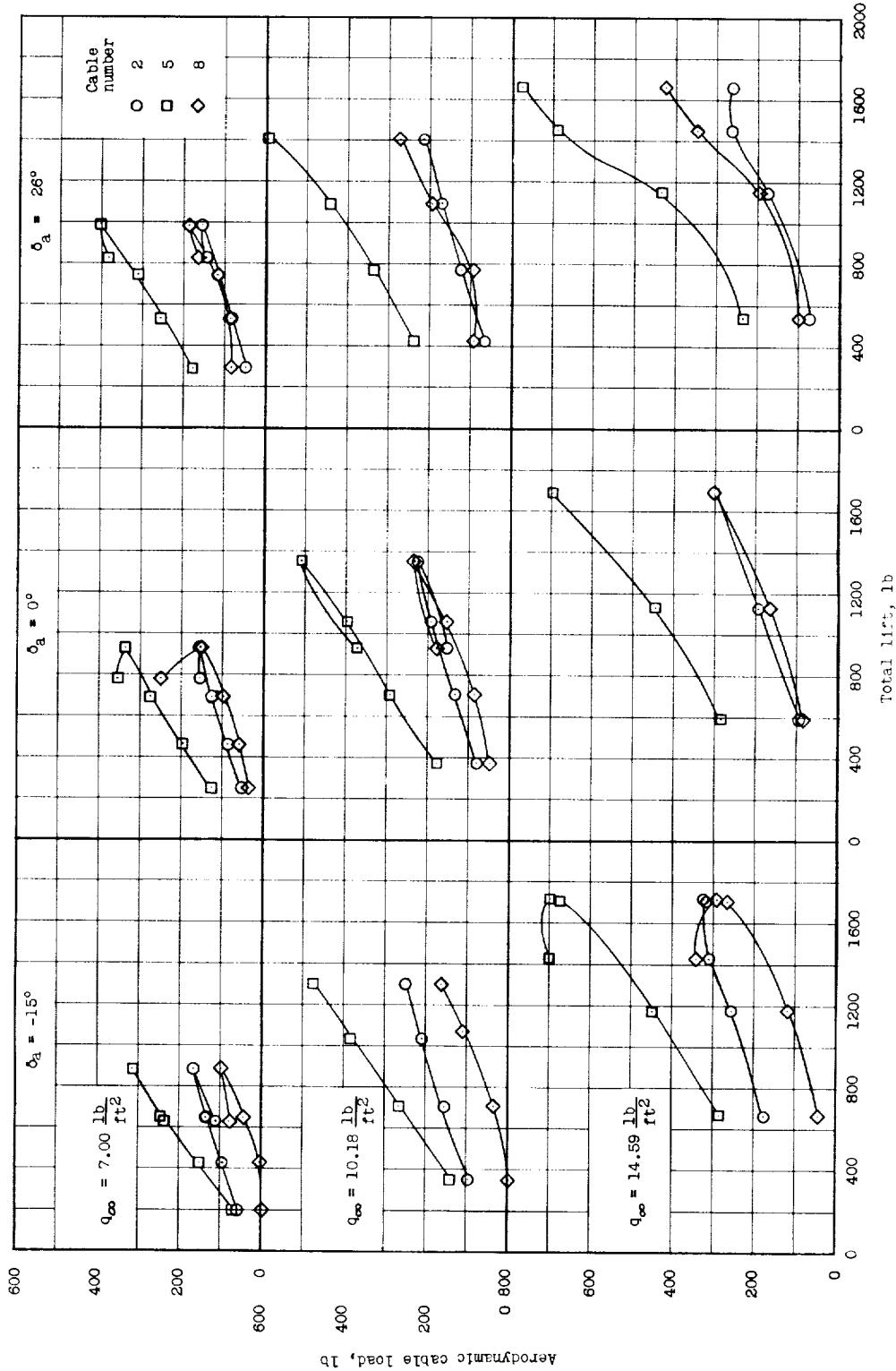
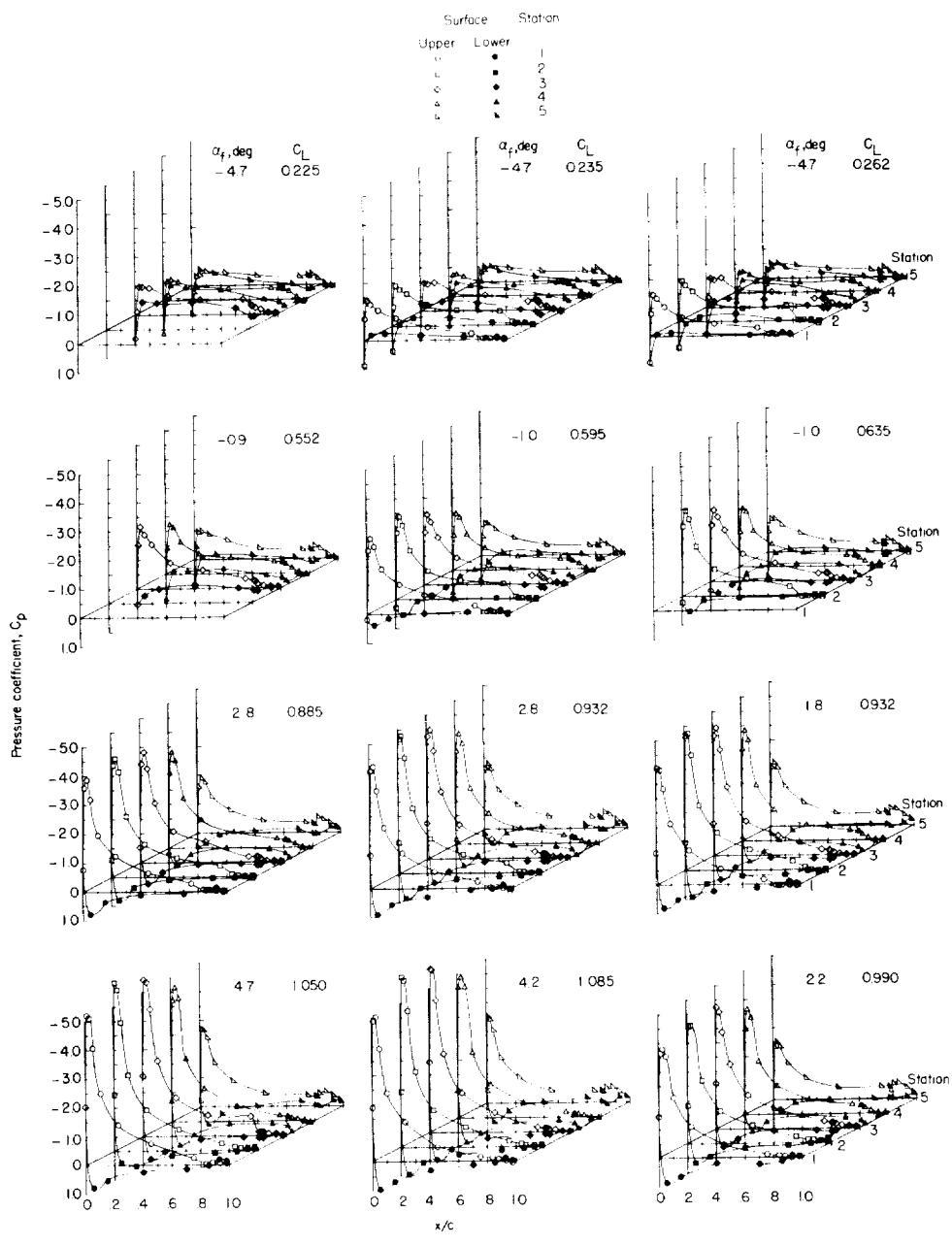


Figure 15.- Variation of the right-hand-wing cable loads with total lift of the configuration for several right-hand-aileron deflections and for several tunnel speeds. Belly plate installed; lower forward cables heavily tightened; $p = 7.0 \text{ lb/sq in.}$



(a) Cable configuration 1-3-6-8;
 $V_\infty = 123.6 \text{ ft/sec.}$

(b) Cable configuration 2-5-8;
 $V_\infty = 122.0 \text{ ft/sec.}$

(c) Cable configuration 2-5-8;
 $V_\infty = 134.3 \text{ ft/sec.}$

Figure 16.- Variation of the chordwise pressure distribution for two wing cable configurations and for increased test velocities. Belly plate installed; lower forward cables heavily tightened; $p = 7.0 \text{ lb/sq in.}$

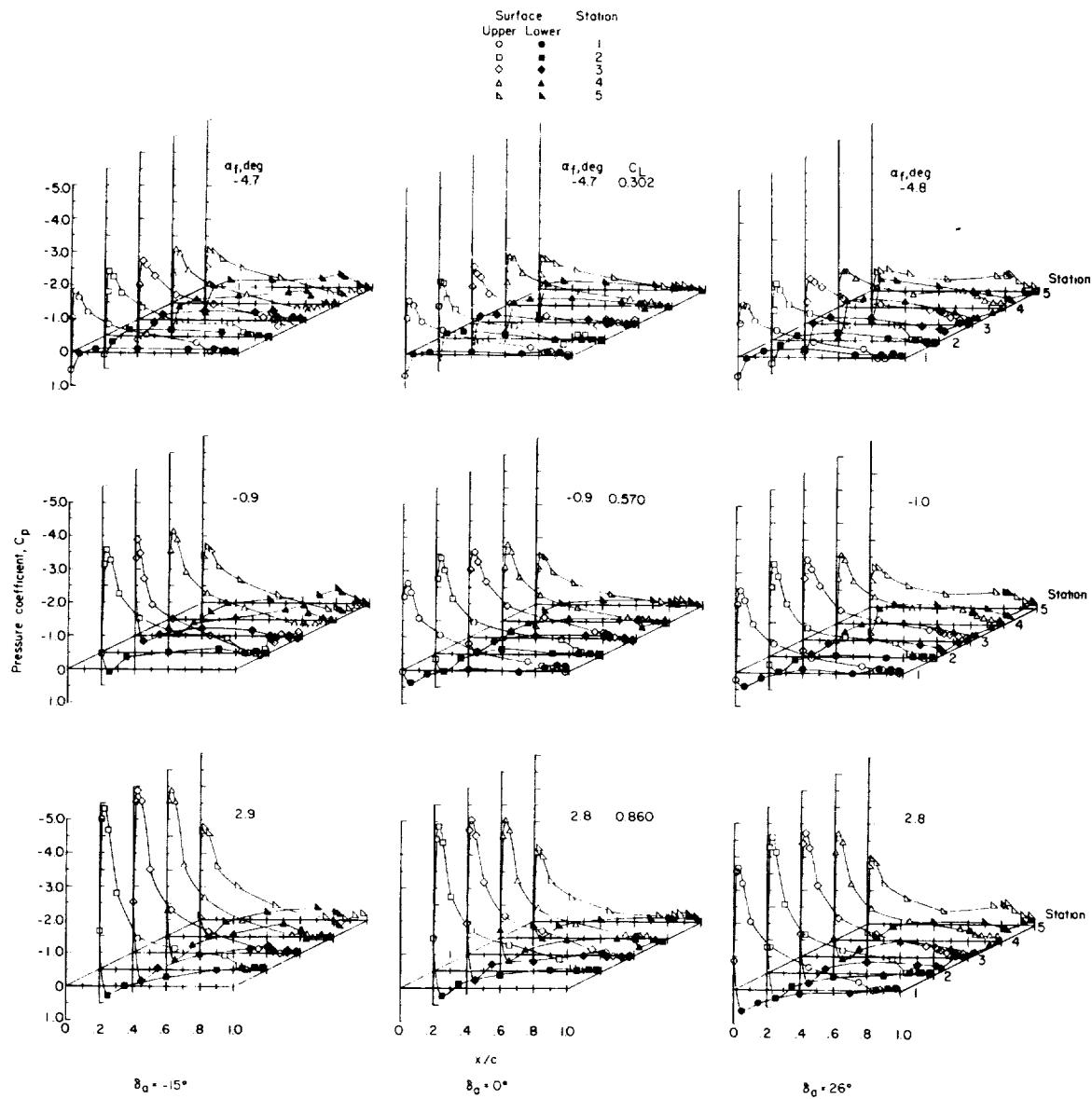
(a) $V_\infty = 94.0 \text{ ft/sec.}$

Figure 17.- Variation of the chordwise pressure distributions with aileron deflection of cable configuration 2-5-8. Belly plate installed; lower forward cables heavily tightened; $p = 7.0 \text{ lb/sq in.}$

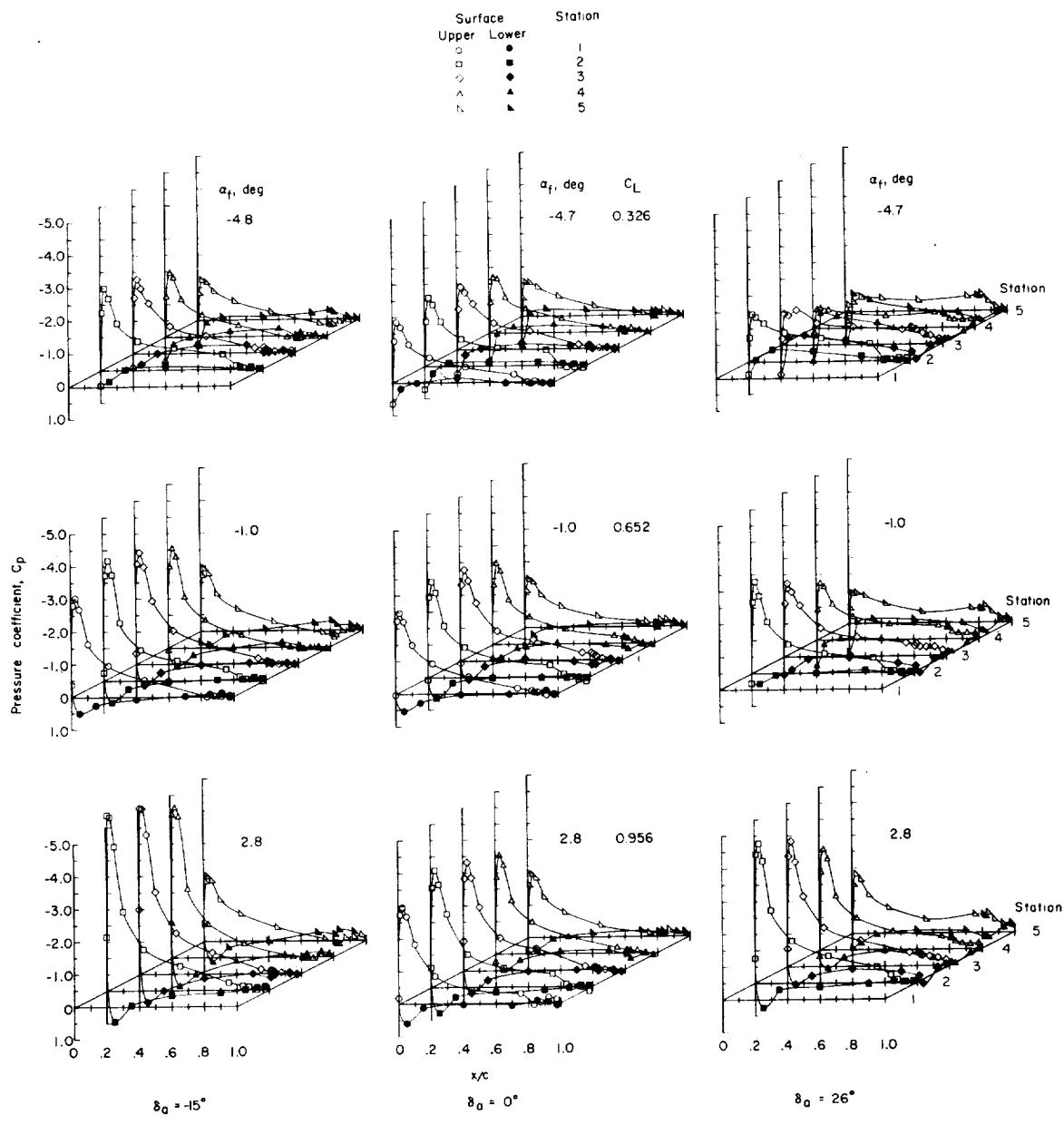


Figure 17.- Concluded.

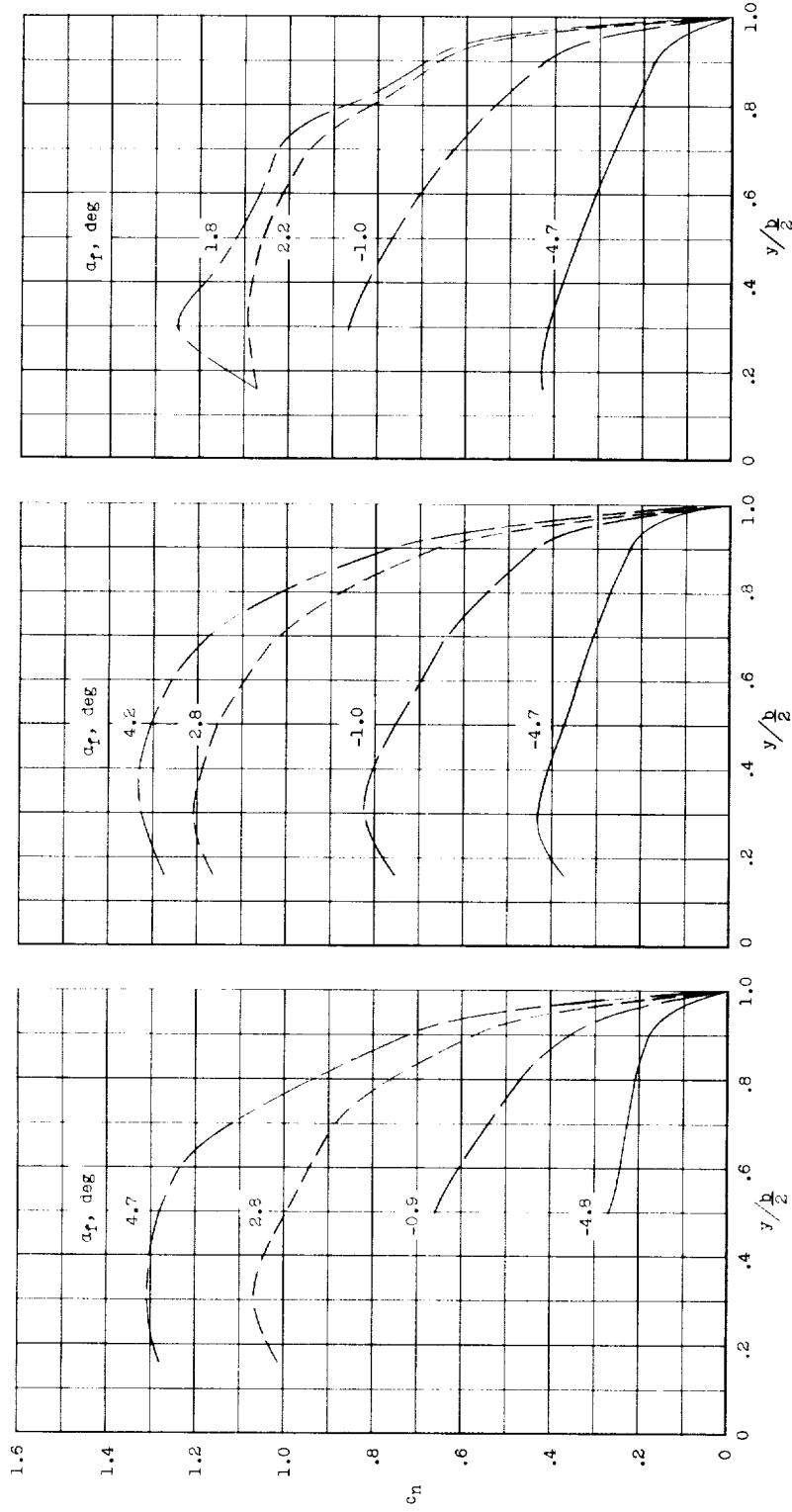
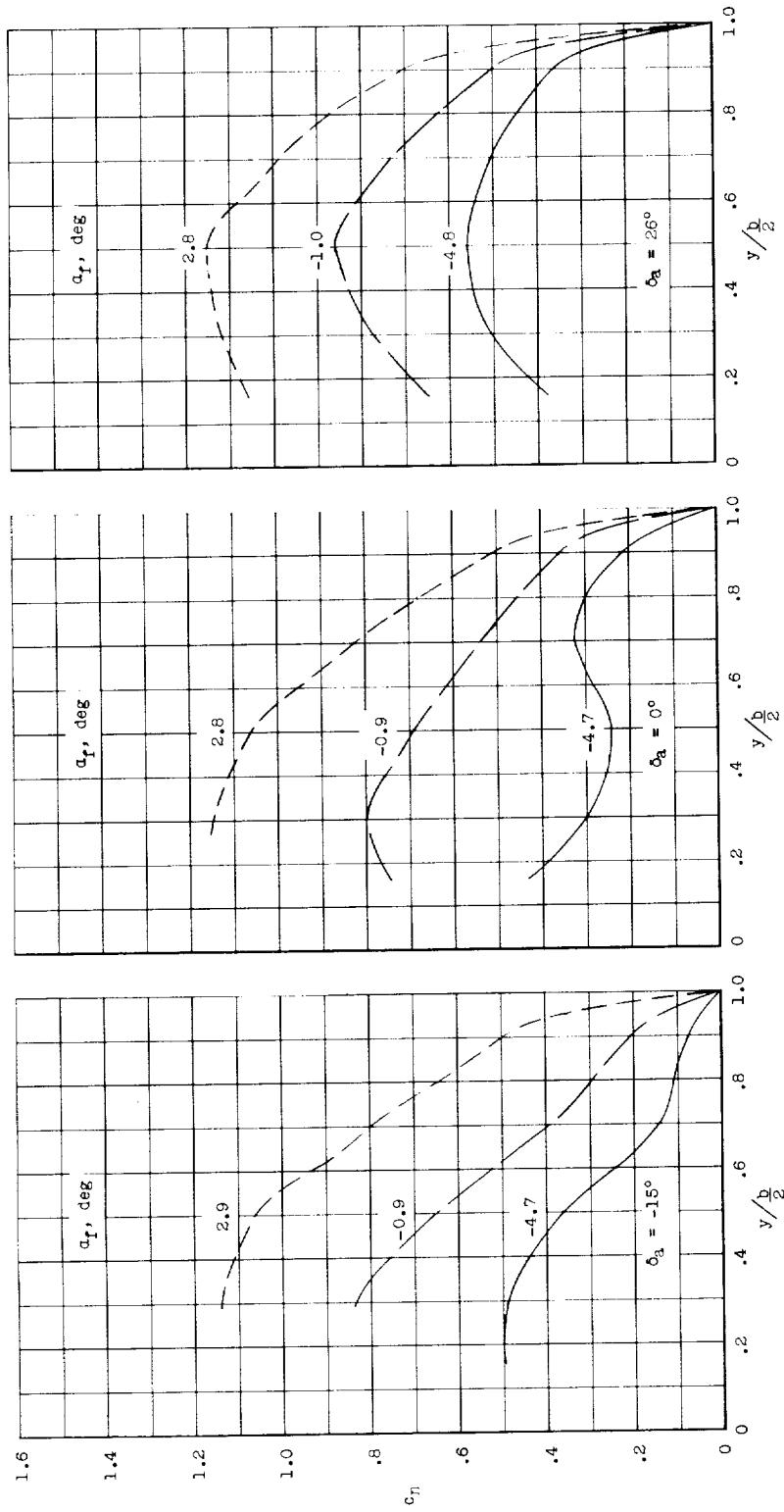


Figure 18.- Span loading characteristics of two cable configurations. Belly plate installed; lower forward cables heavily tightened; $P = 7.0 \text{ lb/sq in.}$



(a) $V_\infty = 94.0 \text{ ft/sec.}$

Figure 19.- Span loading characteristics of cable configuration 2-5-8. Belly plate installed; lower forward cables heavily tightened; $P = 7.0 \text{ lb/sq in.}$

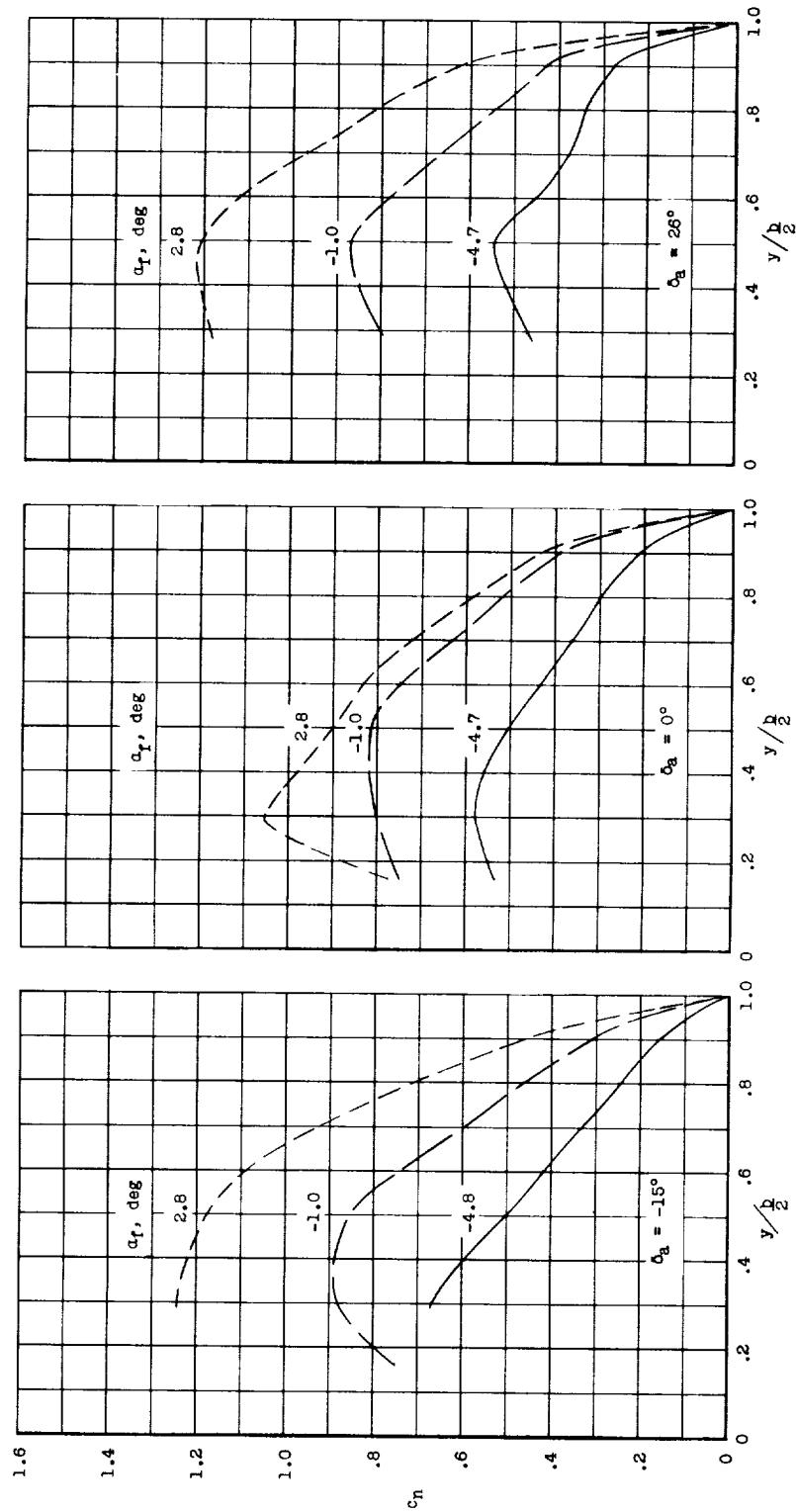
(b) $V_\infty = 113.7 \text{ ft/sec.}$

Figure 19.- Concluded.

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